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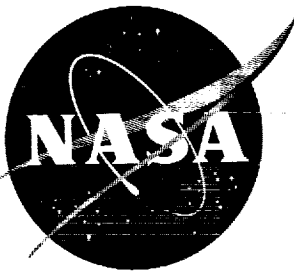
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TECHNICAL MEMORANDUM

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AN INVESTIGATION OF A 0.05-SCALE MODEL OF THE
XSM-64A NAVAHO MISSILE AND BOOSTER
AT SUPERSONIC MACH NUMBERS

PART II - PRESSURE STUDY

By James D. Church and Nancy L. Taylor

Langley Research Center
Langley Field, Va.

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TECHNICAL MEMORANDUM X-17

AN INVESTIGATION OF A 0.05-SCALE MODEL OF THE
XSM-64A NAVAHO MISSILE AND BOOSTER
AT SUPERSONIC MACH NUMBERS

PART II - PRESSURE STUDY*

By James D. Church and Nancy L. Taylor

SUMMARY

An investigation of the pressure distribution over an 0.05-scale model of the XSM-64A Navaho missile and booster has been conducted in the Langley Unitary Plan wind tunnel. Pressures were measured on the missile wing and body and on the booster body. Data were obtained over a Mach number range of 2.29 to 3.22 at a Reynolds number of approximately 2.6×10^6 . Results are presented for an angle-of-attack range of -8° to 4° for the missile-booster combination and -10° to 10° for the missile-alone configuration. Data are also shown at several angles of attack for angles of sideslip from -8° to 8° . The effects of canard deflections of 0° and $\pm 5^\circ$ were obtained for both the missile-booster and the missile-alone configurations. In addition, pressures on a booster pitch thrust chamber with and without a jet simulator are presented. The data are presented in tabular form at nominal angles of attack and sideslip and are not analyzed.

INTRODUCTION

An investigation has been conducted on a 0.05-scale model of the XSM-64A Navaho missile and missile-booster combination in the Langley Unitary Plan wind tunnel. The results of the force testing phase of this investigation and a description of the intended mission of the missile were reported in reference 1. Results from the pressure tests of this study are contained herein.

*Title, Unclassified.

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The purpose of the present phase of the investigation was to determine the pressure distributions on the missile wing and body and on the booster body. Pressures for both the missile-booster combination and the missile-alone configurations were recorded. Pressures were also measured on a booster pitch thrust chamber with and without a jet simulator. Tests were made at Mach numbers of 2.29, 2.75, and 3.22 for both the missile-booster combination and missile alone at Reynolds numbers of 2.39×10^6 , 2.90×10^6 , and 2.69×10^6 , respectively. The angle-of-attack range for the missile-booster combination was approximately -8° to 4° and was -10° to 10° for the missile-alone configuration. Tests for both configurations were conducted through an angle-of-sideslip range of -8° to 8° .

Pressure coefficients are presented in tabular form without analysis.

SYMBOLS

b	span of wing, in.
C_p	pressure coefficient, $\frac{p_m - p}{q}$
M	free-stream Mach number
p	free-stream static pressure, lb/sq ft
p_m	locally measured static pressure, lb/sq ft
q	free-stream dynamic pressure, lb/sq ft
x/l	general nondimensional longitudinal station
x_B/l_B	longitudinal distance rearward from nose and along booster body center line, as a fraction of booster body length
x_M/l_M	longitudinal distance rearward from nose and along missile-fuselage reference line, as a fraction of missile body length
x_N/l_N	longitudinal distance forward from base and along nozzle axis, as a fraction of nozzle length
x_w/c	longitudinal distance rearward from leading edge and in wing chord plane, as a fraction of wing local chord

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- y spanwise distance outboard from and perpendicular to missile center line in wing chord plane, in.
- α angle of attack referred to missile-fuselage reference line, deg
- β angle of sideslip referred to missile center line, deg
- δ_e angle of canard surface relative to missile-fuselage reference line, deg
- δ_{PT} angle of booster pitch thrust chambers relative to axis shown in figure 1(c), deg
- δ_{YT} angle of booster yaw thrust chamber relative to axis shown in figure 1(c), deg
- θ angular location of orifices, deg ($\theta = 0^\circ$ at top of model and increases clockwise when looking forward along the model)

MODEL AND APPARATUS

The tests were performed in the high Mach number test section of Langley Unitary Plan wind tunnel. This tunnel is of the variable-pressure, return-flow type with a test section measuring 4 feet square and approximately 7 feet long. Mach number may be varied continuously from approximately 2.3 to 4.7 by means of an asymmetrical sliding-block nozzle.

Details of the model are shown in figure 1. Reference 1 contains a comprehensive description and table of geometric characteristics for this configuration. The pressure orifices used in the present test are identified by the angle θ and the longitudinal station x/l for body measurements. The value of θ increases in a clockwise direction from 0° at the top of either the missile or booster body when looking forward along the model. (See fig. 1(b).) The nondimensional longitudinal location x/l indicates the rearward locations of the orifices with respect to the nose of either body. Approximately 90 orifices were employed for these measurements. The missile wing orifices, along the upper and lower surfaces, are identified by spanwise $\left(\frac{y}{b/2}\right)$ and chordwise $\left(\frac{x_w}{c}\right)$ location. Spanwise rows of 5 orifices were utilized for these measurements.

The missile was attached to the central tunnel support system by means of several stings, the details of which are illustrated in

reference 1. A rotary coupling and the central support system were employed to obtain varying angles of attack simultaneously with varying angles of sideslip. Tubing contained within the support system connected the model pressure orifices to liquid manometers located just outside the test section. Pressure readings were photographically recorded from the alkazine-filled manometers and then reduced into pressure-coefficient form.

TESTS

Pressure-distribution tests were made for the missile-booster and the missile-alone configurations through an angle-of-attack range of -8° to 4° and -10° to 10° , respectively, at 0° angle of sideslip. At angles of attack of 0° and 5° , tests were conducted on both configurations for an angle-of-sideslip range from -8° to 8° .

Tests were also performed with canard settings (δ_e) of 0° and 5° for the missile-booster combination and with settings of -5° , 0° , and 5° for the missile-alone configuration. Pressures were measured (during the force tests of ref. 1) on the booster pitch thrust chamber to determine the effect of a special fairing and a jet simulator. (See fig. 1(c).) The effect of these latter two items was obtained at various thrust-chamber settings (δ_{PT}) and (δ_{YT}) over both the angle-of-attack and sideslip ranges.

The purpose of the jet-simulator tests was to determine the effect, if any, of flow through the thrust chamber on the nozzle hinge moments of reference 1. Since the model was too small to permit ducting flow through the nozzles, the jet effect was approximated by fastening a wooden jet simulator to the rear of one of the chambers. This block represents the jet boundary calculated for the following conditions: Mach number in the vicinity of the nozzle equal to approximately 3.2, jet-exit Mach numbers equal to approximately 2.7, jet pressure ratio equal to 9, and ratio of specific heats in jet equal to 1.26.

Average test conditions are as follows:

M	Stagnation pressure, lb/sq in. abs	Dynamic pressure, lb/sq ft	Reynolds number
2.29	11.1	476	2.39×10^6
2.75	17.1	518	2.90
3.22	20.4	419	2.69

The Reynolds number is based on the mean aerodynamic chord of the missile wing. The stagnation temperature was maintained at 150° F for all Mach numbers.

CORRECTIONS AND ACCURACY

Tunnel calibrations indicate that, although no significant pressure discontinuities exist, some flow angularity is present in the test section. Values of the flow angle in the vertical plane of the tunnel have been found to be 0.35°, 0.25°, and 0.10°, respectively, at the test Mach numbers of 2.29, 2.75, and 3.22. Inasmuch as the model wings were aligned with the tunnel vertical plane at $\beta = 0^\circ$, there is no flow correction to α . However, both angles of sideslip and attack would require corrections at $\beta \neq 0^\circ$ for a portion of the previously listed corrections, dependent upon the angle of the roll coupling. Neither angle of attack nor angle of sideslip has been corrected for deflections of the modeling combination under static loads; however, the results for the more flexible arrangement of reference 1 (with balance) indicate the maximum error due to deflection to be approximately $\pm 0.2^\circ$. Maximum deviation of local Mach number in the portion of the tunnel occupied by the model was about ± 0.02 . Although random pressure-coefficient errors due to photographic recording are only about ± 0.005 , the errors arising from the local Mach number deviations are much larger. This effect, since C_p is a function of p_m , p , and q , varies with the magnitude of pressure coefficient that is to be measured. For example, a value of $C_p = 1.80$ can be determined to within 3 percent; however, for $C_p = 0.05$, the error can get as large as 20 percent. Except for very small values of C_p and for regions of large pressure gradients, the probable error is believed to be within 2 percent.

PRESENTATION OF RESULTS

The data obtained on the pressure-distribution tests are presented in the following tables:

	Table
Pressure coefficients measured on -	
Missile body (in presence of booster)	I
Missile wing (in presence of booster)	II
Booster body (in presence of missile)	III
Missile body	IV
Missile wing	V
Booster pitch thrust chamber	VI

Langley Research Center,
National Aeronautics and Space Administration,
Langley Field, Va., March 4, 1959.

REFERENCE

1. Church, James D., and Taylor, Nancy L.: An Investigation of a 0.05-Scale Model of the XSM-64A Navaho Missile and Booster at Supersonic Mach Numbers. Part I - Force Study. NASA TM X-16, 1959.

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER

Body station $\frac{x}{M}$ $\frac{z}{M}$	C_p at $\theta =$ —						Body station $\frac{x}{M}$ $\frac{z}{M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 2.29; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.143					.001		.095
.124	.195					-.028		.124
.153	.212					-.047		.153
.191	.158			.014		-.060		.191
.267				-.004		.222		.267
.344				.078		.204		.344
.439				.028		.080		.439
.538		.166	.141		.022		-.021	.538
.556		.105	-.010		-.060		-.059	.556
.592	.144	.071	-.025		-.044	.049	-.016	.592
.668	.099	.093	.117			-.118	-.111	.668
.744	.078	.102	-.108		.115		-.107	.744
.821	.095				-.084			.821
.897	.123	.107	.048		-.098		-.073	.897
.945		.086						.945
.989		-.005	-.066		-.086		-.090	.989

$M = 2.29; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.114					.003		.095
.124	.159					-.016		.124
.153	.167					-.033		.153
.191	.116			.020		-.044		.191
.267				.010		.186		.267
.344				.087		.128		.344
.439				.018		.075		.439
.538		.123	.149		.039		-.017	.538
.556		.081	-.009		-.054		-.060	.556
.592	.118	.053	-.021		-.039	.042	-.021	.592
.668	.076	.065	.095			-.120	-.108	.668
.744	.051	.071	-.107		.088		-.081	.744
.821	.063				-.070			.821
.897	.086	.071	.020		-.093		-.046	.897
.945		.052						.945
.989		-.026	-.064		-.086		-.095	.989

$M = 2.29; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.089					.015		.095
.124	.122					.001		.124
.153	.126					-.015		.153
.191	.078			.024		-.024		.191
.267				.017		.148		.267
.344				.088		.147		.344
.439				.009		.066		.439
.538		.105	.157		.031		-.019	.538
.556		.042	-.012		-.049		-.058	.556
.592	.098	.033	-.019		-.034	.037	-.023	.592
.668	.056	.046	.060			-.120	-.099	.668
.744	.031	.045	-.090		.061		-.045	.744
.821	.036				-.038			.821
.897	.057	.043	-.006		-.076		-.010	.897
.945		.020						.945
.989		-.046	-.062		-.083		-.113	.989

$M = 2.29; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.069					.030		.095
.124	.087					.022		.124
.153	.086					.009		.153
.191	.042			.024		-.003		.191
.267				.040		.155		.267
.344				.085		.150		.344
.439				-.001		.057		.439
.538		.080	.155		.021		-.021	.538
.556		.017	-.013		-.053		-.055	.556
.592	.080	.013	-.030		-.035	.031	-.022	.592
.668	.038	.027	.031			-.119	-.094	.668
.744	.015	.023	-.068		.030		-.023	.744
.821	.013				-.017			.821
.897	.025	.014	-.029		-.057		.016	.897
.945		-.006						.945
.989		-.063	-.057		-.071		-.095	.989

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{l_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{l_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095	.068					.029	.095	
.124	.086					.021	.124	
.153	.085					.009	.153	
.191	.041			.023		-.004	.191	
.267				.039		.154	.267	
.344				.084		.149	.344	
.439				-.002		.057	.439	
.538		.079	.154		.021		.538	
.556		.017	-.015		-.054		.556	
.592	.079	.012	-.029		-.034	.030	.592	
.668	.037	.027	.031			-.118	.668	
.744	.014	.023	-.067		.031		.744	
.821	.013				-.017		.821	
.897	.026	.014	-.029		-.058	.016	.897	
.945		-.006					.945	
.989		-.063	-.058		-.072	-.095	.989	
$M = 2.29; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095	.032					.064	.095	
.124	.024					.071	.124	
.153	.006					.063	.153	
.191	-.018			.020		.049	.191	
.267				.045		.112	.267	
.344				.058		.173	.344	
.439				-.022		.040	.439	
.538		.050	.085		.029		.538	
.556		-.023	-.036		-.043	-.043	.556	
.592	.048	-.022	-.048		-.038	-.008	.592	
.668	.006	-.006	-.018			-.099	.668	
.744	-.013	-.014	-.007		-.027	.033	.744	
.821	-.024				.053		.821	
.897	-.017	-.028	-.074		-.024	.064	.897	
.945		-.047					.945	
.989		-.107	-.054		-.061	-.060	.989	
$M = 2.29; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095	.016					.083	.095	
.124	.001					.101	.124	
.153	-.022					.098	.153	
.191	-.046			.014		.081	.191	
.267				.027		.115	.267	
.344				.036		.170	.344	
.439				-.034		.037	.439	
.538		.009	.052		.038		.538	
.556		-.046	-.046		-.036	-.029	.556	
.592	.029	-.045	-.055		-.039	.006	.592	
.668	-.009	-.018	-.049			-.083	.668	
.744	-.026	-.032	.024		-.052	.061	.744	
.821	-.040				.091		.821	
.897	-.045	-.057	-.091		-.014	.103	.897	
.945		-.065					.945	
.989		-.115	-.050		-.064	-.042	.989	
$M = 2.29; \alpha = -5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$								
.095	.066					-.028	.095	
.124	.083					-.047	.124	
.153	.092					-.062	.153	
.191	.071			-.009		-.077	.191	
.267				-.007		.120	.267	
.344				.043		.098	.344	
.439				-.010		.021	.439	
.538		.074	.071		-.102		.538	
.556		.046	-.058		-.128	-.104	.556	
.592	.041	.035	-.079		-.051	-.157	.592	
.668	.042	.037	.028			-.157	.668	
.744	.036	.018	-.103		.034	-.114	.744	
.821	.035				-.078		.821	
.897	.059	.007	-.001		-.074	-.108	.897	
.945		-.005					.945	
.989		-.132	-.091		-.090	-.117	.989	

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{z_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{z_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.29; \alpha = -5^\circ; \beta = -6^\circ; \delta_e = 0^\circ$								
.095	.082					-.012	.095	
.124	.113					-.029	.124	
.153	.116					-.045	.153	
.191	.083			-.005		-.057	.191	
.267				-.001		.133	.267	
.344				.053		.114	.344	
.439				-.004		.041	.439	
.538		.087	.075		-.083		.538	
.556		.056	-.051		-.124	-.075	.556	
.592	.079	.036	-.064		-.074	-.072	.592	
.668	.048	.049	.055		-.054	-.114	.668	
.744	.042	.031	-.110		-.086	-.181	.744	
.821	.041			.038		-.099	.821	
.897	.067	.026	.007	-.078			.897	
.945		.011		-.083		-.052	.945	
.989		-.118	-.065		-.081	-.123	.989	
$M = 2.29; \alpha = -5^\circ; \beta = -4^\circ; \delta_e = 0^\circ$								
.095	.092					-.002	.095	
.124	.138					-.018	.124	
.153	.131					-.033	.153	
.191	.088			.002		-.044	.191	
.267				.000		.146	.267	
.344				.061		.100	.344	
.439				.000		.049	.439	
.538		.096	.099		-.026		.538	
.556		.067	-.043		-.091	-.087	.556	
.592	.097	.035	-.048		-.067	-.039	.592	
.668	.058	.049	.067		.051	-.089	.668	
.744	.040	.039	-.121		-.098	-.182	.744	
.821	.044			.056		-.046	.821	
.897	.069	.043	.009	-.097			.897	
.945		.021		-.074		-.066	.945	
.989		-.090	-.062		-.071	-.118	.989	
$M = 2.29; \alpha = -5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$								
.095	.099					.006	.095	
.124	.132					-.009	.124	
.153	.141					-.024	.153	
.191	.094			.013		-.036	.191	
.267				.006		.157	.267	
.344				.074		.114	.344	
.439				.002		.062	.439	
.538		.109	.123		.004		.538	
.556		.066	-.023		-.067	-.077	.556	
.592	.107	.040	-.031		-.045	-.060	.592	
.668	.063	.051	.072			.035	.668	
.744	.040	.048	-.116		.067	-.154	.744	
.821	.049				-.084	-.066	.821	
.897	.071	.057	.006		-.085		.897	
.945		.031				-.050	.945	
.989		-.058	-.068		-.093	-.101	.989	
$M = 2.29; \alpha = -5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$								
.095	.099					.008	.095	
.124	.137					-.008	.124	
.153	.143					-.024	.153	
.191	.093			.017		-.034	.191	
.267				.009		.160	.267	
.344				.080		.128	.344	
.439				.005		.066	.439	
.538		.110	.137		.018		.538	
.556		.061	-.014		-.058	-.033	.556	
.592	.107	.043	-.025		-.039	-.076	.592	
.668	.064	.052	.073			-.037	.668	
.744	.042	.055	-.108		.071	-.129	.744	
.821	.049				-.071	-.063	.821	
.897	.071	.057	.003		-.092		.897	
.945		.045				-.052	.945	
.989		-.051	-.063		-.096	-.118	.989	

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x}{L}$ $\frac{z}{L}$	C_p at $\theta = \text{---}$						Body station $\frac{x}{L}$ $\frac{z}{L}$
	0°		60°	90°	120°	180°	
	Body center line	Nacelle center line				Body center line	Nacelle center line
$M = 2.29; \alpha = -5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.101 .139 .146 .096			.024 .016 .090 .029		.009 .005 -.021 -.031 .163 .124 .073	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
		.117 .060 .043 .053 .055 .070 .058 .037 -.036	.155 -.009 -.023 .073 -.095 .002 -.063		.033 -.049 -.039 .075 -.051 -.083 -.089		-.017 -.059 -.023 -.105 -.058 -.026 -.119
$M = 2.29; \alpha = -5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.100 .137 .144 .096			.031 .020 .099 .022		.009 -.007 -.023 -.034 .162 .099 .070	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
	.107 .063 .038 .068 .055 .068 .034 -.022	.121 .061 .043 .059 .078 .059 .055 .034 -.022	.163 -.001 -.008 .078 -.088 .006 -.054		.049 -.041 -.040 .078 -.039 -.079 -.090		-.011 -.048 .001 -.091 -.089 .030 -.151
$M = 2.29; \alpha = -5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.101 .135 .144 .097			.037 .024 .109 .028		.006 -.010 -.026 -.037 .160 .088 .062	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
	.104 .058 .036 .041 .060	.112 .060 .042 .054 .060 .053 .033 -.008	.172 .006 .002 .085 -.081 .010 -.051		.066 -.036 -.036 .080 -.027 -.069 -.095		-.005 -.038 .021 -.083 -.076 -.012 -.149
$M = 2.29; \alpha = -5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.099 .141 .135 .095			.050 .034 .130 .064		-.001 -.016 -.033 -.045 .151 .082 .054	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
	.092 .050 .035 .034 .054	.113 .066 .045 .048 .057 .046 .028 .009	.191 .028 .015 .096 -.070 .015 -.048		.084 -.020 -.023 .079 -.008 -.053 -.092		.004 -.020 -.002 -.063 -.039 -.034 -.163

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TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{l_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{l_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 2.29; \alpha = -5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.095	.088					-.013		.095
.124	.120					-.029		.124
.153	.119					-.046		.153
.191	.086			.074		-.059		.191
.267				.050		.138		.267
.344				.157		.086		.344
.439				.086		.042		.439
.538		.131	.236		.121		.013	.538
.556		.073	.053		.005		.014	.556
.592	.072	.051	.036		-.001	-.050	.003	.592
.668	.036	.031	.109			-.087	-.041	.668
.744	.033	.049	-.050		.079		.018	.744
.821	.026				.032			.821
.897	.045	.036	.021		-.029		-.022	.897
.945		.026						.945
.989		.025	-.042		-.089		-.176	.989

$M = 2.29; \alpha = -5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.095	.073					-.030		.095
.124	.090					-.050		.124
.153	.094					-.066		.153
.191	.075			.101		-.082		.191
.267				.066		.116		.267
.344				.187		.075		.344
.439				.109		.023		.439
.538		.162	.281		.140		.030	.538
.556		.084	.082		.031		.036	.556
.592	.038	.053	.058		.024	-.049	.011	.592
.668	.031	.020	.117			-.087	-.012	.668
.744	.022	.040	-.029		.086		.055	.744
.821	.016				.072			.821
.897	.034	.023	.024		-.005		-.014	.897
.945		.022						.945
.989		.042	-.032		-.076		-.177	.989

$M = 2.29; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.095	.006					.001		.095
.124	.005					-.006		.124
.153	-.001					-.024		.153
.191	-.011			.091		-.035		.191
.267				.080		.084		.267
.344				.185		.045		.344
.439				.083		-.064		.439
.538		.073	.212		.153		.032	.538
.556		-.011	.052		.043		.029	.556
.592	.029	-.024	.013		.024	-.049	.020	.592
.668	.000	-.046	.023			-.078	-.015	.668
.744	-.014	-.023	.013		.000		.096	.744
.821	-.026				.125			.821
.897	-.020	-.030	-.030		.032		.030	.897
.945		-.024						.945
.989		-.004	-.056		-.071		-.157	.989

$M = 2.29; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.095	.021					.015		.095
.124	.022					.012		.124
.153	.014					-.002		.153
.191	-.004			.066		-.014		.191
.267				.059		.087		.267
.344				.150		.056		.344
.439				.056		.017		.439
.538		.066	.178		.114		.007	.538
.556		-.014	.028		.012		-.002	.556
.592	.036	-.016	-.009		-.006	-.060	.001	.592
.668	.004	-.029	.015			-.080	-.038	.668
.744	-.008	-.012	-.009		-.006		.057	.744
.821	-.018				.090			.821
.897	-.007	-.018	-.036		.014		.020	.897
.945		-.027						.945
.989		-.020	-.056		-.077		-.156	.989

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TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta = \text{---}$						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.29; \alpha = 0^\circ; \beta = 4^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.038 .039 .032 .006 .054 .012 -.001 -.002 -.011 -.002	 .072 -.009 -.015 -.003 -.004 -.012 -.029 -.038	 .160 .006 -.019 .012 -.022 -.042 -.058	 .052 .052 .126 .038 				

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{z_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{z_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 2.29; \alpha = 0^\circ; \beta = -1^\circ; \delta_e = 0^\circ$

.095	.051					.044	.095
.124	.053					.042	.124
.153	.035					.031	.153
.191	.012			.018		.016	.191
.267				.048		.152	.267
.344				.064		.149	.344
.439				-.011		.039	.439
.538		.077	.109		-.002		.538
.556		-.004	-.034		-.063	-.023	.556
.592	.063	-.003	-.047		-.043	-.068	.592
.668	.021	.013	.012			-.044	.668
.744	.007	.004	-.063		-.001	-.116	.744
.821	-.003				-.006	-.029	.821
.897	.006	-.006	-.054		-.059	-.015	.897
.945		-.021					.945
.989		-.098	-.061		-.076	-.094	.989

$M = 2.29; \alpha = 0^\circ; \beta = -2^\circ; \delta_e = 0^\circ$

.095	.050					.043	.095
.124	.051					.041	.124
.153	.036					.030	.153
.191	.010			.015		.016	.191
.267				.044		.124	.267
.344				.056		.150	.344
.439				-.015		.036	.439
.538		.074	.104		-.012		.538
.556		.005	-.040		-.069	-.066	.556
.592	.062	.001	-.053		-.044	-.064	.592
.668	.021	.013	.014			-.061	.668
.744	.006	.005	-.068		-.001	-.137	.744
.821	-.002				-.008	-.039	.821
.897	.006	-.006	-.055		-.049	-.017	.897
.945		-.026					.945
.989		-.104	-.061		-.075	-.097	.989

$M = 2.29; \alpha = 0^\circ; \beta = -4^\circ; \delta_e = 0^\circ$

.095	.042					.036	.095
.124	.041					.032	.124
.153	.033					.027	.153
.191	.005			.007		.009	.191
.267				.040		.104	.267
.344				.044		.139	.344
.439				-.025		.030	.439
.538		.076	.083		-.025		.538
.556		.026	-.057		-.088	-.083	.556
.592	.054	.009	-.062		-.057	-.066	.592
.668	.018	.012	.016			-.072	.668
.744	.005	-.001	-.046		-.001	-.163	.744
.821	-.006				-.019	-.070	.821
.897	.003	-.015	-.055		-.045	-.032	.897
.945		-.037					.945
.989		-.120	-.064		-.069	-.105	.989

$M = 2.29; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$

.095	.028					.022	.095
.124	.031					.019	.124
.153	.018					.010	.153
.191	-.002			-.002		-.006	.191
.267				.037		.093	.267
.344				.035		.115	.344
.439				-.033		.021	.439
.538		.099	.048		-.077		.538
.556		.035	-.075		-.123	-.072	.556
.592	.040	.014	-.078		-.050	-.047	.592
.668	.012	.005	.015			-.099	.668
.744	-.004	-.015	-.042		-.005	-.178	.744
.821	-.016				-.036	-.100	.821
.897	-.006	-.027	-.053		-.046	-.055	.897
.945		-.048					.945
.989		-.144	-.069		-.074	-.111	.989

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x}{L}$		C_p at $\theta = -$						Body station $\frac{x}{L}$		
		0°		60°	90°	120°	180°			
		Body center line	Nacelle center line				Body center line			Nacelle center line
$M = 2.29; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.010 .008 -.000 -.014 .026 .006 -.011 -.025 -.016 									

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta = -$						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.29; \alpha = 5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$								
.095	.003					.092	.095	
.124	-.017					.109	.124	
.153	-.039					.109	.153	
.191	-.063			-.003		.096	.191	
.267				.004		.119	.267	
.344				.006		.158	.344	
.439				-.046		.027	.439	
.538		.007	.024		.022		.538	
.556		-.044	-.070		-.060		.556	
.592	.012	-.045	-.061		-.054		.592	
.668	-.023	-.026	-.058			.035	.668	
.744	-.032	-.048	.037		-.067	-.080	.744	
.821	-.059				.088		.821	
.897	-.061	-.068	-.111		.031		.897	
.945		-.088				.061	.945	
.989		-.137	-.070		-.057	-.057	.989	
$M = 2.29; \alpha = 5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$								
.095	.004					.092	.095	
.124	-.015					.111	.124	
.153	-.039					.111	.153	
.191	-.062			.001		.098	.191	
.267				.008		.119	.267	
.344				.012		.164	.344	
.439				-.045		.025	.439	
.538		-.003	.028		.028		.538	
.556		-.058	-.062		-.048		.556	
.592	.015	-.050	-.062		-.048		.592	
.668	-.022	-.027	-.063			.053	.668	
.744	-.031	-.047	.032		-.070	-.077	.744	
.821	-.058				.087		.821	
.897	-.058	-.072	-.108		-.001		.897	
.945		-.087				.051	.945	
.989		-.133	-.067		-.057	-.052	.989	
$M = 2.29; \alpha = 5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095	.005					.092	.095	
.124	-.015					.113	.124	
.153	-.038					.114	.153	
.191	-.062			.006		.095	.191	
.267				.013		.119	.267	
.344				.019		.159	.344	
.439				-.041		.030	.439	
.538		-.014	.036		.042		.538	
.556		-.063	-.056		-.037		.556	
.592	.017	-.054	-.064		-.043		.592	
.668	-.020	-.028	-.070			.052	.668	
.744	-.030	-.045	.045		-.069	-.082	.744	
.821	-.056				.100		.821	
.897	-.055	-.071	-.105		-.012		.897	
.945		-.082				.121	.945	
.989		-.125	-.055		-.063	-.038	.989	
$M = 2.29; \alpha = 5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$								
.095	.005					.094	.095	
.124	-.015					.114	.124	
.153	-.038					.112	.153	
.191	-.063			.013		.098	.191	
.267				.020		.118	.267	
.344				.031		.163	.344	
.439				-.036		.026	.439	
.538		-.025	.044		.056		.538	
.556		-.071	-.052		-.025		.556	
.592	.016	-.057	-.062		-.037		.592	
.668	-.021	-.030	-.075			.039	.668	
.744	-.030	-.047	.060		-.070	-.082	.744	
.821	-.055				.118		.821	
.897	-.054	-.068	-.102		.000		.897	
.945		-.077				.161	.945	
.989		-.119	-.049		-.067	-.044	.989	

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —							Body station $\frac{x_M}{L_M}$
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line	Nacelle center line	
$M = 2.29; \alpha = 5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.001 -.017 -.040 -.064 .012 -.024 -.032 -.057 -.055	 -.031 -.078 -.065 -.034 -.050 -.067 -.077 -.115	 .054 -.049 -.062 -.080 .068 -.102 -.052	 				

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{l_M}$	C_p at $\theta = -$						Body station $\frac{x_M}{l_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.29; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.196 .284 .269 .202 .132 .090 .068 .088 .119 							

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x}{L}$ $\frac{z}{L}$		C_p at $\theta =$ —						Body station $\frac{x}{L}$ $\frac{z}{L}$		
		0°		60°	90°	120°	180°			
		Body center line	Nacelle center line				Body center line			Nacelle center line
$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = -5^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.070 .092 .092 .041				.017 .058 .068 -.012		.044 .027 .013 .001 .149 .116 .054		.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
		.051 -.021 -.023 .008 -.001	.117 -.014 -.037 .001 -.035		.038 -.038 -.032		.000 -.044 -.013 -.085 .010			
	.057 .017 .000 -.008 -.001					.040 -1.108				
		-.012 -.028 -.085	-.050 -.055		.000 .017 -.040 -.068		.040 -.078			
$M = 2.29; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.045 .057 .054 .009				.014 .046 .054 -.022		.062 .053 .041 .028 .148 .120 .049		.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
		.024 -.037 -.035 -.005 -.019	.085 -.029 -.044 -.028 -.003		.042 -.034 -.033		.008 -.035 -.003 -.079 .037			
	.042 .002 -.010 -.027 -.022					-.096				
		-.034 -.042 -.102	-.071 -.049		-.027 .052 -.024 -.061		.066 -.060			
$M = 2.75; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.123 .173 .201 .159				.014 .000 .078 .045		.004 -.012 -.027 -.037 .008 .165 .138		.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
		.164 .114 .086 .100 .090	.180 .035 .008 .103 -.049		.037 -.015 -.008		.050 -.036 .102 -.062 -.084			
	.134 .102 .082 .077 .103				.104 -.076 -.055	.098 -.032				
		.092 .072 .014	.048 -.030		-.072		-.055 -.111			
$M = 2.75; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.097 .137 .154 .119				.019 .010 .077 .050		.006 -.005 -.019 -.030 -.007 .157 .131		.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
		.099 .067 .056 .078 .056	.150 .013 -.005 .084 -.052		.077 -.021 -.014		.040 -.041 .103 -.062 -.087			
	.119 .078 .056 .051 .071				.078 -.077 -.055	-.036				
		.063 .044 -.006	.025 -.033		-.076		-.052 -.121			

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta = \text{---}$						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.75; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.077 .105 .113 .085 .097 .060 .040 .030 .046 							

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x}{L}$ $\frac{z}{L}$	C_p at $\theta =$ —							Body station $\frac{x}{L}$ $\frac{z}{L}$
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line	Nacelle center line	
$M = 2.75; \alpha = 40^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.013 .010 -.005 -.027 .029 .000 -.009 -.027 -.029	 <						

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{z_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{z_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 2.75; \alpha = -5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$

.095	.085					.003	.095
.124	.114					-.002	.124
.153	.129					-.013	.153
.191	.100			.010		-.025	.191
.267				.005		-.013	.267
.344				.061		.129	.344
.439				.038		.123	.439
.538		.077	.144		.031		.538
.556		.042	.005		-.037	.001	.556
.592	.105	.039	-.019		-.026	.084	.592
.668	.067	.061	.060		-.037	-.110	.668
.744	.047	.045	-.065		.059	-.123	.744
.821	.041				-.077		.821
.897	.055	.047	.010		-.057	-.048	.897
.945		.035					.945
.989		-.034	-.043		-.076	-.101	.989

$M = 2.75; \alpha = -5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$

.095	.085					.007	.095
.124	.118					.001	.124
.153	.131					-.010	.153
.191	.100			.015		-.022	.191
.267				.010		-.009	.267
.344				.071		.145	.344
.439				.045		.120	.439
.538		.090	.148		.051		.538
.556		.041	.013		-.024	.028	.556
.592	.107	.042	-.011		-.022	.072	.592
.668	.068	.059	.064			-.085	.668
.744	.047	.049	-.055		.063	-.101	.744
.821	.041				-.072		.821
.897	.057	.045	.010		-.053	-.039	.897
.945		.039					.945
.989		-.029	-.038		-.081	-.104	.989

$M = 2.75; \alpha = -5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.087					.009	.095
.124	.121					.002	.124
.153	.134					-.010	.153
.191	.103			.021		-.022	.191
.267				.015		-.008	.267
.344				.078		.152	.344
.439				.053		.125	.439
.538		.101	.151		.070		.538
.556		.048	.020		-.013	.037	.556
.592	.109	.048	-.004		-.018	.095	.592
.668	.069	.060	.071			-.065	.668
.744	.049	.053	-.051		.067	-.085	.744
.821	.041				-.072		.821
.897	.057	.049	.012		-.049	-.037	.897
.945		.034					.945
.989		-.016	-.035		-.075	-.119	.989

$M = 2.75; \alpha = -5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$

.095	.085					.008	.095
.124	.118					.001	.124
.153	.132					-.009	.153
.191	.100			.027		-.022	.191
.267				.019		-.010	.267
.344				.081		.147	.344
.439				.059		.127	.439
.538		.105	.158		.088		.538
.556		.050	.028		-.003	.049	.556
.592	.106	.051	.005		-.012	.105	.592
.668	.066	.060	.074			-.042	.668
.744	.046	.055	-.044		.068	-.070	.744
.821	.038				-.076		.821
.897	.055	.049	.012		-.037	.006	.897
.945		.030					.945
.989		-.004	-.031		-.071	-.122	.989

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{z_M}$	C_p at $\theta = -$						Body station $\frac{x_M}{z_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.75; \alpha = -5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.084 .115 .130 .099 .103 .063 .044 .036 .048 	 						

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{z_M}$	C_p at $\theta =$ —							Body station $\frac{x_M}{z_M}$
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line	Nacelle center line	

$M = 2.75; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.095	.011					.004		.095
.124	.008					.000		.124
.153	-.001					-.012		.153
.191	-.006			.101		-.023		.191
.267				.086		.099		.267
.344				.102		.107		.344
.439				.118		.004		.439
.538		.124	.251		.188		.109	.538
.556		.017	.095		.080		.026	.556
.592	.026	.009	.051		.066	.051	.148	.592
.668	.013	-.032	.044			-.057	.054	.668
.744	.001	-.041	.057		.018		.103	.744
.821	-.011				.021			.821
.897	-.009	-.015	-.022		.117		.132	.897
.945		-.022						.945
.989		-.004	-.035		-.046		-.106	.989

$M = 2.75; \alpha = 0^\circ; \beta = 6^\circ; \delta_e = 0^\circ$

.095	.022					.017		.095
.124	.026					.013		.124
.153	.015					.004		.153
.191	.009			.076		-.004		.191
.267				.066		.105		.267
.344				.102		.112		.344
.439				.092		-.005		.439
.538		.107	.216		.149		.087	.538
.556		.008	.070		.052		.003	.556
.592	.042	.005	.028		.041	.037	.126	.592
.668	.019	-.019	.033			-.069	.036	.668
.744	.006	-.009	.030		.006		.068	.744
.821	-.008				-.001			.821
.897	-.003	-.010	-.024		.093		.098	.897
.945		-.022						.945
.989		-.016	-.034		-.055		-.110	.989

$M = 2.75; \alpha = 0^\circ; \beta = 4^\circ; \delta_e = 0^\circ$

.095	.022					.016		.095
.124	.025					.012		.124
.153	.014					.004		.153
.191	.009			.075		-.005		.191
.267				.065		.104		.267
.344				.101		.111		.344
.439				.091		-.005		.439
.538		.106	.215		.148		.086	.538
.556		.007	.068		.052		.002	.556
.592	.041	.004	.027		.040	.036	.125	.592
.668	.018	-.020	.031			-.069	.035	.668
.744	.005	-.010	.029		.005		.067	.744
.821	-.008				-.002			.821
.897	-.003	-.011	-.023		.094		.098	.897
.945		-.021						.945
.989		-.015	-.034		-.054		-.110	.989

$M = 2.75; \alpha = 0^\circ; \beta = 2^\circ; \delta_e = 0^\circ$

.095	.044					.036		.095
.124	.045					.034		.124
.153	.043					.030		.153
.191	.023			.039		.021		.191
.267				.037		.105		.267
.344				.073		.115		.344
.439				.051		.024		.439
.538		.088	.153		.083		.040	.538
.556		.000	.028		.003		-.031	.556
.592	.064	.012	-.006		-.005	.063	.074	.592
.668	.030	.014	.019			-.052	-.026	.668
.744	.014	.010	-.001		.000		-.010	.744
.821	.002				-.038			.821
.897	.007	.000	-.030		.020		.061	.897
.945		-.015						.945
.989		-.035	-.036		-.059		-.116	.989

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{z_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{z_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 2.75; \alpha = 0^\circ; \beta = 1^\circ; \delta_e = 0^\circ$

.095	.046					.039		.095
.124	.049					.036		.124
.153	.040					.031		.153
.191	.026			.031		.021		.191
.267				.031		.105		.267
.344				.066		.121		.344
.439				.042		.028		.439
.538		.087	.141		.067		.027	.538
.556		.001	.020		-.009		-.037	.556
.592	.067	.016	-.011		-.015	.075	.055	.592
.668	.033	.020	.018			-.044	-.037	.668
.744	.015	.011	-.011		.001		-.034	.744
.821	.002				-.044			.821
.897	.008	.000	-.031		.005		.049	.897
.945		-.016						.945
.989		-.042	-.039		-.060		-.108	.989

$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.047					.039		.095
.124	.051					.038		.124
.153	.040					.033		.153
.191	.028			.026		.022		.191
.267				.026		.108		.267
.344				.059		.124		.344
.439				.033		.056		.439
.538		.084	.129		.050		.018	.538
.556		.003	.013		-.021		-.045	.556
.592	.067	.017	-.016		-.022	.066	.057	.592
.668	.034	.024	.015			-.045	-.056	.668
.744	.016	.012	-.017		.003		-.052	.744
.821	.003				-.050			.821
.897	.008	.001	-.033		-.010		.015	.897
.945		-.014						.945
.989		-.052	-.042		-.062		-.088	.989

$M = 2.75; \alpha = 0^\circ; \beta = -1^\circ; \delta_e = 0^\circ$

.095	.045					.039		.095
.124	.050					.037		.124
.153	.039			.020		.032		.153
.191	.026			.019		.022		.191
.267				.052		.105		.267
.344				.025		.115		.344
.439					.033	.056		.439
.538		.083	.118		-.012		.016	.538
.556		.004	.006		-.027		-.055	.556
.592	.067	.020	-.022			.063	.054	.592
.668	.035	.026	.012			-.047	-.072	.668
.744	.016	.012	-.022		.006		-.035	.744
.821	.002				-.044			.821
.897	.009	.000	-.034		-.024		.007	.897
.945		-.011						.945
.989		-.059	-.045		-.063		-.080	.989

$M = 2.75; \alpha = 0^\circ; \beta = -2^\circ; \delta_e = 0^\circ$

.095	.045					.038		.095
.124	.047					.037		.124
.153	.040			.016		.032		.153
.191	.024			.015		.023		.191
.267				.047		.104		.267
.344				.018		.105		.344
.439					.018	.041		.439
.538		.081	.108		-.040		-.008	.538
.556		.008	-.001		-.028		-.060	.556
.592	.065	.021	-.027			.077	.027	.592
.668	.032	.026	.008			-.047	-.090	.668
.744	.014	.008	-.032		.007		-.071	.744
.821	.001				-.042			.821
.897	.008	-.001	-.036		-.029		.004	.897
.945		-.010						.945
.989		-.063	-.045		-.060		-.080	.989

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station x_M z_M	C_p at $\theta = \text{---}$						Body station x_M z_M	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 2.75; \alpha = 0^\circ; \beta = -4^\circ; \delta_e = 0^\circ$

.095	.038					.032	.095
.124	.037					.029	.124
.153	.038					.029	.153
.191	.021			.010		.016	.191
.267				.005		.104	.267
.344				.038		.100	.344
.439				.006		.001	.439
.538		.072	.086		-.005		.538
.556		.017	-.015		-.058		.556
.592	.058	.021	-.039		-.048		.592
.668	.029	.024	.011			.036	.668
.744	.012	.000	-.040			-.046	.744
.821	-.002				.006		.821
.897	.004	-.008	-.037		-.046		.897
.945		-.018			-.028	.005	.945
.989		-.076	-.043		-.055	-.085	.989

$M = 2.75; \alpha = 0^\circ; \beta = -6^\circ; \delta_e = 0^\circ$

.095	.026					.020	.095
.124	.028					.017	.124
.153	.020					.015	.153
.191	.010			.000		.002	.191
.267				-.003		.102	.267
.344				.033		.099	.344
.439				-.005		-.015	.439
.538		.090	.057		-.042		.538
.556		.035	-.039		-.085		.556
.592	.043	.023	-.052		-.070		.592
.668	.022	.017	.013			.017	.668
.744	.006	-.008	-.056			-.072	.744
.821	-.008				.002		.821
.897	-.002	-.017	-.039		-.039		.897
.945		-.030			-.020	-.050	.945
.989		-.090	-.048		-.055	-.089	.989

$M = 2.75; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$

.095							.095
.124							.124
.153							.153
.191							.191
.267							.267
.344							.344
.439							.439
.538							.538
.556							.556
.592		.019	-.062		-.079	.034	.592
.668	.014	.002	.013			-.065	.668
.744	.001	-.020	-.060		-.003	-.112	.744
.821	-.012				-.041	-.110	.821
.897	-.008	-.026	-.046		-.027		.897
.945		-.039				-.064	.945
.989		-.103	-.054		-.065	-.084	.989

$M = 2.75; \alpha = 5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$

.095	-.022					.072	.095
.124	-.034					.069	.124
.153	-.043					.064	.153
.191	-.056			-.010		.049	.191
.267				-.002		.108	.267
.344				-.021		.122	.344
.439				-.046		.000	.439
.538		.025	-.039		-.044		.538
.556		-.014	-.072		-.077	-.073	.556
.592	.002	-.023	-.066		-.072	-.067	.592
.668	-.035	-.031	-.038			-.038	.668
.744	-.036	-.056	-.019		-.052	-.042	.744
.821	-.053				.008		.821
.897	-.051	-.067	-.075		.019	-.009	.897
.945		-.074					.945
.989		-.102	-.070		-.052	-.059	.989

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x}{L}$ $\frac{z}{L}$	C_p at $\theta = -$						Body station $\frac{x}{L}$ $\frac{z}{L}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.75; \alpha = 5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	-.014 -.028 -.038 -.052			-.007 -.002 -.017 -.044		.076 .071 .074 .064 .110 .100 -.016	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
		.004 -.019 -.035 -.007 -.041	-.021 -.079 -.061 -.039 .027		-.031 -.074 -.072	-.046 -.086 -.091 -.118 -.057		
	-.003 -.032 -.031 -.049 -.050				-.057 .019 .036			
		-.057 -.073 -.108	-.083 -.070			-.008 -.054		
$M = 2.75; \alpha = 5^\circ; \beta = -4^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	-.004 -.013 -.026 -.045			-.006 .004 -.013 -.040		.081 .082 .088 .080 .115 .096 .020	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
		.002 -.018 -.022 -.015 -.029	.018 -.059 -.057 -.042 .043		-.023 -.063 -.065	.002 -.066 -.006 -.109 -.024		
	.006 -.025 -.045 -.048					-.014 -.026		
		-.048 -.066 -.100	-.084 -.060		.005 .033 -.046	.025		
$M = 2.75; \alpha = 5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.004 -.001 -.021 -.041			-.001 .015 -.002 -.036		.088 .094 .099 .096 .123 .098 .017	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
		-.001 -.036 -.030 -.015 -.033	.029 -.049 -.049 -.054 .028		.020 -.038 -.050	.005 -.047 .038 -.082 .011		
	.015 -.014 -.018 -.037 -.041					-.087 -.023		
		-.048 -.061 -.093	-.079 -.051		.030 .005 -.043	.054		
$M = 2.75; \alpha = 5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.007 .000 -.020 -.040			.001 .017 .004 -.033		.089 .096 .102 .095 .124 .109 .016	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
		-.006 -.043 -.034 -.014 -.032	.036 -.045 -.047 -.057 .040		.042 -.026 -.041	.023 -.042 .071 -.058 .020		
	.017 -.010 -.015 -.035 -.037					.085 -.028		
		-.054 -.063 -.093	-.078 -.053		.032 .018 -.044	.049		

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{z_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{z_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 2.75; \alpha = 5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.008					.088	.095
.124	.001					.096	.124
.153	-.019					.103	.153
.191	-.040			.005		.093	.191
.267				.020		.125	.267
.344				.013		.120	.344
.439				-.026		.015	.439
.538		-.008	.046		.059		.538
.556		-.052	-.041		-.016	-.037	.556
.592	.017	-.038	-.044		-.034	.062	.592
.668	-.007	-.015	-.060		-.023	-.037	.668
.744	-.014	-.029	.055		-.055	-.005	.744
.821	-.034				.021		.821
.897	-.036	-.054	-.076		.037	.069	.897
.945		-.064					.945
.989		-.091	-.055		-.044	-.027	.989

$M = 2.75; \alpha = 5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$

.095							.095
.124							.124
.153							.153
.191							.191
.267							.267
.344							.344
.439							.439
.538							.538
.556							.556
.592							.592
.668							.668
.744							.744
.821							.821
.897							.897
.945		-.060	-.074		.052	.117	.945
.989		-.084	-.053		-.042	-.039	.989

$M = 2.75; \alpha = 5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$

.095	.004					.087	.095
.124	-.002					.094	.124
.153	-.021					.095	.153
.191	-.041			.016		.093	.191
.267				.026		.120	.267
.344				.030		.111	.344
.439				-.008		.018	.439
.538		-.014	.069		.098		.538
.556		-.065	-.030		.013	.058	.556
.592	.015	-.056	-.038		-.013	-.016	.592
.668	-.013	-.021	-.063		-.091	.072	.668
.744	-.018	-.034	.080		-.021	-.001	.744
.821	-.037					.026	.821
.897	-.041	-.051	-.073			.115	.897
.945		-.057					.945
.989		-.080	-.052		-.041	-.066	.989

$M = 2.75; \alpha = 5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$

.095	-.003					.081	.095
.124	-.011					.081	.124
.153	-.025					.088	.153
.191	-.044			.031		.079	.191
.267				.027		.114	.267
.344				.047		.111	.344
.439				.012		.008	.439
.538		-.015	.095		.136		.538
.556		-.073	-.015		.041	.073	.556
.592	.009	-.070	-.029		.009	.001	.592
.668	-.021	-.031	-.062		-.030	.095	.668
.744	-.023	-.040	.093		-.034	.025	.744
.821	-.042				-.067	.078	.821
.897	-.045	-.050	-.071		.047		.897
.945		-.059			.089	.153	.945
.989		-.070	-.050		-.043	-.075	.989

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{l_M}$	C_p at $\theta = -$						Body station $\frac{x_M}{l_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.75; \alpha = 5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	-.014 -.026 -.035 -.050 .002 -.028 -.031 -.047 -.048 	 -.004 -.071 -.079 -.063 -.051 -.056 -.064 -.052	 .130 .007 -.013 -.053 .118 -.071 -.051	 .052 .036 .067 .037 <				

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x}{M}$ $\frac{z}{M}$	C_p at $\theta =$ —						Body station $\frac{x}{M}$ $\frac{z}{M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 2.75; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.095	.084					.011		.095
.124	.140					-.005		.124
.153	.165					-.020		.153
.191	.123					-.026		.191
.267				.013		.059		.267
.344				.025		.138		.344
.439				.069		.128		.439
.538				.039				.538
.556		.074	.150		.080		.039	.556
.592		.016	.015		-.005		-.039	.592
.668	.084	.039	-.005		-.018	.083	.097	.668
.744	.054	.047	.035			-.036	-.056	.744
.821	.035	.035	-.040		.052		-.069	.821
.897	.025				-.069			.897
.945	.042	.035	.007		-.036		-.027	.945
.989		.021						.989
		-.024	-.031		-.075		-.113	

$M = 2.75; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.095	.073					.023		.095
.124	.103					.011		.124
.153	.127					-.002		.153
.191	.085					-.010		.191
.267				.013		.101		.267
.344				.025		.128		.344
.439				.072		.116		.439
.538				.034				.538
.556		.079	.152		.066		.033	.556
.592		.004	.026		-.010		-.041	.592
.668	.071	.014	-.017		-.014	.076	.088	.668
.744	.038	.031	.016			-.038	-.057	.744
.821	.022	.019	-.027		.026		-.058	.821
.897	.010				-.062			.897
.945	.021	.014	-.013		-.021		-.005	.945
.989		-.001						.989
		-.040	-.033		-.068		-.101	

$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.095	.056					.039		.095
.124	.072					.030		.124
.153	.089					.019		.153
.191	.052					.010		.191
.267				.016		.126		.267
.344				.027		.122		.344
.439				.057		.074		.439
.538				.023				.538
.556		.061	.130		.062		.028	.556
.592		-.007	.014		-.014		-.038	.592
.668	.058	-.007	-.014		-.017	.073	.073	.668
.744	.025	.016	.004			-.036	-.054	.744
.821	.011	.005	-.005		.000		-.049	.821
.897	-.001				-.047			.897
.945	.003	-.005	-.031		-.005		.018	.945
.989		-.018						.989
		-.055	-.038		-.059		-.079	

$M = 2.75; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.095	.038					.056		.095
.124	.044					.051		.124
.153	.058					.042		.153
.191	.022					.032		.191
.267				.011		.117		.267
.344				.023		.109		.344
.439				.043		.020		.439
.538				.005				.538
.556		.034	.100		.064		.025	.556
.592		-.025	-.006		-.012		-.032	.592
.668	.042	-.018	-.023		-.021	.073	.055	.668
.744	.011	.001	-.020			-.035	-.048	.744
.821	-.001	-.008	.019		-.022		-.034	.821
.897	-.014				-.024			.897
.945	-.013	-.023	-.048		.015		.042	.945
.989		-.030						.989
		-.070	-.043		-.052		-.059	

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{L_M}$		C_p at $\theta =$ —						Body station $\frac{x_M}{L_M}$	
		0°		60°	90°	120°	180°		
		Body center line	Nacelle center line				Body center line	Nacelle center line	
$M = 2.75; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$									
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.021 .035 .029 -.001 .025 .001 -.009 -.024 -.026 								

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{z_M}$	C_p at $\theta = \text{---}$						Body station $\frac{x_M}{z_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.050					.019	.095
.124	.059					.015	.124
.153	.059					.012	.153
.191	.051					.004	.191
.267				.016		-.005	.267
.344				.016		.211	.344
.439				.026		.056	.439
.538				.051			.538
.556		.099	.163		.077		.556
.592	.075	.020	.031		.014	.073	.592
.668	.041	.035	.003		.004	-.022	.668
.744	.030	.033	.028		.004	.005	.744
.821	.017	.026	-.002		.026	-.039	.821
.897	.024				-.037	-.047	.897
.945		.019	-.003		-.016	.001	.945
.989		.005	-.021		-.051	-.080	.989
		-.021					

$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.038					.032	.095
.124	.040					.030	.124
.153	.038					.030	.153
.191	.023					.020	.191
.267				.016		.029	.267
.344				.016		.194	.344
.439				.041		.056	.439
.538				.048			.538
.556		.096	.133		.071		.556
.592	.063	.004	.022		.008	.069	.592
.668	.030	.013	-.007		-.003	-.018	.668
.744	.019	.018	.007		.094	.000	.744
.821	.007	.011	.012		.001	-.039	.821
.897	.011				.003	-.043	.897
.945		.002	-.016		-.025	.013	.945
.989		-.008	-.024		.006	-.068	.989
		-.031			-.052		

$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.025					.047	.095
.124	.019					.047	.124
.153	.017					.049	.153
.191	.004					.039	.191
.267				.013		.062	.267
.344				.014		.119	.344
.439				.041		.045	.439
.538				.032			.538
.556		.060	.099		.071		.556
.592	.047	-.008	.002		.006	.064	.592
.668	.018	-.012	-.014		-.007	-.013	.668
.744	.011	.006	-.014		.094	-.003	.744
.821	-.003	.006	-.014		.002	-.032	.821
.897	-.003	-.001	.030			-.029	.897
.945		-.012	-.030		-.019		.945
.989		-.022	-.031		-.006	.032	.989
		-.044			.034	-.051	
					-.045		

$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.012					.066	.095
.124	.009					.067	.124
.153	.004					.070	.153
.191	-.015					.062	.191
.267				.004		.123	.267
.344				.010		.111	.344
.439				.026		.053	.439
.538				.010			.538
.556		.021	.074		.075		.556
.592	.030	-.033	-.015		.007	.074	.592
.668	.006	-.031	-.025		-.011	-.006	.668
.744	.000	-.004	-.034		.097	.004	.744
.821	-.014	-.012	.055		.007	-.015	.821
.897	-.015					-.012	.897
.945		-.027	-.040			.050	.945
.989		-.034	-.038			-.032	.989
		-.052			-.040		

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station		C_p at $\theta =$ —						Body station	
$\frac{x_M}{z_M}$		0°		60°	90°	120°	180°		$\frac{x_M}{z_M}$
z_M		Body center line	Nacelle center line				Body center line	Nacelle center line	
$M = 3.22; \alpha = -5^0; \beta = -8^0; \delta_e = 0^0$									
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.068 .072 .059 .046								.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 3.22; \alpha = -5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$

.095	.077					.002		.095
.124	.096					.002		.124
.153	.098					-.003		.153
.191	.092			.006		-.016		.191
.267				.008		-.015		.267
.344				.011		.203		.344
.439				.039		.025		.439
.538		.079	.162		.085		.061	.538
.556		.037	.027		.012		-.017	.556
.592	.097	.039	-.008		-.000	.128	.039	.592
.668	.063	.057	.059			.010	-.049	.668
.744	.050	.046	-.017		.057		-.080	.744
.821	.041				-.052			.821
.897	.054	.046	.019		-.041		-.028	.897
.945		.040			-.060			.945
.989		-.011	-.025				-.076	.989

$M = 3.22; \alpha = -5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.078					.004		.095
.124	.096					.002		.124
.153	.100					-.003		.153
.191	.094			.012		-.016		.191
.267				.012		-.015		.267
.344				.010		.213		.344
.439				.049		.038		.439
.538		.086	.172		.101		.084	.538
.556		.042	.032		.019		-.016	.556
.592	.096	.065	-.002		.006	.128	.023	.592
.668	.062	.054	.064			.010	-.035	.668
.744	.049	.048	-.014		.060		-.060	.744
.821	.040				-.049			.821
.897	.053	.048	.020		-.035		-.022	.897
.945		.035			-.053			.945
.989		-.002	-.019				-.084	.989

$M = 3.22; \alpha = -5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$

.095	.077					.002		.095
.124	.096					.001		.124
.153	.098					-.004		.153
.191	.092			.018		-.016		.191
.267				.016		-.016		.267
.344				.010		.211		.344
.439				.055		.038		.439
.538		.094	.185		.111		.103	.538
.556		.049	.041		.026		-.009	.556
.592	.096	.050	.007		.015	.127	.023	.592
.668	.061	.054	.071			.012	-.021	.668
.744	.047	.050	-.008		.063		-.040	.744
.821	.038				-.042			.821
.897	.051	.049	.021		-.020		-.016	.897
.945		.032						.945
.989		.006	-.015		-.046		-.086	.989

$M = 3.22; \alpha = -5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$

.095	.078					.001		.095
.124	.094					-.001		.124
.153	.097					-.005		.153
.191	.091			.025		-.018		.191
.267				.021		-.017		.267
.344				.015		.206		.344
.439				.064		.042		.439
.538		.112	.214		.114		.112	.538
.556		.060	.056		.032		.001	.556
.592	.093	.056	.016		.023	.132	.040	.592
.668	.056	.049	.078			.014	-.002	.668
.744	.045	.051	-.001		.064		-.016	.744
.821	.036				-.034			.821
.897	.047	.047	.024		-.010		-.009	.897
.945		.032						.945
.989		.015	-.008		-.040		-.086	.989

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{l_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{l_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 3.22; \alpha = -5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.075 .083 .093 .081			.041 .032 .025 .082		.130 .035 .045 .070 -.019 .012 -.032	-.004 -.009 -.011 .024 .027 .217 .054 .130 .016 .040 .002 .029 .020 -.083	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
$M = 3.22; \alpha = -5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.071 .070 .078 .067 .055 .035 .034 .031 .042 			.062 .048 .039 .106 .193 .116 .090 .038 .032 .038 .029 .031	.307 .111 .060 .112 .029 .030 .004	.152 .054 .066 .082 -.001 .040 -.021	-.012 -.018 -.021 -.033 -.037 .218 .057 .155 .032 .060 .022 .066 .077 -.078	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
$M = 3.22; \alpha = -5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.068 .068 .060 .048 .023 .024 .030 .029 .042 			.090 .068 .056 .138 .244 .143 .107 .044 .027 .031 .027 .039	.352 .145 .088 .133 .053 .041 .003	.174 .076 .090 .097 .021 .071 -.008	-.024 -.031 -.035 -.049 -.053 .203 .070 .177 .051 .139 .069 .101 .136 -.073	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
$M = 3.22; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.017 .014 .006 -.004 .018 .015 .009 -.002 .002 			.093 .080 .068 .117 .156 .031 .028 -.006 -.024 -.010 -.016 -.001	.281 .115 .071 .059 .073 -.002 -.002 -.022	.196 .089 .085 .034 .037 .106 -.017	.003 -.003 -.007 -.020 -.028 .177 .056 .170 .085 .029 .116 .168 -.066	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x}{L}$ $\frac{z}{L}$	C_p at $\theta =$ —						Body station $\frac{x}{L}$ $\frac{z}{L}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.022					.014	.095
.124	.017					.007	.124
.153	.015					.006	.153
.191	.005			.068		.006	.191
.267				.064		.014	.267
.344				.051		.186	.344
.439				.096		.044	.439
.538		.134	.242		.151		.538
.556		.021	.089		.062	.139	.556
.592	.035	.031	.048		.061	.058	.592
.668	.021	-.010	.047		.117	-.002	.668
.744	.015	-.018	.053		.005	.086	.744
.821	.003			.022			.821
.897	.006	.001	-.011	.018			.897
.945		-.012		.079		.128	.945
.989		-.007	-.022		-.028	-.070	.989

$M = 3.22; \alpha = 0^\circ; \beta = 4^\circ; \delta_e = 0^\circ$

.095	.031					.022	.095
.124	.026					.018	.124
.153	.030					.016	.153
.191	.016			.047		.006	.191
.267				.046		-.002	.267
.344				.036		.177	.344
.439				.075		.030	.439
.538		.126	.202		.109		.538
.556		.015	.063		.039	.109	.556
.592	.056	.021	.026		.036	.032	.592
.668	.025	-.006	.031		.090	-.024	.668
.744	.019	.006	.029		.010	.047	.744
.821	.007			.011			.821
.897	.010	.005	-.014	.000			.897
.945		-.009		.049		.074	.945
.989		-.014	-.019		-.037	-.074	.989

$M = 3.22; \alpha = 0^\circ; \beta = 2^\circ; \delta_e = 0^\circ$

.095	.036					.028	.095
.124	.036					.025	.124
.153	.033					.022	.153
.191	.025			.031		.014	.191
.267				.029		.003	.267
.344				.033		.192	.344
.439				.059		.033	.439
.538		.115	.168		.080		.538
.556		.007	.044		.023	.085	.556
.592	.065	.019	.009		.017	-.003	.592
.668	.029	.008	.021		.100	.012	.668
.744	.020	.013	.016		.003	-.027	.744
.821	.009			.008		.003	.821
.897	.015	.008	-.016	-.012			.897
.945		-.007		.024		.024	.945
.989		-.021	-.018		-.045	-.080	.989

$M = 3.22; \alpha = 0^\circ; \beta = 1^\circ; \delta_e = 0^\circ$

.095	.038					.031	.095
.124	.040					.028	.124
.153	.036					.025	.153
.191	.026			.034		.016	.191
.267				.022		.006	.267
.344				.032		.194	.344
.439				.052		.034	.439
.538		.108	.153		.071		.538
.556		.007	.034		.013	.077	.556
.592	.069	.021	.002		.007	-.013	.592
.668	.034	.019	.018		.094	.005	.668
.744	.022	.015	.009		.002	-.031	.744
.821	.011			.009		-.022	.821
.897	.016	.008	-.017	-.021			.897
.945		-.005		.009		.027	.945
.989		-.025	-.020		-.048	-.079	.989

TABLE I.- PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x}{L}$	C_p at $\theta =$ —						Body station $\frac{x}{L}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.039					.031	.095
.124	.041					.029	.124
.153	.038					.026	.153
.191	.025			.016		.017	.191
.267				.017		.029	.267
.344				.031		.193	.344
.439				.045		.056	.439
.538		.099	.136		.061		.538
.556		.007	.025		.003		.556
.592	.069	.022	-.005		-.004	.090	.592
.668	.035	.026	.013			-.003	.668
.744	.023	.016	.004		.009		.744
.821	.011				-.029		.821
.897	.017	.009	-.018		-.000		.897
.945		-.005				.015	.945
.989		-.032	-.024		-.052	-.071	.989

$M = 3.22; \alpha = 0^\circ; \beta = -1^\circ; \delta_e = 0^\circ$

.095	.039					.031	.095
.124	.040					.029	.124
.153	.037					.026	.153
.191	.026			.013		.017	.191
.267				.014		.014	.267
.344				.035		.187	.344
.439				.039		.051	.439
.538		.089	.123		.052		.538
.556		.010	.018		-.006		.556
.592	.069	.023	-.012		-.013	.100	.592
.668	.035	.029	.006			-.003	.668
.744	.021	.015	-.001		.009		.744
.821	.010				-.037		.821
.897	.016	.008	-.020		-.014		.897
.945		-.002				.012	.945
.989		-.038	-.029		-.049	-.067	.989

$M = 3.22; \alpha = 0^\circ; \beta = -2^\circ; \delta_e = 0^\circ$

.095	.037					.030	.095
.124	.038					.027	.124
.153	.034					.024	.153
.191	.027			.009		.016	.191
.267				.011		.010	.267
.344				.034		.180	.344
.439				.032		.037	.439
.538		.081	.109		.039		.538
.556		.014	.010		-.016		.556
.592	.068	.024	-.015		-.018	.087	.592
.668	.032	.029	.007			.002	.668
.744	.021	.013	-.009		.011		.744
.821	.010				-.042		.821
.897	.016	.008	-.021		-.022		.897
.945		-.002				.016	.945
.989		-.040	-.030		-.045	-.064	.989

$M = 3.22; \alpha = 0^\circ; \beta = -4^\circ; \delta_e = 0^\circ$

.095	.033					.025	.095
.124	.031					.021	.124
.153	.034					.018	.153
.191	.020			.004		.011	.191
.267				.004		.001	.267
.344				.031		.184	.344
.439				.021		.021	.439
.538		.071	.088		.014		.538
.556		.017	-.004		-.034		.556
.592	.059	.024	-.024		-.036	.073	.592
.668	.026	.028	.010			.010	.668
.744	.019	.008	-.023		.008		.744
.821	.007				-.043		.821
.897	.011	.000	-.021		-.015		.897
.945		-.010				-.009	.945
.989		-.048	-.031		-.043	-.063	.989

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{L_M}$ $\frac{z_M}{L_M}$	C_p at $\theta = \text{---}$						Body station $\frac{x_M}{L_M}$ $\frac{z_M}{L_M}$
	0°		60°	90°	120°	180°	
	Body center line	Nacelle center line				Body center line	Nacelle center line

$M = 3.22; \alpha = 0^\circ; \beta = -6^\circ; \delta_e = 0^\circ$

.095	.025					.017	.095
.124	.018					.012	.124
.153	.020					.011	.153
.191	.009			-.002		.001	.191
.267				-.002		-.009	.267
.344				.028		.185	.344
.439				.014		.031	.439
.538		.062	.064		-.018		.538
.556		.024	-.016		-.055	-.015	.556
.592	.039	.025	-.036		-.051	-.063	.592
.668	.023	.025	.012			-.061	.668
.744	.015	.002	-.044			-.081	.744
.821	.003				.005	-.090	.821
.897	.007	-.007	-.022		-.043		.897
.945		-.018			-.017	-.031	.945
.989		-.057	-.035		-.043	-.063	.989

$M = 3.22; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$

.095	.017					.008	.095
.124	.013					.003	.124
.153	.007					.001	.153
.191	-.002			-.006		-.010	.191
.267				-.007		-.020	.267
.344				.024		.172	.344
.439				.006		.039	.439
.538		.072	.036		-.032		.538
.556		.025	-.031		-.062	-.039	.556
.592	.020	.021	-.041		-.058	-.072	.592
.668	.017	.010	.008			.115	.668
.744	.009	-.007	-.045			.006	.744
.821	-.003				.002	-.085	.821
.897	.002	-.015	-.024		-.038	-.093	.897
.945		-.024			-.018	-.048	.945
.989		-.062	-.039		-.045	-.068	.989

$M = 3.22; \alpha = 5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$

.095	-.014					.063	.095
.124	-.024					.058	.124
.153	-.026					.058	.153
.191	-.040			-.013		.043	.191
.267				-.016		.116	.267
.344				-.014		.101	.344
.439				-.031		.055	.439
.538		-.010	-.015		-.029		.538
.556		-.018	-.054		-.056	-.031	.556
.592	-.006	-.019	-.052		-.062	-.068	.592
.668	-.032	-.012	-.031			.119	.668
.744	-.020	-.039	.001		-.039	.015	.744
.821	-.033				-.004	-.081	.821
.897	-.033	-.047	-.043		.014	-.019	.897
.945		-.051					.945
.989		-.065	-.047		-.045	-.040	.989

$M = 3.22; \alpha = 5^\circ; \beta = -6^\circ; \delta_e = 0^\circ$

.095	-.012					.070	.095
.124	-.022					.067	.124
.153	-.029					.063	.153
.191	-.040			-.012		.056	.191
.267				-.016		.122	.267
.344				-.016		.111	.344
.439				-.031		.042	.439
.538		-.004	.001		-.026		.538
.556		-.017	-.061		-.056	-.005	.556
.592	.005	-.026	-.052		-.060	-.063	.592
.668	-.019	.005	-.035			.099	.668
.744	-.019	-.029	-.022		-.045	.004	.744
.821	-.034				-.005	-.082	.821
.897	-.034	-.040	-.050		.033	.005	.897
.945		-.051					.945
.989		-.066	-.048		-.035	-.039	.989

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —							Body station $\frac{x_M}{L_M}$
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line	Nacelle center line	
$M = 3.22; \alpha = 5^\circ; \beta = -4^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	-.004 -.013 -.019 -.034				-.010 -.012 -.014 -.028	.073 .073 .070 .072 .118 .112 .067		.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
		-.001 -.020 -.019 -.013 -.020 -.032 -.046 -.065	.011 -.048 -.046 -.042 .036		.002 -.044 -.054		.033 -.041 -.021 -.067 -.081 -.029 -.031	
$M = 3.22; \alpha = 5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.001 -.002 -.011 -.030				-.008 -.006 -.002 -.021	.078 .078 .081 .076 .121 .113 .070		.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
		-.002 -.034 -.026 -.008 -.022 -.037 -.045 -.062	.029 -.041 -.041 -.048 .033		.040 -.020 -.038	.106 .018	.044 -.027 .035 -.050 -.074 .051 -.025	
$M = 3.22; \alpha = 5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.003 .001 -.009 -.029				-.007 -.001 .006 -.013	.080 .080 .082 .078 .122 .110 .064		.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
		-.002 -.044 -.036 -.009 -.020 -.040 -.047 -.063	.042 -.034 -.039 -.049 .050		.061 -.004 -.026	.104 .010	.074 -.013 .009 -.029 -.048 .051 -.018	
$M = 3.22; \alpha = 5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.004 .001 -.009 -.029				-.005 .002 .012 -.006	.078 .079 .081 .076 .124 .109 .061		.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
		-.001 -.049 -.043 -.012 -.019 -.037 -.045 -.060	.056 -.028 -.036 -.050 .066		.077 .007 -.016	.101 .009	.075 -.005 .008 -.008 -.013 .059 -.018	

TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 3.22; \alpha = 5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$								
.095	.003					.079	.095	
.124	.000					.079	.124	
.153	-.009					.082	.153	
.191	-.030			.000		.076	.191	
.267				.006		.122	.267	
.344				.018		.108	.344	
.439				-.001		.064	.439	
.538		-.001	.067		.092		.538	
.556		-.051	-.022		.018	.088	.556	
.592	.019	-.047	-.032		-.006	.102	.592	
.668	-.004	-.014	-.051			.015	.668	
.744	-.008	-.020	.079		-.052	.007	.744	
.821	-.022				.039		.821	
.897	-.025	-.037	-.050		.077	.072	.897	
.945		-.045					.945	
.989		-.058	-.046		-.039	-.034	.989	
$M = 3.22; \alpha = 5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$								
.095	.001					.078	.095	
.124	-.002					.078	.124	
.153	-.010					.078	.153	
.191	-.030			.006		.073	.191	
.267				.008		.120	.267	
.344				.023		.104	.344	
.439				.007		.069	.439	
.538		.002	.082		.107		.538	
.556		-.052	-.013		.029	.100	.556	
.592	.018	-.053	-.027		.006	.032	.592	
.668	-.008	-.020	-.050			.017	.668	
.744	-.011	-.022	.084		-.054	.030	.744	
.821	-.024				.044		.821	
.897	-.027	-.037	-.049		.084	.072	.897	
.945		-.044					.945	
.989		-.056	-.045		-.040	-.038	.989	
$M = 3.22; \alpha = 5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$								
.095	-.002					.075	.095	
.124	-.009					.075	.124	
.153	-.015					.071	.153	
.191	-.032			.018		.068	.191	
.267				.016		.121	.267	
.344				.036		.098	.344	
.439				.022		.050	.439	
.538		.012	.112		.133		.538	
.556		-.050	.003		.051	.128	.556	
.592	.015	-.057	-.017		.025	.050	.592	
.668	-.013	-.038	-.045			.013	.668	
.744	-.014	-.029	.093		-.053	.070	.744	
.821	-.028				.053		.821	
.897	-.031	-.038	-.048		.100	.123	.897	
.945		-.045					.945	
.989		-.053	-.042		-.044	-.039	.989	
$M = 3.22; \alpha = 5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$								
.095	-.009					.070	.095	
.124	-.017					.065	.124	
.153	-.022					.061	.153	
.191	-.038			.038		.052	.191	
.267				.032		.120	.267	
.344				.055		.101	.344	
.439				.045		.053	.439	
.538		.028	.151		.169		.538	
.556		-.042	.027		.074	.152	.556	
.592	.011	-.053	.002		.052	.079	.592	
.668	-.017	-.053	-.032			.012	.668	
.744	-.021	-.045	.115		-.045	.112	.744	
.821	-.034				.070		.821	
.897	-.032	-.043	-.048		.128	.168	.897	
.945		-.048					.945	
.989		-.047	-.045		-.042	-.037	.989	

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TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 3.22; \alpha = 5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$								
.095	-.018					.065		.095
.124	-.028					.057		.124
.153	-.031					.057		.153
.191	-.046				.056	.042		.191
.267					.051	.118		.267
.344					.076	.092		.344
.439					.070	.057		.439
.538		.046	.192				.189	.538
.556		-.033	.052		.213	.102	.079	.556
.592	-.009	-.045	.021		.102	.080	.110	.592
.668	-.035	-.061	-.018				.059	.668
.744	-.026	-.056	.142		-.034		.159	.744
.821	-.036				.094			.821
.897	-.035	-.053	-.045		.163		.210	.897
.945		-.054						.945
.989		-.038	-.045		-.031		-.030	.989
$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095	.118					-.002		.095
.124	.182					-.015		.124
.153	.212					-.025		.153
.191	.191				.014	-.039		.191
.267					.007	-.019		.267
.344					.003	.221		.344
.439					.042	.032		.439
.538		.151	.167				.108	.538
.556		.096	.023		.072		-.002	.556
.592	.109	.079	.002		.005		.089	.592
.668	.084	.086	.085		.013	.134	.024	.668
.744	.075	.074	-.009			.092	-.025	.744
.821	.067					-.042		.821
.897	.088	.084	-.004		-.042	-.021	-.024	.897
.945		.065						.945
.989		.024	-.013		-.055		-.082	.989
$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095	.097					.000		.095
.124	.145					-.010		.124
.153	.172					-.022		.153
.191	.151				.014	-.034		.191
.267					.016	-.024		.267
.344					.008	.211		.344
.439					.048	.013		.439
.538		.089	.125				.095	.538
.556		.050	.025		.108		-.006	.556
.592	.097	.049	-.002		.023		.058	.592
.668	.068	.063	.072		.005	.134	.018	.668
.744	.057	.055	-.012			.074	-.023	.744
.821	.047					-.045		.821
.897	.065	.059	.036		-.045	-.026	-.024	.897
.945		.046						.945
.989		.010	-.014		-.055		-.084	.989
$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095	.073					.005		.095
.124	.110					-.003		.124
.153	.130					-.011		.153
.191	.111				.013	-.022		.191
.267					.023	-.022		.267
.344					.015	.209		.344
.439					.043	.010		.439
.538		.079	.132				.085	.538
.556		.020	.029		.094		-.013	.556
.592	.080	.045	.303		.025		.025	.592
.668	.051	.043	.048		.006	.129	-.025	.668
.744	.040	.037	-.007			.051	-.044	.744
.821	.029				.051			.821
.897	.040	.035	.017		-.043	-.021	-.014	.897
.945		.024			-.021			.945
.989		-.007	-.017		-.052		-.085	.989

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TABLE I. - PRESSURE COEFFICIENTS ON MISSILE BODY MEASURED
IN PRESENCE OF BOOSTER - Concluded

Body station $\frac{x_M}{L_M}$ $\frac{z_M}{L_M}$	C_p at $\theta = -$						Body station $\frac{x_M}{L_M}$ $\frac{z_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.058 .078 .098 .080 .069 .037 .028 .017 .024 	 .092 .006 .012 .027 .021 .016 .005 -.020 	 .159 .035 -.004 .016 .001 .000 -.020 	 .010 .026 .030 .046 	 .084 .020 .007 .027 -.034 -.011 -.052 	 .112 .007 <		

TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER

Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
$M = 2.29; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.284	-.078	.048	.330	-.040	.069	.399	.000	.100	.419	.024
.200	.173	-.097	.200	.244	.000	.200	.322	-.037	.200	.362	.004
.450	.133	-.100	.450	.148	-.122	.450	.208	-.094	.450	.252	-.060
.750	.077	-.150	.750	.071	-.145	.750	.106	-.138	.750	.148	-.105
.963	.060	-.068	.953	.071	-.145	.931	.092	-.138	.901	.116	-.111
$M = 2.29; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.205	-.044	.048	.248	-.018	.069	.323	.030	.100	.342	.049
.200	.123	-.076	.200	.191	.000	.200	.249	-.010	.200	.295	.034
.450	.093	-.095	.450	.104	-.108	.450	.158	-.075	.450	.192	-.039
.750	.042	-.148	.750	.040	-.141	.750	.066	-.127	.750	.102	-.091
.963	.032	-.033	.953	.040	-.141	.931	.046	-.132	.901	.075	-.102
$M = 2.29; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.130	-.018	.048	.192	.020	.069	.264	.072	.100	.265	.051
.200	.083	-.052	.200	.137	.000	.200	.196	.028	.200	.240	.065
.450	.059	-.079	.450	.062	-.086	.450	.106	-.049	.450	.145	-.003
.750	.017	-.135	.750	.009	-.123	.750	.030	-.107	.750	.060	-.065
.963	.002	.003	.953	.009	-.123	.931	.010	-.118	.901	.036	-.082
$M = 2.29; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.076	.016	.048	.128	.068	.069	.199	.127	.100	.156	.068
.200	.049	-.026	.200	.074	.000	.200	.141	.073	.200	.170	.088
.450	.022	-.049	.450	.028	-.057	.450	.053	-.015	.450	.099	.037
.750	-.014	-.113	.750	-.023	-.104	.750	-.007	-.083	.750	.017	-.034
.963	-.023	.052	.953	-.023	-.104	.931	-.022	-.098	.901	-.006	-.054
$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.075	-.015	.048	.127	.067	.069	.198	.126	.100	.155	.068
.200	.049	-.025	.200	.073	.000	.200	.141	.073	.200	.170	.087
.450	.022	-.050	.450	.027	-.058	.450	.052	-.015	.450	.098	.037
.750	-.015	-.113	.750	-.023	-.105	.750	-.008	-.083	.750	.017	-.034
.963	-.023	.052	.953	-.023	-.105	.931	-.023	-.097	.901	-.006	-.054
$M = 2.29; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.010	.116	.048	.026	.167	.069	.086	.225	.100	.028	.141
.200	-.025	.053	.200	-.022	.032	.200	.045	.164	.200	.044	.172
.450	-.042	.012	.450	-.051	.032	.450	-.028	.058	.450	.012	.114
.750	-.066	-.059	.750	-.077	-.050	.750	-.083	-.019	.750	-.050	.030
.963	-.071	.100	.953	-.077	-.050	.931	-.092	-.018	.901	-.068	.002
$M = 2.29; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.053	.173	.048	-.016	.230	.069	.030	.297	.100	-.018	.204
.200	-.069	.101	.200	-.063	.230	.200	.000	.220	.200	-.015	.232
.450	-.072	.046	.450	-.092	.081	.450	-.060	.092	.450	-.031	.150
.750	-.088	-.026	.750	-.100	-.015	.750	-.112	.037	.750	-.079	.062
.963	-.090	.121	.953	-.100	-.015	.931	-.121	.024	.901	-.093	.036
$M = 2.29; \alpha = -5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$											
.037	.158	-.130	.048	.165	-.091	.069	.235	-.070	.100	.262	-.039
.200	.107	-.087	.200	.133	-.096	.200	.186	-.100	.200	.208	-.061
.450	.063	-.134	.450	.084	-.096	.450	.116	-.136	.450	.136	-.115
.750	.016	-.134	.750	.015	-.144	.750	.050	-.145	.750	.064	-.148
.963	.010	-.094	.953	.015	-.144	.931	.032	-.136	.901	.046	-.156

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TABLE II.- PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

M = 2.29; $\alpha = -5^\circ$; $\beta = -0^\circ$; $\delta_e = 0^\circ$

.037	.147	-.105	.048	.176	-.073	.069	.258	-.050	.100	.277	-.014
.200	.101	-.087	.200	.136	-.087	.200	.196	-.078	.200	.226	-.037
.450	.086	-.120	.450	.091	-.108	.450	.115	-.125	.450	.142	-.097
.750	.020	-.125	.750			.750	.052	-.155	.750	.062	-.136
.963	.014	-.079	.953	.016	-.169	.931	.035	-.137	.901	.043	-.146

M = 2.29; $\alpha = -5^\circ$; $\beta = -4^\circ$; $\delta_e = 0^\circ$

.037	.149	-.072	.048	.198	-.055	.069	.274	-.017	.100	.289	.015
.200	.098	-.081	.200	.148	-.055	.200	.208	-.050	.200	.241	-.010
.450	.072	-.108	.450	.094	-.107	.450	.122	-.105	.450	.151	-.074
.750	.024	-.146	.750			.750	.048	-.147	.750	.070	-.117
.963	.015	-.058	.953	.019	-.157	.931	.039	-.145	.901	.043	-.129

M = 2.29; $\alpha = -5^\circ$; $\beta = -2^\circ$; $\delta_e = 0^\circ$

.037	.155	-.049	.048	.212	-.032	.069	.284	.017	.100	.301	.046
.200	.094	-.071	.200	.158	-.032	.200	.215	-.020	.200	.255	.023
.450	.077	-.102	.450	.085	-.098	.450	.127	-.085	.450	.158	-.049
.750	.026	-.154	.750			.750	.043	-.132	.750	.075	-.096
.963	.016	-.019	.953	.021	-.146	.931	.034	-.136	.901	.049	-.109

M = 2.29; $\alpha = -5^\circ$; $\beta = -1^\circ$; $\delta_e = 0^\circ$

.037	.157	-.040	.048	.215	-.015	.069	.285	.035	.100	.303	.058
.200	.096	-.068	.200	.159	-.015	.200	.215	-.004	.200	.261	.039
.450	.075	-.092	.450	.082	-.098	.450	.129	-.074	.450	.163	-.034
.750	.029	-.151	.750			.750	.042	-.124	.750	.076	-.087
.963	.013	-.013	.953	.022	-.139	.931	.026	-.132	.901	.051	-.102

M = 2.29; $\alpha = -5^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$

.037	.166	-.036	.048	.220	.002	.069	.294	.033	.100	.304	.045
.200	.102	-.054	.200	.165	.002	.200	.223	.010	.200	.267	.051
.450	.078	-.086	.450	.083	-.093	.450	.134	-.062	.450	.168	-.018
.750	.028	-.140	.750			.750	.048	-.115	.750	.080	-.075
.963	.016	-.017	.953	.022	-.133	.931	.027	-.123	.901	.057	-.087

M = 2.29; $\alpha = -5^\circ$; $\beta = 1^\circ$; $\delta_e = 0^\circ$

.037	.175	-.028	.048	.220	.019	.069	.301	.071	.100	.298	.033
.200	.105	-.060	.200	.168	.019	.200	.228	.026	.200	.272	.052
.450	.075	-.079	.450	.084	-.090	.450	.135	-.051	.450	.173	-.006
.750	.027	-.134	.750			.750	.050	-.108	.750	.084	-.066
.963	.016	-.020	.953	.022	-.126	.931	.028	-.117	.901	.058	-.081

M = 2.29; $\alpha = -5^\circ$; $\beta = 2^\circ$; $\delta_e = 0^\circ$

.037	.186	-.017	.048	.229	.033	.069	.312	.083	.100	.286	.022
.200	.113	-.055	.200	.173	.033	.200	.236	.038	.200	.279	.045
.450	.074	-.073	.450	.087	-.082	.450	.140	-.041	.450	.180	.006
.750	.030	-.126	.750			.750	.054	-.100	.750	.088	-.057
.963	.018	-.017	.953	.023	-.120	.931	.032	-.112	.901	.062	-.074

M = 2.29; $\alpha = -5^\circ$; $\beta = 4^\circ$; $\delta_e = 0^\circ$

.037	.201	.012	.048	.240	.071	.069	.333	.124	.100	.256	.044
.200	.125	-.016	.200	.179	.071	.200	.255	.018	.200	.278	.044
.450	.074	-.068	.450	.096	-.065	.450	.146	-.018	.450	.197	.028
.750	.033	-.107	.750			.750	.065	-.085	.750	.100	-.035
.963	.024	-.022	.953	.023	-.118	.931	.043	-.100	.901	.073	-.055

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TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$M = 2.29; \alpha = -5^\circ; \beta = 80^\circ; \delta_e = 0^\circ$

.037	.206	.048	.048	.252	.110	.069	.349	.160	.100	.259	.077
.200	.134	-.007	.200	.183		.200	.268	.108	.200	.277	.061
.450	.076	-.048	.450	.104	-.039	.450	.156	.012	.450	.214	.057
.750	.032	-.093	.750			.750	.074	-.061	.750	.118	-.010
.963	.032	.062	.953	.024	-.103	.931	.057	-.078	.901	.087	-.029

$M = 2.29; \alpha = -5^\circ; \beta = 80^\circ; \delta_e = 0^\circ$

.037	.217	.088	.048	.271	.145	.069	.373	.198	.100	.283	.110
.200	.140	.024	.200	.196		.200	.291	.141	.200	.313	.093
.450	.083	-.025	.450	.112	-.016	.450	.171	.041	.450	.234	.087
.750	.035	-.076	.750			.750	.081	-.039	.750	.131	.015
.963	.033	.118	.953	.026	-.085	.931	.060	-.058	.901	.099	-.007

$M = 2.29; \alpha = 0^\circ; \beta = 80^\circ; \delta_e = 0^\circ$

.037	.066	.213	.048	.133	.250	.069	.205	.280	.100	.140	.185
.200	.016	.105	.200	.065		.200	.164	.236	.200	.120	.170
.450	-.008	.028	.450	-.006	.064	.450	.070	.116	.450	.104	.153
.750	-.033	-.037	.750			.750	-.010	.021	.750	.043	.080
.963	-.029	.169	.953	-.048	-.047	.931	-.030	-.004	.901	.020	.050

$M = 2.29; \alpha = 0^\circ; \beta = 80^\circ; \delta_e = 0^\circ$

.037	.049	.152	.048	.110	.200	.069	.179	.234	.100	.111	.141
.200	.011	.062	.200	.047		.200	.137	.190	.200	.093	.132
.450	-.013	.002	.450	-.015	.020	.450	.051	.081	.450	.082	.115
.750	-.042	-.056	.750			.750	-.025	-.007	.750	.022	.048
.963	-.032	.125	.953	-.053	-.065	.931	-.044	-.031	.901	.000	.022

$M = 2.29; \alpha = 0^\circ; \beta = 40^\circ; \delta_e = 0^\circ$

.037	.041	.115	.048	.096	.168	.069	.170	.220	.100	.096	.120
.200	.010	.032	.200	.034		.200	.123	.166	.200	.086	.119
.450	-.019	-.015	.450	-.015	-.003	.450	.035	.057	.450	.076	.100
.750	-.041	-.067	.750			.750	-.036	-.026	.750	.011	.029
.963	-.037	.106	.953	-.056	-.074	.931	-.052	-.048	.901	-.012	.005

$M = 2.29; \alpha = 0^\circ; \beta = 20^\circ; \delta_e = 0^\circ$

.037	.043	.073	.048	.081	.132	.069	.158	.194	.100	.085	.098
.200	.013	.009	.200	.027		.200	.108	.134	.200	.093	.113
.450	-.018	-.020	.450	-.010	-.021	.450	.022	.030	.450	.069	.083
.750	-.039	-.086	.750			.750	-.043	-.047	.750	-.004	.005
.963	-.040	.066	.953	-.051	-.081	.931	-.053	-.067	.901	-.026	-.018

$M = 2.29; \alpha = 0^\circ; \beta = 10^\circ; \delta_e = 0^\circ$

.037	.041	.058	.048	.075	.116	.069	.150	.182	.100	.080	.090
.200	.013	-.003	.200	.023		.200	.099	.118	.200	.096	.110
.450	-.017	-.028	.450	-.011	-.029	.450	.017	.021	.450	.062	.072
.750	-.041	-.088	.750			.750	-.045	-.056	.750	-.010	-.003
.963	-.042	.059	.953	-.049	-.086	.931	-.053	-.073	.901	-.030	-.025

$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.042	.043	.048	.067	.098	.069	.139	.162	.100	.082	.088
.200	.015	-.011	.200	.021		.200	.090	.104	.200	.104	.108
.450	-.012	-.032	.450	-.010	-.030	.450	.011	.008	.450	.056	.060
.750	-.041	-.097	.750			.750	-.046	-.066	.750	-.016	-.015
.963	-.044	.052	.953	-.047	-.091	.931	-.049	-.079	.901	-.037	-.037

TABLE II.- PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

M = 2.29; $\alpha = 0^\circ$; $\beta = -10^\circ$; $\delta_e = 0^\circ$

.037	.041	.027	.048	.063	.085	.069	.134	.150	.100	.094	.095
.200	.014	-.015	.200	.019		.200	.083	.091	.200	.111	.112
.450	-.011	-.040	.450	-.011	-.022	.450	.006	-.002	.450	.049	.047
.750	-.042	-.104	.750			.750	-.046	-.074	.750	-.021	-.024
.963	-.045	.047	.953	-.047	-.098	.931	-.050	-.081	.901	-.041	-.045

M = 2.29; $\alpha = 0^\circ$; $\beta = -20^\circ$; $\delta_e = 0^\circ$

.037	.041	.016	.048	.058	.070	.069	.127	.134	.100	.115	.109
.200	.015	-.018	.200	.020		.200	.077	.078	.200	.115	.116
.450	-.006	-.049	.450	-.009	-.020	.450	.003	-.012	.450	.043	.038
.750	-.042	-.100	.750			.750	-.043	-.077	.750	-.025	-.032
.963	-.047	.035	.953	-.046	-.103	.931	-.048	-.069	.901	-.044	-.053

M = 2.29; $\alpha = 0^\circ$; $\beta = -40^\circ$; $\delta_e = 0^\circ$

.037	.045	-.002	.048	.052	.049	.069	.113	.104	.100	.134	.130
.200	.016	-.013	.200	.025		.200	.065	.057	.200	.105	.099
.450	-.004	-.060	.450	-.006	-.026	.450	-.001	-.031	.450	.033	.021
.750	-.042	-.071	.750			.750	-.040	-.087	.750	-.032	-.048
.963	-.048	.001	.953	-.046	-.107	.931	-.048	-.066	.901	-.049	-.069

M = 2.29; $\alpha = 0^\circ$; $\beta = -60^\circ$; $\delta_e = 0^\circ$

.037	.058	-.060	.048	.043	.012	.069	.100	.067	.100	.118	.103
.200	.021	-.000	.200	.032		.200	.057	.021	.200	.090	.067
.450	-.001	-.080	.450	.004	-.026	.450	.001	-.058	.450	.024	-.008
.750	-.041	-.071	.750			.750	-.039	-.071	.750	-.038	-.071
.963	-.048	-.014	.953	-.048	-.106	.931	-.042	-.072	.901	-.050	-.090

M = 2.29; $\alpha = 0^\circ$; $\beta = -80^\circ$; $\delta_e = 0^\circ$

.037	.074	-.077	.048	.037	-.016	.069	.090	.032	.100	.106	.068
.200	.030	-.041	.200	.037		.200	.049	-.007	.200	.080	.036
.450	-.002	-.087	.450	.009	-.037	.450	.007	-.066	.450	.016	-.032
.750	-.040	-.082	.750			.750	-.031	-.083	.750	-.037	-.087
.963	-.048	-.070	.953	-.050	-.081	.931	-.038	-.082	.901	-.044	-.099

M = 2.29; $\alpha = 5^\circ$; $\beta = -80^\circ$; $\delta_e = 0^\circ$

.037	-.035	.138	.048	-.070	.148	.069	-.054	.181	.100	-.038	.172
.200	-.051	.090	.200	-.070		.200	-.068	.161	.200	-.050	.158
.450	-.064	.027	.450	-.078	.044	.450	-.100	.090	.450	-.093	.112
.750	-.103	-.012	.750			.750	-.116	.010	.750	-.125	.044
.963	-.106	.006	.953	-.109	.008	.931	-.113	-.011	.901	-.130	.028

M = 2.29; $\alpha = 5^\circ$; $\beta = -60^\circ$; $\delta_e = 0^\circ$

.037	-.048	.129	.048	-.074	.134	.069	-.036	.218	.100	-.019	.212
.200	-.060	.121	.200	-.081		.200	-.065	.147	.200	-.035	.191
.450	-.068	.025	.450	-.085	.067	.450	-.107	.102	.450	-.086	.102
.750	-.099	.015	.750			.750	-.127	.029	.750	-.126	.051
.963	-.109	.022	.953	-.110	.020	.931	-.120	.000	.901	-.135	.043

M = 2.29; $\alpha = 5^\circ$; $\beta = -40^\circ$; $\delta_e = 0^\circ$

.037	-.065	.140	.048	-.069	.177	.069	-.020	.264	.100	.001	.259
.200	-.072	.137	.200	-.095		.200	-.051	.180	.200	-.007	.233
.450	-.073	.034	.450		.083	.450	-.102	.115	.450	-.069	.128
.750	-.100	.036	.750			.750	-.135	.043	.750	-.120	.059
.963	-.106	.054	.953	-.106	.008	.931	-.134	.022	.901	-.132	.053

TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$M = 2.29; \alpha = 5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$

.037	-.075	.174	.048	-.053	.223	.069	-.007	.306	.100	-.040	.263
.200	-.084	.147	.200	-.092	.200	.200	-.035	.215	.200	-.027	.256
.450	-.077	.036	.450	-.106	.098	.450	-.092	.113	.450	-.056	.153
.750	-.101	.026	.750			.750	-.137	.045	.750	-.104	.061
.963	-.104	.093	.953	-.111	-.016	.931	-.135	.035	.901	-.115	.048

$M = 2.29; \alpha = 5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$

.037	-.078	.187	.048	-.045	.244	.069	-.001	.314	.100	-.040	.245
.200	-.088	.135	.200	-.088	.200	.200	-.030	.223	.200	-.034	.251
.450	-.085	.055	.450	-.111	.111	.450	-.084	.116	.450	-.054	.164
.750	-.102	.002	.750			.750	-.133	.057	.750	-.099	.073
.963	-.105	.114	.953	-.114	-.009	.931	-.136	.042	.901	-.110	.054

$M = 2.29; \alpha = 5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	-.076	.199	.048	-.037	.251	.069	-.009	.330	.100	-.037	.242
.200	-.092	.112	.200	-.082	.200	.200	-.024	.248	.200	-.038	.266
.450	-.090	.065	.450	-.114	.112	.450	-.076	.121	.450	-.051	.177
.750	-.101	-.009	.750			.750	-.127	.056	.750	-.093	.083
.963	-.104	.128	.953	-.113	-.005	.931	-.133	.043	.901	-.103	.059

$M = 2.29; \alpha = 5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$

.037	-.075	.214	.048	-.037	.277	.069	-.012	.346	.100	-.034	.236
.200	-.100	.111	.200	-.077	.200	.200	-.019	.263	.200	-.036	.268
.450	-.097	.076	.450	-.116	.107	.450	-.071	.132	.450	-.053	.191
.750	-.101	-.004	.750			.750	-.122	.066	.750	-.089	.093
.963	-.104	.140	.953	-.119	.006	.931	-.130	.042	.901	-.095	.065

$M = 2.29; \alpha = 5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$

.037	-.074	.225	.048	-.033	.295	.069	-.019	.356	.100	-.025	.232
.200	-.105	.146	.200	-.072	.200	.200	-.011	.274	.200	-.030	.265
.450	-.103	.086	.450	-.117	.111	.450	-.069	.141	.450	-.053	.202
.750	-.102	.001	.750			.750	-.118	.071	.750	-.086	.102
.963	-.104	.146	.953	-.124	.006	.931	-.125	.045	.901	-.090	.072

$M = 2.29; \alpha = 5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$

.037	-.068	.273	.048	-.020	.331	.069	-.015	.371	.100	-.010	.245
.200	-.108	.170	.200	-.084	.200	.200	-.006	.303	.200	-.020	.269
.450	-.109	.082	.450	-.111	.131	.450	-.056	.165	.450	-.057	.225
.750	-.105	.021	.750			.750	-.108	.090	.750	-.074	.125
.963	-.101	.184	.953	-.134	.024	.931	-.113	.064	.901	-.074	.094

$M = 2.29; \alpha = 5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.037	-.058	.319	.048	-.002	.370	.069	-.024	.396	.100	-.011	.273
.200	-.104	.203	.200	-.050	.200	.200	-.025	.333	.200	-.010	.289
.450	-.106	.090	.450	-.101	.157	.450	-.039	.193	.450	-.045	.250
.750	-.108	.030	.750			.750	-.096	.112	.750	-.062	.153
.963	-.098	.235	.953	-.138	.031	.931	-.103	.087	.901	-.060	.118

$M = 2.29; \alpha = 5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.037	-.047	.371	.048	-.017	.416	.069	-.047	.446	.100	-.031	.312
.200	-.097	.243	.200	-.034	.200	.200	-.047	.375	.200	-.011	.327
.450	-.100	.112	.450	-.090	.195	.450	-.021	.227	.450	-.028	.284
.750	-.103	.051	.750			.750	-.082	.142	.750	-.047	.178
.963	-.097	.258	.953	-.135	.048	.931	-.092	.111	.901	-.043	.147

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TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

M = 2.29; $\alpha = -8^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$

.037	.253	-.081	.048	.331	-.044	.069	.397	-.006	.100	.418	.016
.200	.158	-.074	.200	.235		.200	.320	-.040	.200	.361	-.002
.450	.131	-.102	.450	.147	-.113	.450	.205	-.098	.450	.249	-.061
.750	.073	-.152	.750			.750	.102	-.138	.750	.145	-.105
.963	.058	-.047	.953	.068	-.143	.931	.090	-.134	.901	.113	-.109

M = 2.29; $\alpha = -8^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$

.037	.177	-.033	.048	.261	-.030	.069	.331	.019	.100	.346	.030
.200	.110	-.066	.200	.188		.200	.258	-.020	.200	.302	.026
.450	.093	-.090	.450	.103	-.092	.450	.159	-.078	.450	.198	-.044
.750	.045	-.147	.750			.750	.065	-.126	.750	.105	-.091
.963	.030	-.016	.953	.038	-.138	.931	.046	-.126	.901	.077	-.100

M = 2.29; $\alpha = -4^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$

.037	.057	-.027	.048	.212	.012	.069	.274	.072	.100	.265	.036
.200	.072	-.055	.200	.127		.200	.204	.026	.200	.245	.056
.450	.058	-.079	.450	.060	-.076	.450	.105	-.050	.450	.151	-.002
.750	.012	-.134	.750			.750	.028	-.107	.750	.062	-.062
.963	.002	.019	.953	.007	-.122	.931	.009	-.115	.901	.034	-.079

M = 2.29; $\alpha = -2^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$

.037	.051	.022	.048	.144	.065	.069	.219	.130	.100	.156	.059
.200	.025	-.012	.200	.073		.200	.156	.073	.200	.178	.079
.450	.016	-.043	.450	.027	-.051	.450	.059	-.015	.450	.108	.038
.750	-.018	-.110	.750			.750	-.011	-.083	.750	.023	-.031
.963	-.020	.058	.953	-.020	-.104	.931	-.024	-.097	.901	-.003	-.052

M = 2.29; $\alpha = 0^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$

.037	.025	.068	.048	.086	.118	.069	.154	.183	.100	.081	.093
.200	-.003	.016	.200	.026		.200	.103	.121	.200	.101	.116
.450	-.015	-.014	.450	-.021	.003	.450	.019	.023	.450	.064	.075
.750	-.042	-.076	.750			.750	-.049	-.054	.750	-.010	.001
.963	-.046		.953	-.050	-.082	.931	-.064	-.059	.901	-.032	-.023

M = 2.29; $\alpha = 2^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$

.037	-.012	.131	.048	-.032	.170	.069	.088	.240	.100	.032	.144
.200	-.037	.061	.200	-.018		.200	.051	.173	.200	.041	.172
.450	-.046	.018	.450	-.058	.042	.450	-.020	.060	.450	.016	.116
.750	-.067	-.059	.750			.750	-.080	-.010	.750	-.043	.034
.963	-.069	.103	.953	-.077	-.053	.931	-.092	-.013	.901	-.062	.008

M = 2.75; $\alpha = -8^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$

.037	.252	-.029	.048	.293	.021	.069	.374	.026	.100	.295	.013
.200	.160	-.063	.200	.230		.200	.310	.024	.200	.300	-.003
.450	.111	-.051	.450	.138	-.073	.450	.208	-.035	.450	.244	-.029
.750	.070	-.091	.750			.750	.117	-.078	.750	.152	-.049
.963	.063	-.087	.953	.069	-.089	.931	.093	-.080	.901	.125	-.053

M = 2.75; $\alpha = -8^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$

.037	.172	-.007	.048	.240	.041	.069	.268	.015	.100	.227	.029
.200	.119	-.042	.200	.169		.200	.254	.031	.200	.211	.013
.450	.077	-.052	.450	.101	-.063	.450	.155	-.019	.450	.186	-.026
.750	.041	-.091	.750			.750	.079	-.072	.750	.110	-.044
.963	.034	-.077	.953	.038	-.089	.931	.059	-.079	.901	.083	-.048

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TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
M = 2.75; $\alpha = -40^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$											
.037	.125	.012	.048	.198	.077	.069	.179	.032	.100	.172	.051
.200	.081	-.023	.200	.124	.200	.200	.197	.045	.200	.154	.032
.450	.046	-.042	.450	.065	-.049	.450	.117	.009	.450	.125	-.011
.750	.015	-.083	.750	.011	-.082	.750	.041	-.054	.750	.076	-.027
.963	.010	-.079	.953	.011	-.082	.931	.026	-.071	.901	.054	-.032
M = 2.75; $\alpha = -20^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$											
.037	.073	.046	.048	.147	.120	.069	.119	.070	.100	.121	.077
.200	.038	-.004	.200	.077	.200	.200	.127	.072	.200	.103	.053
.450	.013	-.029	.450	.021	-.025	.450	.078	.037	.450	.065	.017
.750	-.010	-.067	.750	.021	-.025	.750	.006	-.035	.750	.035	-.006
.963	-.014	-.080	.953	-.019	-.065	.931	-.011	-.052	.901	.020	-.012
M = 2.75; $\alpha = 0^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$											
.037	.044	.071	.048	.100	.160	.069	.074	.102	.100	.082	.108
.200	.007	.024	.200	.039	.200	.200	.069	.101	.200	.067	.096
.450	-.015	-.007	.450	-.013	.005	.450	.042	.064	.450	.025	.046
.750	-.032	-.050	.750	-.041	-.047	.750	-.020	-.011	.750	-.003	.017
.963	-.034	-.066	.953	-.041	-.047	.931	-.037	-.029	.901	-.011	.008
M = 2.75; $\alpha = 20^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$											
.037	.005	.109	.048	.047	.198	.069	.033	.141	.100	.045	.142
.200	-.031	.057	.200	.001	.200	.200	.019	.139	.200	.032	.133
.450	-.042	.023	.450	-.045	.032	.450	.002	.095	.450	-.009	.078
.750	-.054	-.027	.750	-.066	-.026	.750	-.047	.016	.750	-.040	.048
.963	-.056	-.040	.953	-.066	-.026	.931	-.060	-.001	.901	-.044	.035
M = 2.75; $\alpha = 40^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$											
.037	-.024	.161	.048	.008	.249	.069	-.000	.185	.100	.011	.182
.200	-.062	.094	.200	-.032	.200	.200	-.015	.188	.200	-.001	.174
.450	-.067	.056	.450	-.070	.062	.450	-.030	.130	.450	-.035	.122
.750	-.072	-.003	.750	-.085	.006	.750	-.068	.048	.750	-.065	.085
.963	-.072	-.007	.953	-.085	.006	.931	-.072	.022	.901	-.066	.068
M = 2.75; $\alpha = -50^\circ$; $\beta = -80^\circ$; $\delta_e = 0^\circ$											
.037	.101	-.062	.048	.126	-.029	.069	.212	-.011	.100	.233	.001
.200	.076	-.093	.200	.095	.200	.200	.153	-.037	.200	.193	-.006
.450	.056	-.077	.450	.071	-.098	.450	.086	-.073	.450	.118	-.057
.750	.011	-.108	.750	.013	-.111	.750	.037	-.104	.750	.051	-.085
.963	.004	-.092	.953	.013	-.111	.931	.033	-.107	.901	.037	-.091
M = 2.75; $\alpha = -50^\circ$; $\beta = -60^\circ$; $\delta_e = 0^\circ$											
.037	.106	-.048	.048	.148	-.016	.069	.246	.012	.100	.198	-.021
.200	.077	-.093	.200	.107	.200	.200	.177	-.021	.200	.205	-.019
.450	.060	-.067	.450	.074	-.099	.450	.095	-.072	.450	.138	-.042
.750	.017	-.116	.750	.018	-.115	.750	.044	-.106	.750	.062	-.080
.963	.012	-.081	.953	.018	-.115	.931	.035	-.112	.901	.043	-.090
M = 2.75; $\alpha = -50^\circ$; $\beta = -40^\circ$; $\delta_e = 0^\circ$											
.037	.115	-.028	.048	.175	-.002	.069	.257	.014	.100	.184	-.006
.200	.084	-.062	.200	.122	.200	.200	.199	-.003	.200	.191	-.019
.450	.063	-.061	.450	.074	-.085	.450	.110	-.056	.450	.150	-.038
.750	.023	-.113	.750	.024	-.106	.750	.050	-.095	.750	.071	-.065
.963	.017	-.087	.953	.024	-.106	.931	.033	-.101	.901	.049	-.075

TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$M = 2.75; \alpha = -5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$

.037	.124	-.015	.048	.201	.023	.069	.243	.011	.100	.189	.012
.200	.091	-.044	.200	.133		.200	.216	.025	.200	.179	.000
.450	.063	-.054	.450	.078	-.072	.450	.120	-.032	.450	.157	-.031
.750	.026	-.102	.750			.750	.054	-.081	.750	.081	-.049
.963	.019	-.088	.953	.025	-.097	.931	.037	-.091	.901	.056	-.057

$M = 2.75; \alpha = -5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$

.037	.133	-.002	.048	.210	.036	.069	.226	.013	.100	.190	.026
.200	.095	-.041	.200	.136		.200	.222	.038	.200	.176	.010
.450	.061	-.050	.450	.080	-.064	.450	.127	-.017	.450	.156	-.025
.750	.027	-.095	.750			.750	.055	-.070	.750	.087	-.042
.963	.020	-.086	.953	.025	-.090	.931	.041	-.084	.901	.061	-.048

$M = 2.75; \alpha = -5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.145	.007	.048	.220	.054	.069	.222	.019	.100	.200	.034
.200	.103	-.034	.200	.145		.200	.229	.031	.200	.183	.018
.450	.064	-.051	.450	.085	-.057	.450	.136	-.004	.450	.159	-.020
.750	.029	-.088	.750			.750	.059	-.061	.750	.093	-.035
.963	.022	-.075	.953	.025	-.085	.931	.045	-.075	.901	.068	-.039

$M = 2.75; \alpha = -5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$

.037	.152	.016	.048	.225	.074	.069	.211	.030	.100	.203	.052
.200	.105	-.026	.200	.147		.200	.229	.044	.200	.184	.034
.450	.061	-.046	.450	.086	-.047	.450	.139	.009	.450	.154	-.014
.750	.028	-.082	.750			.750	.059	-.053	.750	.096	-.029
.963	.022	-.058	.953	.022	-.082	.931	.045	-.069	.901	.071	-.034

$M = 2.75; \alpha = -5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$

.037	.161	.027	.048	.235	.093	.069	.212	.043	.100	.213	.066
.200	.111	-.017	.200	.154		.200	.231	.054	.200	.193	.047
.450	.064	-.041	.450	.089	-.037	.450	.147	.023	.450	.156	-.003
.750	.028	-.073	.750			.750	.063	-.042	.750	.103	-.019
.963	.023	-.046	.953	.022	-.077	.931	.048	-.059	.901	.078	-.024

$M = 2.75; \alpha = -5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$

.037	.180	.052	.048	.255	.127	.069	.225	.067	.100	.234	.095
.200	.122	.006	.200	.170		.200	.245	.078	.200	.212	.075
.450	.072	-.031	.450	.098	-.013	.450	.162	.048	.450	.168	.018
.750	.030	-.062	.750			.750	.075	-.022	.750	.119	-.001
.963	.027	-.015	.953	.028	-.066	.931	.059	-.040	.901	.093	-.004

$M = 2.75; \alpha = -5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$

.037	.201	.083	.048	.278	.162	.069	.245	.092	.100	.257	.121
.200	.140	.033	.200	.190		.200	.268	.106	.200	.233	.102
.450	.082	-.011	.450	.114	.007	.450	.184	.072	.450	.182	.042
.750	.037	-.050	.750			.750	.091	-.001	.750	.135	.020
.963	.034	.018	.953	.035	-.048	.931	.071	-.021	.901	.112	.017

$M = 2.75; \alpha = -5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.037	.219	.121	.048	.301	.202	.069	.269	.124	.100	.278	.151
.200	.156	.066	.200	.211		.200	.298	.143	.200	.254	.126
.450	.097	.019	.450	.129	.034	.450	.205	.098	.450	.204	.064
.750	.048	-.031	.750			.750	.107	.019	.750	.157	.046
.963	.043	.068	.953	.045	-.027	.931	.088	-.003	.901	.132	.038

TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
$M = 2.75; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$											
.037	.102	.224	.048	.179	.305	.069	.153	.207	.100	.172	.216
.200	.044	.148	.200	.110	.200	.200	.140	.213	.200	.149	.196
.450	.013	.074	.450	.037	.106	.450	.112	.165	.450	.089	.128
.750	-.014	.000	.750			.750	.035	.071	.750	.047	.095
.963	-.015	.064	.953	-.024	.013	.931	.013	.052	.901	.044	.086
$M = 2.75; \alpha = 0^\circ; \beta = 6^\circ; \delta_e = 0^\circ$											
.037	.084	.176	.048	.159	.262	.069	.132	.184	.100	.146	.179
.200	.028	.109	.200	.091	.200	.200	.115	.168	.200	.125	.160
.450	.003	.041	.450	.022	.076	.450	.090	.129	.450	.070	.104
.750	-.025	-.022	.750			.750	.019	.045	.750	.029	.067
.963	-.024	-.006	.953	-.032	-.007	.931	-.002	.027	.901	.025	.060
$M = 2.75; \alpha = 0^\circ; \beta = 4^\circ; \delta_e = 0^\circ$											
.037	.085	.175	.048	.159	.260	.069	.132	.182	.100	.146	.178
.200	.028	.108	.200	.092	.200	.200	.115	.166	.200	.124	.160
.450	.004	.041	.450	.022	.076	.450	.090	.129	.450	.070	.102
.750	-.024	-.023	.750			.750	.019	.044	.750	.029	.066
.963	-.024	-.006	.953	-.031	-.008	.931	-.002	.026	.901	.025	.060
$M = 2.75; \alpha = 0^\circ; \beta = 2^\circ; \delta_e = 0^\circ$											
.037	.058	.101	.048	.126	.187	.069	.094	.120	.100	.102	.115
.200	.014	.042	.200	.059	.200	.200	.081	.106	.200	.085	.106
.450	-.007	-.000	.450	-.001	.021	.450	.058	.076	.450	.040	.057
.750	-.029	-.041	.750			.750	-.008	.001	.750	.007	.022
.963	-.029	-.059	.953	-.041	-.040	.931	-.026	-.017	.901	.001	.016
$M = 2.75; \alpha = 0^\circ; \beta = 1^\circ; \delta_e = 0^\circ$											
.037	.053	.082	.048	.118	.169	.069	.086	.105	.100	.094	.105
.200	.012	.029	.200	.052	.200	.200	.077	.096	.200	.078	.097
.450	-.009	-.008	.450	-.003	.009	.450	.051	.064	.450	.034	.045
.750	-.027	-.049	.750			.750	-.014	-.009	.750	.004	.015
.963	-.030	-.064	.953	-.040	-.047	.931	-.030	-.026	.901	-.003	.007
$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.048	.065	.048	.109	.153	.069	.079	.091	.100	.087	.098
.200	.011	.015	.200	.046	.200	.200	.076	.088	.200	.072	.084
.450	-.010	-.013	.450	-.005	-.003	.450	.046	.052	.450	.030	.035
.750	-.026	-.055	.750			.750	-.019	-.019	.750	.003	.008
.963	-.032	-.069	.953	-.041	-.052	.931	-.035	-.035	.901	-.006	.001
$M = 2.75; \alpha = 0^\circ; \beta = -1^\circ; \delta_e = 0^\circ$											
.037	.040	.047	.048	.103	.138	.069	.075	.078	.100	.081	.090
.200	.013	.011	.200	.041	.200	.200	.076	.082	.200	.066	.073
.450	-.008	-.016	.450	-.005	-.009	.450	.040	.041	.450	.028	.027
.750	-.027	-.061	.750			.750	-.024	-.028	.750	.001	.002
.963	-.032	-.064	.953	-.035	-.058	.931	-.038	-.042	.901	-.009	-.005
$M = 2.75; \alpha = 0^\circ; \beta = -2^\circ; \delta_e = 0^\circ$											
.037	.035	.034	.048	.097	.121	.069	.071	.066	.100	.075	.079
.200	.014	.007	.200	.036	.200	.200	.077	.075	.200	.060	.061
.450	-.005	-.020	.450	-.006	-.014	.450	.035	.030	.450	.026	.021
.750	-.026	-.069	.750			.750	-.027	-.036	.750	-.001	-.002
.963	-.032	-.060	.953	-.036	-.063	.931	-.039	-.048	.901	-.013	-.011

TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

M = 2.75; $\alpha = 0^\circ$; $\beta = -4^\circ$; $\delta_e = 0^\circ$

.037	.033	.014	.048	.084	.086	.069	.067	.049	.100	.066	.058
.200	.019	-.023	.200	.027		.200	.085	.067	.200	.051	.040
.450	.001	-.020	.450	.001	-.043	.450	.024	.011	.450	.029	.015
.750	-.027	-.077	.750			.750	-.032	-.049	.750	-.004	-.011
.963	-.032	-.054	.953	-.033	-.072	.931	-.037	-.064	.901	-.020	-.024

M = 2.75; $\alpha = 0^\circ$; $\beta = -6^\circ$; $\delta_e = 0^\circ$

.037	.036	-.015	.048	.060	.058	.069	.095	.061	.100	.053	.033
.200	.021	-.047	.200	.019		.200	.079	.062	.200	.046	.023
.450	.005	-.028	.450	-.000	-.057	.450	.013	-.014	.450	.036	.014
.750	-.029	-.064	.750			.750	-.031	-.068	.750	-.012	-.026
.963	-.035	-.049	.953	-.033	-.084	.931	-.034	-.078	.901	-.029	-.043

M = 2.75; $\alpha = 0^\circ$; $\beta = -8^\circ$; $\delta_e = 0^\circ$

.037	.046	-.049	.048	.045	.017	.069	.107	.069	.100	.058	.028
.200	.019	-.043	.200	.019		.200	.059	.026	.200	.073	.034
.450	.007	-.061	.450	-.004	-.035	.450	.001	-.035	.450	.034	.005
.750	-.032	-.065	.750			.750	-.028	-.079	.750	-.021	-.048
.963	-.038	-.024	.953	-.036	-.076	.931	-.035	-.078	.901	-.037	-.061

M = 2.75; $\alpha = 5^\circ$; $\beta = -8^\circ$; $\delta_e = 0^\circ$

.037	-.033	.071	.048	-.049	.125	.069	-.023	.178	.100	-.012	.165
.200	-.046	.059	.200	-.060		.200	-.044	.130	.200	-.016	.157
.450	-.052	.019	.450	-.066	.060	.450	-.071	.045	.450	-.058	.101
.750	-.076	-.047	.750			.750	-.091	.031	.750	-.087	.031
.963	-.080	.027	.953	-.081	-.003	.931	-.091	.013	.901	-.093	.012

M = 2.75; $\alpha = 5^\circ$; $\beta = -6^\circ$; $\delta_e = 0^\circ$

.037	-.046	.083	.048	-.047	.151	.069	-.008	.217	.100	-.036	.144
.200	-.052	.079	.200	-.070		.200	-.033	.163	.200	-.036	.170
.450	-.056	.023	.450	-.076	.076	.450	-.073	.070	.450	-.050	.126
.750	-.076	-.008	.750			.750	-.100	.027	.750	-.081	.051
.963	-.085	.028	.953	-.084	-.007	.931	-.103	.027	.901	-.090	.027

M = 2.75; $\alpha = 5^\circ$; $\beta = -4^\circ$; $\delta_e = 0^\circ$

.037	-.052	.107	.048	-.027	.193	.069	-.035	.195	.100	-.028	.157
.200	-.064	.087	.200	-.063		.200	-.029	.191	.200	-.038	.157
.450	-.054	.038	.450	-.086	.084	.450	-.058	.095	.450	-.064	.135
.750	-.077	.015	.750			.750	-.097	.018	.750	-.079	.066
.963	-.082	.012	.953	-.084	-.003	.931	-.099	.036	.901	-.082	.043

M = 2.75; $\alpha = 5^\circ$; $\beta = -2^\circ$; $\delta_e = 0^\circ$

.037	-.043	.160	.048	-.022	.235	.069	-.027	.199	.100	-.017	.180
.200	-.073	.099	.200	-.056		.200	-.033	.213	.200	-.028	.179
.450	-.072	.057	.450	-.087	.086	.450	-.051	.127	.450	-.057	.145
.750	-.076	.002	.750			.750	-.088	.042	.750	-.081	.088
.963	-.079	.029	.953	-.086	.007	.931	-.088	.032	.901	-.080	.066

M = 2.75; $\alpha = 5^\circ$; $\beta = -1^\circ$; $\delta_e = 0^\circ$

.037	-.044	.175	.048	-.017	.255	.069	-.022	.209	.100	-.012	.192
.200	-.077	.108	.200	-.053		.200	-.035	.216	.200	-.024	.186
.450	-.079	.066	.450	-.087	.086	.450	-.050	.138	.450	-.055	.145
.750	-.081	.003	.750			.750	-.085	.049	.750	-.081	.095
.963	-.080	.013	.953	-.092	.015	.931	-.084	.034	.901	-.079	.074

TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$M = 2.75; \alpha = 5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	-.044	.187	.048	-.010	.274	.069	-.016	.212	.100	-.006	.201
.200	-.079	.113	.200	-.048		.200	-.031	.216	.200	-.018	.194
.450	-.082	.070	.450	-.083	.085	.450	-.046	.148	.450	-.049	.145
.750	-.081	.010	.750			.750	-.080	.059	.750	-.078	.103
.963	-.081	.008	.953	-.096	.018	.931	-.080	.036	.901	-.076	.083

$M = 2.75; \alpha = 5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$

.037	-.042	.205	.048	-.003	.293	.069	-.009	.222	.100	.003	.217
.200	-.080	.123	.200	-.043		.200	-.027	.220	.200	-.011	.208
.450	-.085	.077	.450	-.080	.093	.450	-.044	.160	.450	-.043	.147
.750	-.085	.014	.750			.750	-.077	.063	.750	-.074	.101
.963	-.081	.019	.953	-.099	.021	.931	-.077	.041	.901	-.072	.085

$M = 2.75; \alpha = 5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$

.037	-.038	.226	.048	-.006	.315	.069	-.001	.237	.100	.013	.227
.200	-.079	.137	.200	-.036		.200	-.019	.226	.200	-.003	.214
.450	-.085	.083	.450	-.077	.102	.450	-.040	.165	.450	-.038	.160
.750	-.087	.021	.750			.750	-.071	.083	.750	-.069	.120
.963	-.082	.025	.953	-.100	.023	.931	-.071	.055	.901	-.068	.102

$M = 2.75; \alpha = 5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$

.037	-.027	.268	.048	-.027	.354	.069	.013	.262	.100	.033	.271
.200	-.073	.162	.200	-.021		.200	-.004	.249	.200	.016	.256
.450	-.084	.105	.450	-.068	.120	.450	-.029	.197	.450	-.024	.179
.750	-.089	.038	.750			.750	-.060	.104	.750	-.058	.133
.963	-.084	.033	.953	-.098	.035	.931	-.062	.072	.901	-.053	.118

$M = 2.75; \alpha = 5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$

.037	-.014	.321	.048	-.049	.403	.069	.033	.305	.100	.047	.302
.200	-.063	.198	.200	-.004		.200	.012	.289	.200	.039	.293
.450	-.076	.131	.450	-.057	.157	.450	-.013	.233	.450	-.007	.209
.750	-.087	.058	.750			.750	-.047	.128	.750	-.045	.164
.963	-.083	.066	.953	-.094	.052	.931	-.053	.097	.901	-.040	.148

$M = 2.75; \alpha = 5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.037	.004	.374	.048	-.071	.454	.069	.054	.339	.100	.067	.328
.200	-.049	.235	.200	.015		.200	.030	.338	.200	.058	.330
.450	-.064	.161	.450	-.042	.184	.450	.007	.270	.450	.010	.250
.750	-.075	.074	.750			.750	-.033	.158	.750	-.030	.201
.963	-.073	.141	.953	-.086	.073	.931	-.038	.124	.901	-.023	.179

$M = 2.75; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.037	.230	-.027	.048	.290	.008	.069	.378	.018	.100	.299	.009
.200	.154	-.039	.200	.226		.200	.310	.016	.200	.305	-.008
.450	.110	-.053	.450	.138	-.065	.450	.207	-.043	.450	.245	-.035
.750	.070	-.090	.750			.750	.117	-.080	.750	.152	-.056
.963	.063	-.084	.953	.070	-.085	.931	.093	-.079	.901	.126	-.057

$M = 2.75; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.037	.164	-.006	.048	.250	.023	.069	.263	.007	.100	.231	.025
.200	.113	-.025	.200	.172		.200	.257	.014	.200	.214	.009
.450	.079	-.052	.450	.103	-.055	.450	.159	-.026	.450	.188	-.035
.750	.042	-.090	.750			.750	.078	-.073	.750	.112	-.054
.963	.036	-.068	.953	.040	-.087	.931	.061	-.077	.901	.086	-.054

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TABLE II.- PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

 $M = 2.75; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.037	.089	.014	.048	.213	.061	.069	.176	.021	.100	.174	.043
.200	.068	-.015	.200	.128	.061	.200	.195	.025	.200	.155	.025
.450	.045	-.047	.450	.063	-.042	.450	.121	.004	.450	.123	-.022
.750	.015	-.084	.750			.750	.040	-.055	.750	.078	-.039
.963	.010	-.080	.953	.011	-.079	.931	.023	-.070	.901	.056	-.039

 $M = 2.75; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.037	.073	.044	.048	.169	.118	.069	.118	.046	.100	.129	.071
.200	.014	.009	.200	.091	.091	.200	.126	.055	.200	.109	.049
.450	.014	-.025	.450	.018	-.022	.450	.087	.033	.450	.066	.002
.750	-.009	-.067	.750			.750	.013	-.034	.750	.037	-.015
.963	-.012	-.078	.953	-.016	-.064	.931	-.006	-.052	.901	.024	-.018

 $M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.037	.048	.084	.048	.105	.163	.069	.076	.096	.100	.084	.095
.200	-.002	.035	.200	.048		.200	.064	.089	.200	.069	.086
.450	-.016	-.002	.450	-.010	.011	.450	.046	.063	.450	.027	.041
.750	-.032	-.047	.750			.750	-.014	-.009	.750	-.004	.012
.963	-.032	-.063	.953	-.042	-.045	.931	-.032	-.026	.901	-.011	.005

 $M = 2.75; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.037	.009	.120	.048	.055	.205	.069	.039	.135	.100	.051	.137
.200	-.029	.063	.200	.009		.200	.024	.132	.200	.037	.127
.450	-.042	.028	.450	-.037	.034	.450	.006	.096	.450	-.004	.074
.750	-.052	-.027	.750			.750	-.040	.016	.750	-.036	.044
.963	-.052	-.040	.953	-.064	-.025	.931	-.054	.001	.901	-.041	.035

 $M = 2.75; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.037	-.019	.170	.048	-.016	.254	.069	.007	.182	.100	.017	.179
.200	-.058	.101	.200	-.025		.200	-.009	.185	.200	.004	.170
.450	-.065	.060	.450	-.062	.066	.450	-.024	.133	.450	-.030	.117
.750	-.071	-.003	.750			.750	-.062	.047	.750	-.061	.083
.963	-.068	-.009	.953	-.083	.007	.931	-.069	.023	.901	-.062	.067

 $M = 3.22; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.237	-.009	.048	.316	.071	.069	.295	.051	.100	.252	-.006
.200	.158	-.029	.200	.228		.200	.291	.028	.200	.254	.011
.450	.106	-.023	.450	.145	-.039	.450	.210	.002	.450	.216	.002
.750	.065	-.050	.750			.750	.123	-.041	.750	.159	-.028
.963	.061	-.060	.953	.062	-.053	.931	.100	-.041	.901	.135	-.029

 $M = 3.22; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.179	-.001	.048	.263	.085	.069	.230	.062	.100	.235	.076
.200	.124	-.017	.200	.176		.200	.217	.037	.200	.215	.066
.450	.077	-.024	.450	.109	-.029	.450	.166	.007	.450	.162	.013
.750	.042	-.050	.750			.750	.085	-.035	.750	.115	-.025
.963	.040	-.060	.953	.039	-.053	.931	.065	-.039	.901	.097	-.031

 $M = 3.22; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.137	.027	.048	.210	.106	.069	.172	.077	.100	.184	.108
.200	.077	.003	.200	.133		.200	.152	.050	.200	.164	.078
.450	.046	-.021	.450	.064	-.009	.450	.121	.020	.450	.113	.024
.750	.018	-.046	.750			.750	.053	-.020	.750	.069	-.015
.963	.014	-.057	.953	.012	-.051	.931	.032	-.029	.901	.058	-.024

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TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.094	.052	.048	.163	.141	.069	.124	.097	.100	.136	.123
.200	.036	.026	.200	.094	.200	.200	.102	.068	.200	.116	.098
.450	.018	-.015	.450	.031	.008	.450	.075	.039	.450	.073	.042
.750	-.005	-.040	.750			.750	.025	-.001	.750	.029	.000
.963	-.008	-.051	.953	-.014	-.045	.931	.006	-.015	.901	.019	-.008

$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.066	.086	.048	.106	.166	.069	.084	.118	.100	.096	.131
.200	.011	.047	.200	.059	.200	.200	.103	.079	.200	.117	.117
.450	-.011	.002	.450	.004	.030	.450	.036	.063	.450	.038	.061
.750	-.025	-.030	.750			.750	.002	.018	.750	.002	.023
.963	-.025	-.041	.953	-.032	-.027	.931	-.013	.001	.901	-.007	.013

$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.033	.121	.048	.066	.201	.069	.050	.157	.100	.061	.153
.200	-.018	.077	.200	.024	.200	.200	.036	.140	.200	.047	.141
.450	-.034	.028	.450	-.022	.058	.450	.006	.095	.450	.010	.092
.750	-.045	-.013	.750			.750	-.023	.040	.750	-.021	.049
.963	-.042	-.026	.953	-.050	-.005	.931	-.029	.020	.901	-.029	.037

$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.026	.159	.048	.035	.253	.069	.024	.207	.100	.033	.182
.200	-.043	.112	.200	-.003	.200	.200	.008	.177	.200	.020	.173
.450	-.054	.058	.450	-.043	.090	.450	-.018	.134	.450	-.014	.132
.750	-.059	.012	.750			.750	-.039	.065	.750	-.039	.079
.963	-.057	-.006	.953	-.052	.019	.931	-.039	.046	.901	-.043	.069

$M = 3.22; \alpha = -5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$

.037	.098	-.024	.048	.144	.015	.069	.186	-.035	.100	.158	.002
.200	.077	-.049	.200	.087	.200	.200	.165	-.001	.200	.144	-.013
.450	.054	-.039	.450	.060	-.058	.450	.088	-.036	.450	.122	-.032
.750	.011	-.073	.750			.750	.029	-.062	.750	.057	-.047
.963	.008	-.059	.953	.016	-.071	.931	.025	-.063	.901	.037	-.051

$M = 3.22; \alpha = -5^\circ; \beta = -6^\circ; \delta_e = 0^\circ$

.037	.114	-.020	.048	.168	.019	.069	.169	-.006	.100	.168	.020
.200	.081	-.053	.200	.103	.200	.200	.174	-.012	.200	.148	-.003
.450	.058	-.034	.450	.068	-.063	.450	.101	-.032	.450	.120	-.038
.750	.019	-.081	.750			.750	.039	-.065	.750	.067	-.055
.963	.016	-.069	.953	.021	-.076	.931	.031	-.069	.901	.046	-.055

$M = 3.22; \alpha = -5^\circ; \beta = -4^\circ; \delta_e = 0^\circ$

.037	.123	-.010	.048	.192	.033	.069	.178	-.016	.100	.180	.037
.200	.086	-.040	.200	.120	.200	.200	.177	-.004	.200	.161	.015
.450	.057	-.029	.450	.076	-.051	.450	.117	-.024	.450	.122	-.025
.750	.024	-.072	.750			.750	.047	-.059	.750	.079	-.051
.963	.019	-.069	.953	.024	-.071	.931	.035	-.063	.901	.057	-.053

$M = 3.22; \alpha = -5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$

.037	.136	-.001	.048	.217	.058	.069	.189	.041	.100	.193	.061
.200	.093	-.022	.200	.136	.200	.200	.177	.016	.200	.172	.036
.450	.054	-.023	.450	.078	-.039	.450	.130	-.011	.450	.126	-.008
.750	.027	-.062	.750			.750	.055	-.047	.750	.084	-.037
.963	.022	-.071	.953	.022	-.062	.931	.038	-.048	.901	.066	-.043

TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

 $M = 3.22; \alpha = -5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$

.037	.141	.004	.048	.225	.076	.069	.192	.053	.100	.202	.082
.200	.095	-.014	.200	.143	.200	.177	.177	.031	.200	.179	.057
.450	.056	-.026	.450	.080	-.028	.450	.136	.002	.450	.128	.006
.750	.028	-.056	.750	.021	-.056	.750	.060	-.038	.750	.086	-.029
.963	.024	-.065	.953			.931	.041	-.042	.901	.070	-.036

 $M = 3.22; \alpha = -5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.153	.010	.048	.237	.094	.069	.198	.068	.100	.211	.098
.200	.100	-.008	.200	.152	.200	.181	.181	.042	.200	.187	.071
.450	.058	-.026	.450	.083	-.019	.450	.143	.012	.450	.134	.016
.750	.028	-.050	.750	.022	-.055	.750	.065	-.030	.750	.090	-.023
.963	.024	-.061	.953			.931	.044	-.036	.901	.075	-.031

 $M = 3.22; \alpha = -5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$

.037	.160	.023	.048	.245	.115	.069	.204	.084	.100	.208	.087
.200	.103	-.001	.200	.157	.200	.183	.183	.055	.200	.193	.086
.450	.059	-.023	.450	.085	-.009	.450	.147	.024	.450	.140	.030
.750	.026	-.045	.750	.021	-.055	.750	.069	-.021	.750	.091	-.014
.963	.023	-.055	.953			.931	.047	-.030	.901	.079	-.022

 $M = 3.22; \alpha = -5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$

.037	.172	.037	.048	.258	.135	.069	.215	.101	.100	.181	.045
.200	.111	.008	.200	.168	.200	.191	.191	.068	.200	.185	.059
.450	.064	-.016	.450	.092	-.006	.450	.155	.034	.450	.146	.043
.750	.029	-.039	.750	.025	-.049	.750	.079	-.011	.750	.097	-.003
.963	.025	-.046	.953			.931	.056	-.023	.901	.086	-.013

 $M = 3.22; \alpha = -5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$

.037	.190	.068	.048	.279	.174	.069	.232	.130	.100	.170	.028
.200	.124	.025	.200	.185	.200	.206	.206	.092	.200	.162	.025
.450	.074	-.002	.450	.105	.011	.450	.173	.055	.450	.139	.027
.750	.032	-.029	.750	.033	-.041	.750	.092	.008	.750	.097	.016
.963	.027	-.032	.953			.931	.068	-.008	.901	.091	.006

 $M = 3.22; \alpha = -5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$

.037	.210	.107	.048	.306	.221	.069	.245	.099	.100	.144	.018
.200	.142	.051	.200	.210	.200	.225	.225	.120	.200	.134	.003
.450	.088	.018	.450	.121	.037	.450	.193	.080	.450	.119	.003
.750	.042	-.013	.750	.044	-.028	.750	.110	.031	.750	.083	.000
.963	.036	-.015	.953			.931	.085	.010	.901	.087	.000

 $M = 3.22; \alpha = -5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.037	.233	.148	.048	.339	.269	.069	.283	.042	.100	.143	.017
.200	.163	.085	.200	.236	.200	.200	.213	.035	.200	.126	-.006
.450	.104	.037	.450	.142	.065	.450	.115	.015	.450	.101	-.014
.750	.056	.006	.750	.060	-.013	.750	.133	.056	.750	.077	-.008
.963	.051	.019	.953			.931	.107	.035	.901	.089	.003

 $M = 3.22; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.037	.134	.226	.048	.221	.341	.069	.174	.219	.100	.074	.062
.200	.065	.165	.200	.147	.200	.200	.150	.198	.200	.061	.053
.450	.030	.073	.450	.064	.120	.450	.111	.165	.450	.046	.042
.750	.003	.029	.750	-.001	.022	.750	.066	.098	.750	.021	.049
.963	.001	.017	.953			.931	.047	.074	.901	.021	.039

TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

M = 3.22; $\alpha = 0^\circ$; $\beta = 6^\circ$; $\delta_e = 0^\circ$

.037	.115	.177	.048	.190	.289	.069	.154	.206	.100	.076	.088
.200	.048	.126	.200	.124	.200	.200	.126	.169	.200	.081	.085
.450	.019	.045	.450	.045	.086	.450	.090	.131	.450	.064	.077
.750	-.008	.009	.750			.750	.048	.072	.750	.039	.071
.963	-.011	-.003	.953	-.011	.002	.931	.030	.050	.901	.034	.061

M = 3.22; $\alpha = 0^\circ$; $\beta = 4^\circ$; $\delta_e = 0^\circ$

.037	.094	.132	.048	.161	.238	.069	.128	.167	.100	.134	.150
.200	.034	.092	.200	.102	.200	.200	.104	.134	.200	.122	.159
.450	.006	.021	.450	.028	.063	.450	.071	.104	.450	.077	.099
.750	-.018	-.009	.750			.750	.031	.046	.750	.031	.046
.963	-.021	-.020	.953	-.020	-.017	.931	.013	.026	.901	.020	.037

M = 3.22; $\alpha = 0^\circ$; $\beta = 2^\circ$; $\delta_e = 0^\circ$

.037	.080	.096	.048	.138	.193	.069	.108	.136	.100	.123	.154
.200	.025	.064	.200	.081	.200	.200	.086	.109	.200	.104	.138
.450	-.002	.004	.450	.015	.042	.450	.056	.078	.450	.058	.077
.750	-.024	-.023	.750			.750	.017	.025	.750	.018	.029
.963	-.023	-.030	.953	-.026	-.029	.931	.000	.010	.901	.009	.021

M = 3.22; $\alpha = 0^\circ$; $\beta = 1^\circ$; $\delta_e = 0^\circ$

.037	.076	.081	.048	.129	.171	.069	.100	.122	.100	.114	.138
.200	.022	.048	.200	.073	.200	.200	.080	.100	.200	.094	.122
.450	-.004	-.003	.450	.011	.030	.450	.050	.065	.450	.052	.065
.750	-.024	-.031	.750			.750	.011	.018	.750	.012	.023
.963	-.022	-.038	.953	-.027	-.034	.931	-.006	.001	.901	.003	.014

M = 3.22; $\alpha = 0^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$

.037	.073	.067	.048	.119	.152	.069	.093	.109	.100	.103	.121
.200	.018	.035	.200	.086	.200	.200	.073	.091	.200	.085	.107
.450	-.005	-.006	.450	.006	.019	.450	.046	.054	.450	.045	.054
.750	-.022	-.036	.750			.750	.006	.010	.750	.007	.016
.963	-.023	-.045	.953	-.030	-.034	.931	-.011	-.007	.901	-.002	.006

M = 3.22; $\alpha = 0^\circ$; $\beta = -1^\circ$; $\delta_e = 0^\circ$

.037	.069	.062	.048	.111	.133	.069	.087	.096	.100	.093	.105
.200	.016	.024	.200	.058	.200	.200	.067	.080	.200	.077	.092
.450	-.006	-.008	.450	.002	.008	.450	.041	.043	.450	.039	.044
.750	-.020	-.039	.750			.750	.000	.001	.750	.002	.008
.963	-.022	-.047	.953	-.030	-.036	.931	-.015	-.015	.901	-.006	-.002

M = 3.22; $\alpha = 0^\circ$; $\beta = -2^\circ$; $\delta_e = 0^\circ$

.037	.062	.052	.048	.106	.119	.069	.082	.091	.100	.085	.092
.200	.014	.015	.200	.053	.200	.200	.063	.068	.200	.070	.081
.450	-.006	-.007	.450	-.001	-.001	.450	.039	.034	.450	.035	.038
.750	-.019	-.045	.750			.750	-.004	-.006	.750	-.001	.000
.963	-.023	-.052	.953	-.029	-.038	.931	-.019	-.022	.901	-.008	-.008

M = 3.22; $\alpha = 0^\circ$; $\beta = -4^\circ$; $\delta_e = 0^\circ$

.037	.052	.035	.048	.100	.099	.069	.071	.069	.100	.076	.074
.200	.011	-.006	.200	.040	.200	.200	.055	.045	.200	.063	.063
.450	-.003	-.007	.450	-.005	-.017	.450	.035	.024	.450	.027	.022
.750	-.019	-.052	.750			.750	-.011	-.019	.750	-.005	-.014
.963	-.023	-.054	.953	-.024	-.045	.931	-.026	-.034	.901	-.012	-.018

TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
M = 3.22; $\alpha = 0^\circ$; $\beta = -6^\circ$; $\delta_e = 0^\circ$											
.037	.045	.012	.048	.088	.089	.069	.060	.043	.100	.069	.061
.200	.012	-.030	.200	.028		.200	.051	.030	.200	.056	.045
.450	.003	-.009	.450	-.006	-.037	.450	.031	.014	.450	.022	.004
.750	-.021	-.068	.750			.750	-.019	-.035	.750	-.006	-.022
.963	-.024	-.053	.953	-.022	-.053	.931	-.031	-.049	.901	-.013	-.025
M = 3.22; $\alpha = 0^\circ$; $\beta = -8^\circ$; $\delta_e = 0^\circ$											
.037	.043	-.015	.048	.062	.055	.069	.056	.024	.100	.058	.040
.200	.017	-.043	.200	.020		.200	.059	.029	.200	.044	.021
.450	.006	-.025	.450	-.004	-.047	.450	.022	-.002	.450	.021	-.005
.750	-.021	-.058	.750			.750	-.023	-.047	.750	-.005	-.025
.963	-.026	-.060	.953	-.022	-.060	.931	-.029	-.058	.901	-.014	-.031
M = 3.22; $\alpha = 5^\circ$; $\beta = -8^\circ$; $\delta_e = 0^\circ$											
.037	-.006	.079	.048	-.021	.134	.069	.001	.193	.100	-.013	.123
.200	-.038	.055	.200	-.043		.200	-.013	.149	.200	-.020	.134
.450	-.044	.018	.450	-.051	.056	.450	-.042	.071	.450	-.035	.110
.750	-.057	-.021	.750			.750	-.062	.010	.750	-.055	.048
.963	-.059	.011	.953	-.059	.002	.931	-.069	.020	.901	-.059	.031
M = 3.22; $\alpha = 5^\circ$; $\beta = -6^\circ$; $\delta_e = 0^\circ$											
.037	-.017	.084	.048	-.016	.189	.069	-.021	.160	.100	-.017	.134
.200	-.050	.061	.200	-.046		.200	-.027	.157	.200	-.023	.140
.450	-.046	.046	.450	-.063	.053	.450	-.042	.095	.450	-.046	.107
.750	-.060	.008	.750			.750	-.068	.025	.750	-.062	.064
.963	-.064	.029	.953	-.065	-.006	.931	-.070	.014	.901	-.060	.046
M = 3.22; $\alpha = 5^\circ$; $\beta = -4^\circ$; $\delta_e = 0^\circ$											
.037	-.015	.120	.048	-.011	.229	.069	-.017	.182	.100	-.013	.156
.200	-.059	.078	.200	-.043		.200	-.029	.165	.200	-.022	.156
.450	-.058	.058	.450	-.067	.058	.450	-.046	.119	.450	-.045	.118
.750	-.062	.025	.750			.750	-.067	.043	.750	-.064	.078
.963	-.066	-.008	.953	-.066	.007	.931	-.066	.026	.901	-.062	.063
M = 3.22; $\alpha = 5^\circ$; $\beta = -2^\circ$; $\delta_e = 0^\circ$											
.037	-.013	.156	.048	-.000	.259	.069	-.010	.201	.100	-.001	.191
.200	-.060	.104	.200	-.034		.200	-.023	.181	.200	-.010	.186
.450	-.065	.069	.450	-.066	.079	.450	-.043	.142	.450	-.038	.133
.750	-.067	.015	.750			.750	-.063	.065	.750	-.061	.089
.963	-.066	-.001	.953	-.070	.015	.931	-.059	.043	.901	-.060	.077
M = 3.22; $\alpha = 5^\circ$; $\beta = -1^\circ$; $\delta_e = 0^\circ$											
.037	-.011	.171	.048	-.008	.273	.069	-.003	.214	.100	.008	.195
.200	-.063	.117	.200	-.028		.200	-.018	.189	.200	-.002	.200
.450	-.068	.078	.450	-.063	.092	.450	-.039	.151	.450	-.031	.142
.750	-.070	.022	.750			.750	-.061	.074	.750	-.056	.093
.963	-.068	.012	.953	-.073	.024	.931	-.056	.051	.901	-.056	.082
M = 3.22; $\alpha = 5^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$											
.037	-.006	.183	.048	-.018	.283	.069	-.008	.229	.100	.018	.200
.200	-.061	.127	.200	-.020		.200	-.009	.198	.200	.006	.200
.450	-.068	.078	.450	-.058	.102	.450	-.033	.157	.450	-.025	.152
.750	-.070	.028	.750			.750	-.054	.081	.750	-.050	.098
.963	-.067	.007	.953	-.071	.032	.931	-.049	.060	.901	-.052	.086

TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Continued

Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
$M = 3.22; \alpha = 5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$											
.037	-.004	.197	.048	.022	.293	.069	-.013	.239	.100	.025	.210
.200	-.062	.140	.200	-.014	.200	.200	-.003	.208	.200	.011	.204
.450	-.069	.079	.450	-.055	.113	.450	-.030	.164	.450	-.022	.160
.750	-.072	.031	.750			.750	-.053	.088	.750	-.048	.103
.963	-.068	.013	.953	-.073	.035	.931	-.047	.066	.901	-.050	.091
$M = 3.22; \alpha = 5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$											
.037	.001	.211	.048	.031	.301	.069	.020	.248	.100	.034	.223
.200	-.057	.155	.200	-.006	.200	.200	.006	.221	.200	.019	.213
.450	-.067	.084	.450	-.050	.127	.450	-.023	.170	.450	-.015	.165
.750	-.073	.031	.750			.750	-.047	.097	.750	-.044	.109
.963	-.069	.018	.953	-.070	.040	.931	-.040	.075	.901	-.045	.096
$M = 3.22; \alpha = 5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$											
.037	.011	.241	.048	.049	.343	.069	.034	.260	.100	.041	.249
.200	-.048	.186	.200	.008	.200	.200	.018	.254	.200	.030	.238
.450	-.062	.105	.450	-.040	.155	.450	-.013	.196	.450	-.002	.177
.750	-.070	.040	.750			.750	-.038	.115	.750	-.034	.125
.963	-.068	.033	.953	-.066	.058	.931	-.031	.091	.901	-.034	.114
$M = 3.22; \alpha = 5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$											
.037	.025	.278	.048	.074	.392	.069	.055	.294	.100	.054	.254
.200	-.034	.218	.200	.028	.200	.200	.034	.277	.200	.042	.254
.450	-.050	.125	.450	-.026	.184	.450	.002	.221	.450	.011	.199
.750	-.062	.057	.750			.750	-.026	.138	.750	-.021	.151
.963	-.060	.048	.953	-.058	.073	.931	-.017	.119	.901	-.028	.140
$M = 3.22; \alpha = 5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$											
.037	.040	.332	.048	.101	.458	.069	.075	.343	.100	.016	.160
.200	-.020	.260	.200	.047	.200	.200	.050	.313	.200	.012	.186
.450	-.040	.154	.450	-.013	.220	.450	.018	.268	.450	-.003	.164
.750	-.052	.080	.750			.750	-.012	.165	.750	-.022	.157
.963	-.050	.071	.953	-.051	.100	.931	-.006	.142	.901	-.022	.166
$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	.220	-.012	.048	.312	.052	.069	.301	.043	.100	.258	-.011
.200	.153	-.023	.200	.224	.200	.200	.296	.020	.200	.259	.002
.450	.104	-.018	.450	.145	-.042	.450	.210	-.004	.450	.220	-.006
.750	.067	-.052	.750			.750	.123	-.046	.750	.160	-.032
.963	.062	-.062	.953	.063	-.055	.931	.100	-.045	.901	.134	-.035
$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	.170	.001	.048	.271	.072	.069	.235	.057	.100	.240	.066
.200	.115	-.006	.200	.179	.200	.200	.217	.030	.200	.221	.061
.450	.075	-.021	.450	.106	-.034	.450	.167	-.001	.450	.165	.010
.750	.043	-.051	.750			.750	.086	-.039	.750	.116	-.028
.963	.038	-.061	.953	.036	-.052	.931	.063	-.044	.901	.099	-.036
$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	.137	.016	.048	.221	.091	.069	.179	.074	.100	.192	.106
.200	.065	.007	.200	.138	.200	.200	.155	.045	.200	.173	.077
.450	.045	-.016	.450	.062	-.011	.450	.124	.009	.450	.119	.023
.750	.018	-.044	.750			.750	.055	-.025	.750	.072	-.017
.963	.014	-.053	.953	.011	-.049	.931	.032	-.033	.901	.060	-.028

TABLE II. - PRESSURE COEFFICIENTS ON MISSILE WING MEASURED IN PRESENCE OF BOOSTER - Concluded

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.037	.090	.045	.048	.169	.110	.069	.133	.095	.100	.144	.124
.200	.026	.032	.200	.108	.032	.200	.108	.065	.200	.124	.098
.450	.013	-.011	.450	.033	.007	.450	.077	.027	.450	.078	.042
.750	-.004	-.039	.750			.750	.030	-.005	.750	.031	-.004
.963	-.006	-.050	.953	-.014	-.044	.931	.010	-.017	.901	.021	-.014

$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.037	.068	.092	.048	.106	.143	.069	.094	.119	.100	.105	.133
.200	.010	.053	.200	.071		.200	.073	.090	.200	.086	.118
.450	-.014	.004	.450	.009	.031	.450	.039	.056	.450	.045	.062
.750	-.026	-.026	.750			.750	.006	.016	.750	.006	.017
.963	-.022	-.038	.953	-.034	-.026	.931	-.010	.001	.901	-.005	.007

$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.037	.034	.126	.048	.066	.188	.069	.055	.148	.100	.067	.155
.200	-.015	.084	.200	.030		.200	.039	.136	.200	.054	.143
.450	-.035	.033	.450	-.017	.061	.450	.009	.090	.450	.015	.087
.750	-.045	-.008	.750			.750	-.020	.042	.750	-.016	.047
.963	-.042	-.023	.953	-.050	-.002	.931	-.028	.023	.901	-.025	.035

$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.037	.014	.161	.048	.040	.241	.069	.030	.195	.100	.041	.181
.200	-.036	.116	.200	.005		.200	.016	.173	.200	.028	.171
.450	-.050	.058	.450	-.035	.090	.450	-.012	.128	.450	-.006	.122
.750	-.056	.015	.750			.750	-.035	.064	.750	-.034	.075
.963	-.055	-.006	.953	-.059	.018	.931	-.037	.043	.901	-.039	.062

TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE

Body station $\frac{x_B}{L_B}$	C_p at $\theta =$				
	0°	60°	90°	120°	180°
$M = 2.29; \alpha = -80^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.372		.133		.038
.059	.208		.033		-.020
.099	.209	.088		-.027	.002
.237		-.041		-.082	-.052
.296	-.009	.041		-.066	-.040
.434	.094	-.007			
.519		.049			-.014
.592		-.048		-.050	-.012
.714				-.064	-.008
.765		-.082		-.112	-.021
.897					-.066
.957	-.128			-.040	.018
.992	-.112	-.076		-.082	-.072
$M = 2.29; \alpha = -40^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.283		.144		.062
.059	.141		.043		-.004
.099	.144	.078		.008	.009
.237		-.017		-.048	-.040
.296	-.023	-.044			-.023
.434	.078	-.014		-.031	
.519		.043			-.019
.592		-.045		-.033	-.007
.714				-.038	-.005
.765		-.017		-.104	-.012
.897					-.071
.957	-.142			.014	-.014
.992	-.104	-.100		-.048	-.073
$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.201		.151		.115
.059	.082		.053		.026
.099	.089	.069		.043	.037
.237		.028		-.020	-.032
.296	-.030	-.007			-.017
.434	.052	-.001		.008	
.519		.026			-.014
.592		-.016		.001	-.007
.714				-.021	.004
.765		.051		-.071	-.008
.897					-.033
.957	-.093			.070	-.018
.992	-.101	-.060		-.014	-.043
$M = 2.29; \alpha = 40^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.137		.145		.178
.059	.039		.049		.065
.099	.049	.053		.065	.072
.237		.079		-.012	-.011
.296	-.037	.047			.000
.434	.011	.012		.030	
.519		.020			.002
.592		.020		.020	.002
.714				.001	.014
.765		.131		-.039	.008
.897				-.015	-.015
.957	-.043			.109	.020
.992	-.076	-.027		.009	-.005
$M = 2.29; \alpha = -50^\circ; \beta = -80^\circ; \delta_e = 0^\circ$					
.043	.314		.050		.037
.059	.161		-.028		-.030
.099	.146	-.007		-.019	-.022
.237		-.097		-.060	-.063
.296	.129	-.105			-.046
.434	.018	-.097		-.014	
.519		-.076			-.056
.592		-.204		-.031	-.057
.714				-.057	-.076
.765		-.112		-.098	-.084
.897					-.090
.957	-.163			-.095	-.111
.992	-.137	-.127		-.120	-.158
$M = 2.29; \alpha = -60^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.326		.135		.045
.059	.173		.036		-.015
.099	.175	.082		-.010	.007
.237		-.033		-.072	-.047
.296	-.019	.026			-.031
.434	.090	-.018		-.053	
.519		.051			-.021
.592		-.049		-.047	-.010
.714				-.052	-.007
.765		-.056		-.116	-.014
.897					-.071
.957	-.135			-.023	-.013
.992	-.105	-.107		-.063	-.066
$M = 2.29; \alpha = -20^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.241		.148		.084
.059	.110		.049		.008
.099	.113	.073		.027	.023
.237		.006		-.034	-.043
.296	-.027	-.029			-.016
.434	.067	-.011		-.009	
.519		.039			-.016
.592		-.032		-.017	-.007
.714				-.026	-.001
.765		.011		-.085	-.013
.897					-.047
.957	-.119			.042	-.016
.992	-.105	-.087		-.025	-.060
$M = 2.29; \alpha = 20^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.169		.149		.144
.059	.058		.052		.042
.099	.066	.062		.056	.053
.237		.053		-.014	-.025
.296	-.033	.020			-.007
.434	.024	.004		.016	
.519		.021			-.007
.592		-.002		.014	-.007
.714				-.013	.009
.765		.095		-.057	-.002
.897					-.029
.957	-.071			.089	-.003
.992	-.091	-.041		-.003	-.026
$M = 2.29; \alpha = -50^\circ; \beta = -40^\circ; \delta_e = 0^\circ$					
.043	.316		.028		.025
.059	.162		-.038		-.044
.099	.140	-.029		-.025	-.044
.237		-.109		-.058	-.074
.296	.160	-.100			-.060
.434	.023	-.113		-.027	
.519		-.049			-.075
.592		-.174		-.029	-.083
.714				-.080	-.103
.765		-.062		-.045	-.121
.897					-.122
.957	-.145			-.087	-.116
.992	-.132	-.103		-.123	-.135

TABLE III.- PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x_B}{l_B}$	C_p at $\theta = \gamma$				
	0°	60°	90°	120°	180°
$M = 2.29; \alpha = -5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$					
.043	.308		.110		.050
.059	.159		.016		-.012
.099	.157	.053		-.005	-.000
.237		-.050		-.059	-.046
.296	-.011	-.069			-.031
.434	.073	-.042		-.028	
.519		-.005			-.027
.592		-.117		-.027	-.019
.714				-.049	-.028
.765		-.031		-.107	-.036
.897					-.087
.957	-.152			-.044	-.049
.992	-.108	-.086		-.091	-.091
$M = 2.29; \alpha = -5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.302		.140		.051
.059	.153		.037		-.012
.099	.156	.077		-.002	.007
.237		-.025		-.067	-.046
.296	-.022	-.048			-.030
.434	.082	-.018		-.042	
.519		.099			.035
.592		.008		.014	.047
.714				.011	.051
.765		.028		-.055	.044
.897					.019
.957	-.093			.055	.037
.992	-.055	-.056		.001	-.024
$M = 2.29; \alpha = -5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$					
.043	.301		.174		.055
.059	.154		.065		-.013
.099	.161	.109		.002	.004
.237		.000		-.062	-.042
.296	.007	.034			-.027
.434	.068	.016		-.052	
.519		.060			-.028
.592		-.027		-.069	-.018
.714				-.053	-.025
.765		-.036		-.121	-.035
.897					-.091
.957	-.147			.047	-.061
.992	-.110	-.112		-.020	-.097
$M = 2.29; \alpha = -5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$					
.043	.279		.250		.040
.059	.136		.122		-.030
.099	.150	.167		.017	-.018
.237		.053		-.055	-.062
.296	.136	.104			-.047
.434	.018	.085		-.079	
.519		.227		.002	.002
.592		.025		-.067	-.002
.714				-.074	-.011
.765		.097		-.086	-.024
.897					-.034
.957	-.109			.223	-.053
.992	-.092	-.056		.153	-.094
$M = 2.29; \alpha = 0^\circ; \beta = 6^\circ; \delta_e = 0^\circ$					
.043	.158		.302		.083
.059	.046		.163		-.009
.099	.062	.166		.096	-.015
.237		.132		.013	-.075
.296	.153	.045			-.066
.434	-.003	.094		.003	
.519		.139			-.103
.592		-.036		-.073	-.100
.714				-.108	-.086
.765		.131		-.161	-.102
.897					-.124
.957	-.162			.203	-.140
.992	-.151	-.073		.145	-.153

Body station $\frac{x_B}{l_B}$	C_p at $\theta = \gamma$				
	0°	60°	90°	120°	180°
$M = 2.29; \alpha = -5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$					
.043	.307		.126		.052
.059	.158		.029		-.010
.099	.158	.066		-.002	.003
.237		-.038		-.057	-.044
.296	-.022	-.065			-.029
.434	.081	-.030		-.036	
.519		.026			-.023
.592		-.073		-.034	-.010
.714				-.048	-.013
.765		-.027		-.111	-.020
.897					-.078
.957	-.160			-.026	-.028
.992	-.108	-.092		-.076	-.082
$M = 2.29; \alpha = -5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$					
.043	.290		.147		.044
.059	.143		.040		-.021
.099	.149	.083		-.009	-.003
.237		-.022		-.072	-.055
.296	-.029	-.017			-.034
.434	.070	-.010		-.054	
.519		.055			-.031
.592		-.043		-.061	-.019
.714				-.053	-.019
.765		-.058		-.124	-.025
.897					-.087
.957	-.162			.008	-.038
.992	-.115	-.117		-.063	-.092
$M = 2.29; \alpha = -5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$					
.043	.289		.203		.045
.059	.144		.083		-.022
.099	.154	.131		.003	-.007
.237		.018		-.063	-.053
.296	.093	.049			-.038
.434	.031	.046		-.067	
.519		.155			.018
.592		.007		-.040	.024
.714				-.021	.005
.765		.056		-.070	-.006
.897					-.017
.957	-.104			.115	-.027
.992	-.074	-.058		.086	-.063
$M = 2.29; \alpha = -5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$					
.043	.272		.305		.026
.059	.131		.167		-.048
.099	.146	.210		.037	-.033
.237		.092		-.041	-.073
.296	.188	.168			-.063
.434	.025	.122		-.065	
.519		.226			-.092
.592		-.022		-.137	-.099
.714				-.159	-.120
.765		.076		-.167	-.140
.897					-.146
.957	-.176			.249	-.157
.992	-.161	-.109		.108	-.159
$M = 2.29; \alpha = 0^\circ; \beta = 6^\circ; \delta_e = 0^\circ$					
.043	.169		.257		.094
.059	.058		.132		.002
.099	.071	.138		.079	-.002
.237		.101		.002	-.060
.296	.128	.055			-.046
.434	-.004	.076		-.003	
.519		.053			-.089
.592		-.079		-.078	-.088
.714				-.104	-.076
.765		.092		-.160	-.092
.897					-.103
.957	-.166			.100	-.104
.992	-.156	-.082		.034	-.132

TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x_B}{L_B}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 2.29; \alpha = 0^\circ; \beta = 4^\circ; \delta_e = 0^\circ$					
.043	.187		.227		.107
.059	.069			.068	.013
.099	.082	.118	.107	.019	
.237		.080		-.007	-.048
.296	.030	.037		-.004	-.029
.434	-.006	.060			-.070
.519		.007		-.071	-.070
.592		-.098		-.091	-.060
.714				-.147	-.075
.765		.055			-.079
.897				.087	-.087
.957	-.165			-.013	-.116
.992	-.148	-.091			
$M = 2.29; \alpha = 0^\circ; \beta = 1^\circ; \delta_e = 0^\circ$					
.043	.193		.166		.167
.059	.073		.061		.017
.099	.084	.078		.044	.027
.237		.038		-.022	-.041
.296	-.038	.007			-.021
.434	.027	.008		-.005	
.519		.042			.029
.592		.007		.038	.028
.714				.012	.044
.765		.092		-.043	.027
.897					.009
.957	-.060			.127	.013
.992	-.058	-.025		.030	-.011
$M = 2.29; \alpha = 0^\circ; \beta = -1^\circ; \delta_e = 0^\circ$					
.043	.203		.198		.109
.059	.091		.042		.020
.099	.089	.061		.036	.032
.237		.016		-.024	-.037
.296	-.030	-.022			-.016
.434	.046	-.016		.003	
.519		.063			.021
.592		-.001		.035	.021
.714				.008	.036
.765		.060		-.042	.021
.897					-.001
.957	-.064			.071	.010
.992	-.065	-.036		.005	-.015
$M = 2.29; \alpha = 0^\circ; \beta = -4^\circ; \delta_e = 0^\circ$					
.043	.200		.092		.099
.059	.079		.012		.009
.099	.079	.029		.018	.020
.237		-.017		-.032	-.044
.296	-.028	-.053			-.028
.434	-.012	-.051		.002	
.519		.034			-.003
.592		-.048		.025	-.003
.714				-.013	.009
.765		-.003		-.043	-.006
.897					-.010
.957	-.093			.000	-.022
.992	-.080	-.071		-.048	-.046
$M = 2.29; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.196		.037		.066
.059	.070		-.024		-.022
.099	.054	-.017		-.007	-.013
.237		-.063		-.046	-.075
.296	.133	-.079			-.066
.434	-.013	-.092		-.009	
.519		-.018			-.066
.592		-.151		.003	-.061
.714				-.050	-.049
.765		-.053		-.060	-.080
.897				-.077	-.103
.957	-.121			-.117	-.111
.992	-.117	-.120			

Body station $\frac{x_B}{L_B}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 2.29; \alpha = 0^\circ; \beta = 2^\circ; \delta_e = 0^\circ$					
.043	.199		.189		.114
.059	.078		.077		.020
.099	.089	.093		.055	.030
.237		.054		-.015	-.040
.296	-.034	.015			-.019
.434	.026	.026		-.002	
.519		.037			.018
.592		-.013		.025	.019
.714				-.001	.033
.765		.092		-.057	.015
.897					.001
.957	-.078			.140	-.002
.992	-.069	-.027		.027	-.026
$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.201		.154		.111
.059	.082		.055		.022
.099	.090	.073		.043	.033
.237		.031		-.020	-.035
.296	-.029	-.006			-.015
.434	.048	-.001		.001	
.519		.048			.010
.592		-.006		.020	.010
.714				-.004	.024
.765		.061		-.049	.010
.897					.013
.957	-.076			.084	.001
.992	-.076	-.044		.005	-.026
$M = 2.29; \alpha = 0^\circ; \beta = -2^\circ; \delta_e = 0^\circ$					
.043	.200		.118		.104
.059	.079		.028		.015
.099	.084	.047		.027	.027
.237		.002		-.030	-.041
.296	-.037	-.040			-.021
.434	.025	-.031		.002	
.519		.064			.020
.592		-.019		.039	.019
.714				.007	.034
.765		.056		-.041	.018
.897					.002
.957	-.068			.047	.001
.992	-.063	-.034		-.011	-.021
$M = 2.29; \alpha = 0^\circ; \beta = -6^\circ; \delta_e = 0^\circ$					
.043	.210		.071		.093
.059	.084		-.001		.022
.099	.075	.013		.013	.014
.237		-.033		-.031	-.051
.296	.068	-.061			-.037
.434	-.006	-.069		.000	
.519		.003			-.059
.592		-.150		-.008	-.056
.714				-.047	-.045
.765		-.063		-.052	-.059
.897					-.063
.957	-.139			-.067	-.077
.992	-.125	-.131		-.110	-.098
$M = 2.29; \alpha = 5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.127		.049		.156
.059	.019		-.002		.041
.099	.008	.010		.005	.055
.237		.019		-.050	-.028
.296	.122	.006			-.027
.434	-.009	-.049		.008	
.519		-.056			-.066
.592		-.115		-.034	-.072
.714				-.079	-.087
.765		-.024		-.093	-.098
.897					-.091
.957	-.102			-.086	-.070
.992	-.101	-.114		-.118	-.068

TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x_B}{l_B}$	C_p at $\theta =$				
	0°	60°	90°	120°	180°
$M = 2.29; \alpha = 5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.129		.065		.169
.059	.025		.004		.054
.099	.022	.018		.017	.068
.237		.036		-.045	-.016
.296	.056	.019			-.011
.434	-.021	-.035	.015		
.519		.016		-.034	
.592		-.077		-.009	-.036
.714				-.041	-.044
.765		-.006		-.071	-.048
.897				-.043	
.957	-.084			-.050	-.026
.992	-.092	-.103		-.086	-.025
$M = 2.29; \alpha = 5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$					
.043	.129		.115		.191
.059	.031		.029		.075
.099	.041	.038		.052	.086
.237		.080		-.022	-.001
.296	-.041	.048			.008
.434	.008	.001	.029		
.519		.044		-.009	
.592		-.010	.000	-.014	-.001
.714			-.019	-.001	
.765		.076	-.044	-.005	
.897				-.022	
.957	-.050		.049	.016	
.992	-.085	-.067	-.034	-.003	
$M = 2.29; \alpha = 5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.123		.144		.197
.059	.030		.046		.079
.099	.041	.048		.070	.086
.237		.093		-.010	-.001
.296	-.040	.062			.007
.434	.010	.020	.032		
.519		-.015		-.022	-.023
.592		-.002	-.013	-.031	
.714			-.027	-.016	
.765		.104	-.060	-.017	
.897				-.045	
.957	-.062		.095	-.000	
.992	-.102	-.048	-.016	-.022	
$M = 2.29; \alpha = 5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$					
.043	.113		.176		.196
.059	.022		.069		.079
.099	.035	.058		.090	.081
.237		.112		.001	-.006
.296	-.035	.078			.001
.434	.000	.039	.043		
.519		-.023		-.005	-.023
.592		-.031	-.005	-.031	
.714			-.019	-.022	
.765		.146	-.063	-.028	
.897				-.037	
.957	-.076		.159	-.009	
.992	-.104	-.030	.023	-.023	
$M = 2.29; \alpha = 5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$					
.043	.100		.252		.199
.059	.009		.126		.074
.099	.022	.099		.139	.063
.237		.183		.038	-.019
.296	.092	.122			-.015
.434	-.020	.081	.070		
.519		.017		-.085	
.592		-.057	-.020	-.091	
.714			-.053	-.102	
.765		.220	-.087	-.103	
.897				-.099	
.957	-.138		.177	-.076	
.992	-.143	-.044	.053	-.083	

Body station $\frac{x_B}{l_B}$	C_p at $\theta =$				
	0°	60°	90°	120°	180°
$M = 2.29; \alpha = 5^\circ; \beta = -4^\circ; \delta_e = 0^\circ$					
.043	.130		.089		.181
.059	.029		.016		.066
.099	.034	.029		.034	.079
.237		.058		-.034	-.005
.296	-.017	.034			.001
.434	-.036	-.016	.022		
.519		.036		-.009	-.027
.592		-.032		-.035	-.023
.714				-.062	-.026
.765		.020		-.027	
.897				-.015	-.008
.957	-.073			-.070	-.014
.992	-.094	-.095			
$M = 2.29; \alpha = 5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$					
.043	.123		.126		.192
.059	.029		.035		.075
.099	.039	.040		.058	.085
.237		.079		-.018	-.003
.296	-.047	.051			.005
.434	.008	.010	.029		
.519		.016		-.023	
.592		-.019	-.014	-.029	
.714			-.030	-.014	
.765		.079	-.063	-.017	
.897				-.043	
.957	-.062		.059	.002	
.992	-.101	-.070	-.032	-.020	
$M = 2.29; \alpha = 5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$					
.043	.117		.160		.198
.059	.026		.057		.080
.099	.038	.051		.079	.084
.237		.106		-.006	-.004
.296	-.045	.071			.006
.434	.007	.030	.036		
.519		-.023		-.018	
.592		-.005	-.007	-.027	
.714			-.019	-.014	
.765		.136	-.058	-.018	
.897				-.038	
.957	-.064		.131	.002	
.992	-.100	-.031	.010	-.016	
$M = 2.29; \alpha = 5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$					
.043	.102		.209		.190
.059	.012		.093		.074
.099	.027	.074		.112	.071
.237		.139		.017	-.015
.296	.013	.098			-.008
.434	-.037	.061	.053		
.519		.001		-.048	
.592		-.045	-.004	-.047	
.714			-.027	-.050	
.765		.184	-.074	-.054	
.897				-.051	
.957	-.105		.199	-.037	
.992	-.116	-.028	.043	-.041	
$M = 2.29; \alpha = 5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$					
.043	.084		.297		.179
.059	-.007		.161		.064
.099	.003	.122		.164	.044
.237		.212		.058	-.038
.296	.134	.147			-.035
.434	-.008	.093	.084		
.519		.094		-.110	
.592		-.013	-.012	-.119	
.714			-.048	-.137	
.765		.261	-.106	-.145	
.897				-.135	
.957	-.148		.176	-.104	
.992	-.140	-.047	.116	-.119	

TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x_B}{L_B}$	C_p at $\theta =$ —				
	0°	60°	90°	120°	180°
$M = 2.29; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.366		.129		.036
.059	.202		.030		-.023
.099	.203	.084		-.029	.002
.237		-.057		-.087	-.051
.296	.028	.047			-.042
.434	.088	-.005		-.064	
.519		.064			-.021
.592		-.045		-.059	-.015
.714				-.065	-.013
.765		-.068		-.117	-.033
.897					-.068
.957	-.128			-.032	.002
.992	-.116	-.071		-.077	-.072
$M = 2.29; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.280		.141		.058
.059	.139		.041		-.007
.099	.141	.076		.006	.010
.237		-.041		-.056	-.047
.296	-.025	-.063			-.024
.434	.085	-.011		-.029	
.519		.045		-.040	-.018
.592		-.035		-.039	-.005
.714				-.011	-.001
.765		-.020		-.106	-.014
.897					-.043
.957	-.147			.017	-.023
.992	-.108	-.107		-.050	-.061
$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.203		.150		.112
.059	.082		.051		.023
.099	.088	.068		.041	.035
.237		-.003		-.022	-.034
.296	-.029	-.024			-.020
.434	.068	.005		.011	
.519		.037		-.017	-.017
.592		-.016		-.003	-.008
.714				-.023	.003
.765		.052		-.075	-.010
.897				-.029	-.029
.957	-.098			.070	-.023
.992	-.103	-.066		-.014	-.048
$M = 2.75; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.362		.133		.033
.059	.201		.040		-.015
.099	.203	.094		-.021	-.001
.237		.001		-.062	-.040
.296	.087	-.028			-.030
.434	.088	-.029		-.049	
.519		-.017			-.025
.592		-.064		-.058	-.019
.714				-.060	-.011
.765		-.070		-.090	-.012
.897					-.042
.957	-.108			-.053	-.023
.992	-.097	-.102		-.055	-.005
$M = 2.75; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.275		.141		.053
.059	.140		.048		-.006
.099	.142	.081		.006	.010
.237		.017		-.037	-.032
.296	.093	-.014			-.023
.434	.040	-.023		-.033	
.519		-.021			-.014
.592		-.065		-.026	-.005
.714				-.038	.002
.765		-.059		-.082	-.001
.897					-.043
.957	-.114			.014	-.009
.992	-.094	-.096		-.041	-.030

Body station $\frac{x_B}{L_B}$	C_p at $\theta =$ —				
	0°	60°	90°	120°	180°
$M = 2.29; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.326		.135		.041
.059	.173		.034		-.018
.099	.175	.081		-.015	.003
.237		-.055		-.077	-.047
.296	-.014	-.049			-.034
.434	.090	-.011		-.054	
.519		.046			-.021
.592		-.043		-.057	-.010
.714				-.053	-.005
.765		-.057		-.120	-.019
.897					-.076
.957	-.142			-.021	-.026
.992	-.111	-.108		-.070	-.082
$M = 2.29; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.241		.148		.083
.059	.110		.048		.007
.099	.113	.072		.026	.022
.237		-.022		-.035	-.041
.296	-.027	-.046			-.022
.434	.077	-.002		-.007	
.519		.043			-.019
.592		-.029		-.022	-.008
.714				-.029	-.001
.765		.012		-.088	-.014
.897					-.043
.957	-.126			.042	-.023
.992	-.108	-.095		-.029	-.061
$M = 2.29; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.168		.148		.142
.059	.059		.051		.041
.099	.066	.061		.055	.051
.237		.016		-.015	-.026
.296	-.031	-.002			-.011
.434	.055	.012		.022	
.519		.035			-.012
.592		.003		.012	-.007
.714				-.012	.010
.765		.095		-.058	.002
.897					-.024
.957	-.073			.091	-.006
.992	-.093	-.043		-.004	-.029
$M = 2.75; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.316		.136		.036
.059	.169		.042		-.015
.099	.171	.086		-.013	.001
.237		.007		-.054	-.040
.296	.080	-.027			-.030
.434	.048	.023		-.047	
.519		-.022			-.018
.592		-.066		-.039	-.008
.714				-.057	-.005
.765		-.075		-.091	-.004
.897					-.048
.957	-.118			-.036	-.036
.992	-.097	-.106		-.060	-.028
$M = 2.75; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.230		.144		.075
.059	.107		.051		.007
.099	.111	.074		.025	.018
.237		.028		-.027	-.026
.296	.100	.003			-.021
.434	.027	.025		-.009	
.519		-.021			-.012
.592		-.061		-.014	-.001
.714				-.024	.002
.765		-.031		-.066	-.004
.897					-.039
.957	-.099			.045	-.007
.992	-.096	-.081		-.008	-.034

TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x}{L}$ $\frac{y}{B}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.193		.145		.100
.059	.083		.052		.021
.099	.087				.031
.237		.068		.039	
.296		.027		-.018	
.434	.102	.023		-.021	
.519	.019	.031		-.016	
.592		-.019		-.009	
.714		-.050		-.006	
.765		.003		-.016	
.897				-.005	
.957	-.081			-.002	
.992	-.088	-.066		.057	-.032
				.007	-.016
					-.033
$M = 2.75; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.128		.140		.165
.059	.039		.050		.062
.099	.047				.068
.237		.051		.062	
.296		.001		-.006	
.434	.106	.072		-.003	
.519	.033	.043		.002	
.592		-.014		.046	
.714		-.014		.013	
.765				.000	
.897		.058		.010	
.957	-.028			-.023	
.992	-.054	-.016		-.012	
				.092	.012
				.028	.005
$M = 2.75; \alpha = -5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.311		.051		.034
.059	.163		-.017		-.023
.099	.148				-.020
.237		.004		-.009	
.296		-.055		-.044	
.434	.054	-.085		-.054	
.519	.025	-.079		-.047	
.592		-.089		-.015	
.714		-.123		-.051	
.765		-.057		-.024	
.897				-.068	
.957	-.120			-.056	
.992	-.110	-.097		-.090	
				-.083	
				-.073	
				-.101	
$M = 2.75; \alpha = -5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$					
.043	.302		.105		.045
.059	.158		.018		-.011
.099	.153				.003
.237		.053		-.006	
.296		-.015		-.046	
.434	.080	-.054		-.041	
.519	.035	-.022		-.029	
.592		-.057		-.031	
.714		-.070		-.025	
.765				-.046	
.897		-.049		-.085	
.957	-.127			-.071	
.992	-.099	-.105		-.033	
				-.008	
				-.064	
				-.070	
$M = 2.75; \alpha = -5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.295		.138		.046
.059	.153		.044		-.010
.099	.155				.006
.237		.082		.000	
.296		.012		-.045	
.434	.088	-.023		-.035	
.519	.043	-.020		-.025	
.592		-.028		-.032	
.714		-.071		-.020	
.765		-.035		-.009	
.897		-.071		-.048	
.957	-.123			-.087	
.992	-.095	-.105		-.052	
				-.012	
				-.054	
				-.032	

Body station $\frac{x}{L}$ $\frac{y}{B}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 2.75; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.155		.142		.135
.059	.055		.052		.043
.099	.062				.049
.237		.059		.053	
.296		.018		-.011	
.434	.108	.055		-.014	
.519	.021	.041		-.007	
.592		-.018		.028	
.714		-.030		.001	
.765				.007	
.897		.036		-.003	
.957	-.049			-.037	
.992	-.067	-.038		.075	-.021
				.019	-.005
					-.014
$M = 2.75; \alpha = -5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.310		.039		.032
.059	.160		-.018		-.024
.099	.140				-.027
.237		-.006		-.009	
.296		-.056		-.040	
.434	.106	.072		-.059	
.519	.033	.043		-.051	
.592		-.014		-.019	
.714		-.014		-.061	
.765				-.071	
.897		.058		-.071	
.957	-.028			-.066	
.992	-.054	-.016		-.069	
				-.080	
				-.068	
				-.073	
				-.081	
$M = 2.75; \alpha = -5^\circ; \beta = -4^\circ; \delta_e = 0^\circ$					
.043	.308		.075		.040
.059	.162		-.003		-.017
.099	.152				-.007
.237		.027		-.010	
.296		-.038		-.046	
.434	.059	-.076		-.036	
.519	.023	-.056		-.010	
.592		-.065		-.040	
.714		-.126		-.026	
.765		-.050		-.042	
.897				-.047	
.957	-.125			-.071	
.992	-.110	-.103		-.072	
				-.094	
				-.063	
				-.103	
$M = 2.75; \alpha = -5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$					
.043	.295		.121		.045
.059	.153		.031		-.010
.099	.152				.005
.237		.068		-.003	
.296		-.001		-.046	
.434	.088	-.038		-.038	
.519	.042	-.002		-.027	
.592		-.044		-.024	
.714		-.073		-.023	
.765				-.025	
.897		-.063		-.047	
.957	-.124			-.089	
.992	-.093	-.101		-.015	
				.013	
				-.061	
				-.037	
$M = 2.75; \alpha = -5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$					
.043	.293		.158		.051
.059	.153		.058		-.007
.099	.156				.008
.237		.098		.006	
.296		.025		-.040	
.434	.084	-.009		-.035	
.519	.042	-.036		-.024	
.592		-.008		-.036	
.714		-.060		-.017	
.765				-.044	
.897		-.042		-.009	
.957	-.124			-.004	
.992	-.093	-.099		-.088	
				-.009	
				-.051	
				-.012	
				-.035	

TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x_B}{l_B}$	C_p at $\theta =$ —				
	0°	60°	90°	120°	180°
$M = 2.75; \alpha = -5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$					
.043	.285		.173		-.049
.059	.147		.070		-.011
.099	.153	.110		.008	.004
.237		.035		-.041	-.039
.296	.082	.001			-.026
.434	.031	.049		-.046	
.519		-.001			-.027
.592		-.053		-.064	-.022
.714				-.055	-.022
.765		-.009		-.096	-.028
.897					-.066
.957	-.130			-.019	-.014
.992	-.098	-.099		.025	-.064
$M = 2.75; \alpha = -5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$					
.043	.275		.258		.041
.059	.140		.133		-.022
.099	.151	.172		.030	-.019
.237		.095		-.026	-.050
.296	.078	.057			-.040
.434	.037	.120		-.042	
.519		.049			-.043
.592		.001		-.080	-.049
.714				-.092	-.072
.765		.116		-.105	-.077
.897					-.085
.957	-.129			.140	-.077
.992	-.115	-.079		.111	-.100
$M = 2.75; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$					
.043	.162		.297		.082
.059	.058		.163		.001
.099	.072	.166		.098	-.008
.237		.129		.024	-.053
.296	.107	.136			-.055
.434	.032	.129		.025	
.519		.058			-.057
.592		.023		-.019	-.051
.714				-.049	-.053
.765		.224		-.093	-.063
.897					-.057
.957	-.114			.236	-.089
.992	-.108	-.018		.177	-.094
$M = 2.75; \alpha = 0^\circ; \beta = 4^\circ; \delta_e = 0^\circ$					
.043	.182		.219		.095
.059	.073		.107		.012
.099	.083	.116		.066	.015
.237		.076		.000	-.032
.296	.087	.082			-.029
.434	.010	.081		.010	
.519		.011			-.024
.592		-.012		-.024	-.018
.714				-.045	-.014
.765		.139		-.085	-.024
.897					-.040
.957	-.114			.106	-.042
.992	-.112	-.037		.068	-.063
$M = 2.75; \alpha = 0^\circ; \beta = 1^\circ; \delta_e = 0^\circ$					
.043	.191		.164		.098
.059	.080		.067		.018
.099	.086	.080		.044	.025
.237		.041		-.013	-.024
.296	.096	.043			-.019
.434	.016	.043		.003	
.519		-.013			-.012
.592		-.041		-.013	-.003
.714				-.024	.001
.765		.029		-.063	-.008
.897					-.037
.957	-.089			.075	-.030
.992	-.096	-.059		.011	-.047

Body station $\frac{x_B}{l_B}$	C_p at $\theta =$ —				
	0°	60°	90°	120°	180°
$M = 2.75; \alpha = -5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$					
.043	.280		.214		.046
.059	.143		.100		-.018
.099	.152	.140		.017	-.005
.237		.064		-.036	-.044
.296	.095	.028			-.032
.434	.028	.086		-.049	
.519		.019			-.036
.592		-.033		-.086	-.035
.714				-.070	-.050
.765		.070		-.103	-.057
.897					-.093
.957	-.129			-.003	-.070
.992	-.110	-.088		.082	-.088
$M = 2.75; \alpha = -5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$					
.043	.259		.298		.027
.059	.127		.163		-.035
.099	.142	.200		.044	-.036
.237		.125		-.016	-.065
.296	.099	.086			-.055
.434	.032	.146		-.032	
.519		.074			-.067
.592		.013		-.073	-.079
.714				-.099	-.081
.765		.143		-.115	-.092
.897					-.085
.957	-.128			.252	-.092
.992	-.113	-.066		.129	-.104
$M = 2.75; \alpha = 0^\circ; \beta = 6^\circ; \delta_e = 0^\circ$					
.043	.171		.257		.089
.059	.065		.135		.007
.099	.078	.141		.082	.005
.237		.103		.011	-.042
.296	.081	.107			-.041
.434	.020	.099		.017	
.519		.036			-.039
.592		.008		-.023	-.033
.714				-.050	-.028
.765		.186		-.093	-.038
.897					-.047
.957	-.114			.112	-.055
.992	-.110	-.026		.137	-.074
$M = 2.75; \alpha = 0^\circ; \beta = 2^\circ; \delta_e = 0^\circ$					
.043	.190		.181		.097
.059	.078		.079		.016
.099	.086	.091		.051	.023
.237		.052		-.010	-.027
.296	.092	.056			-.022
.434	.008	.054		.005	
.519		-.013			-.018
.592		-.036		-.021	-.010
.714				-.033	-.007
.765		.053		-.073	-.016
.897					-.040
.957	-.098			.080	-.039
.992	-.105	-.055		.008	-.055
$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.190		.145		.096
.059	.079		.053		.017
.099	.084	.067		.037	.026
.237		.029		-.017	-.024
.296	.099	.027			-.020
.434	.019	.030		.000	
.519		-.021		-.011	-.013
.592		-.051		-.022	.001
.714				-.061	-.008
.765		.004			-.040
.897					-.025
.957	-.083			.049	-.025
.992	-.094	-.066		.000	-.043

TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x_B}{L_B}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 2.75; \alpha = 0^\circ; \beta = -1^\circ; \delta_e = 0^\circ$					
.043	.194		.132		.097
.059	.081		.045		.019
.099	.085	.060		.034	.028
.237		.022		-.017	-.022
.296	.101	.014			-.018
.434	.021	.021		.002	
.519		-.025			-.011
.592		-.055		-.006	-.002
.714				-.021	.002
.765		-.003		-.057	-.007
.897					-.038
.957	-.081			.033	-.024
.992	-.092	-.058		-.006	-.042
$M = 2.75; \alpha = 0^\circ; \beta = -4^\circ; \delta_e = 0^\circ$					
.043	.198		.089		.091
.059	.082		.017		.011
.099	.079	.031		.018	.021
.237		-.004		-.025	-.029
.296	.081	-.023			-.025
.434	.001	-.015		.001	
.519		-.048			-.019
.592		-.079		-.003	-.014
.714				-.027	-.011
.765		-.026		-.050	-.022
.897					-.039
.957	-.098			-.005	-.041
.992	-.107	-.074		-.043	-.054
$M = 2.75; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.200		.042		.069
.059	.081		-.013		-.009
.099	.065	-.009		.000	-.003
.237		-.032		-.034	-.051
.296	.076	-.058			-.053
.434	.020	-.062		-.008	
.519		-.044			-.050
.592		-.109		-.023	-.044
.714				-.036	-.043
.765		-.037		-.040	-.056
.897					-.053
.957	-.112			-.064	-.081
.992	-.110	-.085		-.079	-.086
$M = 2.75; \alpha = 5^\circ; \beta = -6^\circ; \delta_e = 0^\circ$					
.043	.111		.054		.157
.059	.021		-.002		.053
.099	.014	.008		.007	.063
.237		.010		-.036	-.007
.296	.100	.015			-.005
.434	.051	-.008		.018	
.519		-.062			-.020
.592		-.050		-.018	-.015
.714				-.014	-.022
.765		.000		-.036	-.021
.897					-.025
.957	-.064			-.038	-.024
.992	-.071	-.055		-.052	-.013
$M = 2.75; \alpha = 5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$					
.043	.115		.105		.177
.059	.028		.026		.069
.099	.033	.030		.044	.077
.237		-.013		-.016	.003
.296	.099	.052			.009
.434	.027	.026		.042	
.519		-.028			.011
.592		-.018		.003	.006
.714				.005	.014
.765		.046		-.021	.012
.897					-.011
.957	-.026			.046	.020
.992	-.050	-.023		-.002	.015

Body station $\frac{x_B}{L_B}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 2.75; \alpha = 0^\circ; \beta = -2^\circ; \delta_e = 0^\circ$					
.043	.190		.115		.094
.059	.079		.033		.015
.099	.081	.048		.026	.025
.237		.011		-.022	-.025
.296	.091	-.001			-.021
.434	.014	.007		.000	
.519		-.038			-.015
.592		-.062		-.006	-.007
.714				-.025	-.003
.765		-.001		-.059	-.013
.897					-.040
.957	-.085			.014	-.033
.992	-.097	-.057		-.021	-.049
$M = 2.75; \alpha = 0^\circ; \beta = -6^\circ; \delta_e = 0^\circ$					
.043	.199		.064		.080
.059	.082		.000		.002
.099	.073	.011		.009	.010
.237		-.018		-.030	-.042
.296	.064	-.043			-.037
.434	.002	-.039		-.004	
.519		-.060			-.033
.592		-.103		-.003	-.028
.714				-.030	-.025
.765		-.042		-.032	-.035
.897					-.044
.957	-.107			-.031	-.052
.992	-.111	-.089		-.066	-.066
$M = 2.75; \alpha = 5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.111		.051		.156
.059	.020		.000		.052
.099	.012	.006		.004	.061
.237		.016		-.037	-.007
.296	.152	.008			-.008
.434	.049	-.008		.011	
.519		-.059			-.034
.592		-.065		-.023	-.030
.714				-.019	-.032
.765		.024		-.039	-.036
.897					-.033
.957	-.064			-.032	-.034
.992	-.071	-.051		-.050	-.019
$M = 2.75; \alpha = 5^\circ; \beta = -4^\circ; \delta_e = 0^\circ$					
.043	.115		.076		.167
.059	.025		.010		.060
.099	.025	.017		.024	.069
.237		.003		-.027	-.001
.296	.087	.030			.002
.434	.049	.006		.030	
.519		-.036			-.002
.592		-.012		-.005	.000
.714				-.004	.001
.765		-.005		-.028	.000
.897					-.018
.957	-.056			-.002	.002
.992	-.063	-.048		-.035	.004
$M = 2.75; \alpha = 5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$					
.043	.113		.120		.181
.059	.029		.036		.073
.099	.036	.037		.056	.079
.237		-.014		-.012	.005
.296	.102	.066			.009
.434	.034	.033		.049	
.519		-.018			.014
.592		-.015		.003	.014
.714				.010	.015
.765		.065		-.019	.014
.897					-.008
.957	-.016			.068	.022
.992	-.046	-.013		.014	.018

TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x_B}{l_B}$	C_p at $\theta =$ —				
	0°	60°	90°	120°	180°
$M = 2.75; \alpha = 5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.112		.136		.184
.059	.029		.046		.075
.099	.038	.044		.066	.080
.237		-.015		-.004	.006
.296	.104	.078			.009
.434	.035	.037		.055	
.519		-.014			.016
.592		-.008		.001	.017
.714				.013	.014
.765		.071		-.018	.014
.897					-.007
.957	-.015			.091	.021
.992	-.044	-.004		.030	.018
$M = 2.75; \alpha = 5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$					
.043	.108		.169		.187
.059	.025		.069		.078
.099	.035	.058		.086	.078
.237		-.008		.011	.003
.296	.097	.101			.007
.434	.028	.035		.061	
.519		.003			.013
.592		.015		.013	.009
.714				.020	.016
.765		.191		-.013	.012
.897				-.010	.007
.957	-.035			.157	.019
.992	-.054	.007		.080	.014
$M = 2.75; \alpha = 5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$					
.043	.096		.241		.178
.059	.009		.119		.070
.099	.018	.092		.128	.058
.237		.014		.043	-.009
.296	.121	.161			-.008
.434	.051	.060		.075	
.519		.034			-.022
.592		.051		.023	-.018
.714				.012	-.025
.765		.232		-.021	-.026
.897					-.031
.957	-.075			.270	-.029
.992	-.081	.068		.150	-.023
$M = 2.75; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.364		.134		.036
.059	.203		.042		-.012
.099	.205	.094		-.016	.002
.237		-.002		-.056	-.036
.296	.068	-.036			-.027
.434	.066	.035		-.044	
.519		-.012			-.021
.592		-.057		-.058	-.014
.714				-.056	-.011
.765		-.055		-.080	-.007
.897					-.037
.957	-.103			-.043	.015
.992	-.088	-.097		-.047	-.017
$M = 2.75; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.275		.141		.053
.059	.141		.048		-.005
.099	.143	.082		.006	.010
.237		.010		-.037	-.031
.296	.083	-.029			-.024
.434	.041	.028		-.028	
.519		-.017			-.013
.592		-.057		-.029	-.003
.714				-.036	.004
.765		-.046		-.080	.001
.897					-.042
.957	-.115			.014	.006
.992	-.092	-.096		-.045	-.031

Body station $\frac{x_B}{l_B}$	C_p at $\theta =$ —				
	0°	60°	90°	120°	180°
$M = 2.75; \alpha = 5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$					
.043	.115		.155		.190
.059	.031		.060		.080
.099	.040	.054		.079	.082
.237		-.009		.007	.007
.296	.105	.093			.010
.434	.026	.040		.061	
.519		-.004			.018
.592		.004		.007	.017
.714				.019	.018
.765		.075		-.014	.016
.897					-.006
.957	-.019			.126	.023
.992	-.046	.000		.037	.019
$M = 2.76; \alpha = 5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$					
.043	.101		.201		.184
.059	.018		.090		.075
.099	.028	.072		.105	.069
.237		.000		.024	-.002
.296	.090	.122			.001
.434	.041	.036		.067	
.519		.014			-.001
.592		.029		.016	-.001
.714				.017	.000
.765		.171		-.017	-.003
.897					-.020
.957	-.063			.217	-.003
.992	-.074	.032		.109	-.002
$M = 2.75; \alpha = 5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$					
.043	.080		.286		.171
.059	.001		.154		.066
.099	.008	.116		.154	.047
.237		.034		.064	-.018
.296	.159	.208			-.019
.434	.056	.096		.088	
.519		.058			-.050
.592		.071		.033	-.044
.714				.023	-.048
.765		.290		-.024	-.056
.897					-.055
.957	-.088			.270	-.056
.992	-.085	.083		.186	-.044
$M = 2.75; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.317		.136		.037
.059	.172		.043		-.013
.099	.173	.088		-.012	.002
.237		.000		-.052	-.038
.296	.064	-.040			-.028
.434	.045	.031		-.041	
.519		-.016			-.015
.592		-.056		-.043	-.004
.714				-.050	-.000
.765		-.055		-.086	-.000
.897					-.043
.957	-.117			-.034	.039
.992	-.091	-.102		-.058	-.028
$M = 2.75; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.234		.144		.074
.059	.111		.052		.007
.099	.115	.076		.024	.018
.237		.021		-.026	-.025
.296	.092	-.014			-.020
.434	.030	.030		-.006	
.519		-.013			-.012
.592		-.053		-.013	.001
.714				-.023	.006
.765		-.022		-.066	.000
.897					-.036
.957	-.102			.048	-.006
.992	-.094	-.083		-.010	-.034

TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x_B}{l_B}$	C_p at $\theta = \text{---}$				
	0°	60°	90°	120°	180°
$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.196		.146		.102
.059	.085		.054		.022
.099	.090				.032
.237		.070		.040	
.296	.097	.030		-.017	-.020
.434	.023	.006			-.014
.519		.037		.007	
.592		-.012			-.013
.714		-.043		-.004	.002
.765				-.015	.007
.897		.011		-.052	.001
.957	-.082				-.031
.992	-.088			.060	-.017
		-.065		.006	-.034
$M = 2.75; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.132		.140		.162
.059	.042		.051		.059
.099	.049				.065
.237		.052		.061	
.296	.102	.006		-.007	-.005
.434	.039	.046			.001
.519		.050		.037	
.592		-.007			.001
.714		-.011		.004	.012
.765				.012	.018
.897		.061		-.025	.016
.957	-.033				-.012
.992	-.056			.094	.012
		-.017		.031	.004
$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.308		.134		.031
.059	.166		.046		-.012
.099	.164				-.004
.237		.086		-.009	
.296	.107	.036		-.041	-.028
.434	.084	.002			-.029
.519		.058		-.041	
.592		-.010			-.024
.714		-.020		-.043	-.009
.765				-.001	-.001
.897		-.023		-.063	-.005
.957	-.073			-.029	-.020
.992	-.071			-.034	-.001
		-.073		-.014	
$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.216		.137		.067
.059	.101		.049		.006
.099	.102				.011
.237		.069		.020	
.296	.130	.026		-.022	-.020
.434	.064	.027			-.022
.519		.045		-.013	
.592		-.018			-.016
.714		-.025		-.023	-.005
.765				-.025	.004
.897		-.034		-.053	.001
.957	-.067			-.022	-.007
.992	-.071			.031	-.020
		-.059		.014	
$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.150		.136		.121
.059	.057		.050		.035
.099	.060				.040
.237		.055		.045	
.296	.162	.008		-.008	-.011
.434	.058	.049			-.008
.519		.029		.027	
.592		-.006			-.002
.714		-.004		-.003	.007
.765				-.004	.014
.897		-.007		-.032	.014
.957	-.040				-.010
.992	-.054			.068	.002
		-.036		.038	-.006

Body station $\frac{x_B}{l_B}$	C_p at $\theta = \text{---}$				
	0°	60°	90°	120°	180°
$M = 2.75; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.159		.142		.129
.059	.060		.052		.038
.099	.066				.044
.237		.061		.051	
.296	.100	.020		-.013	-.016
.434	.020	.025			-.009
.519		.043		.019	
.592		-.012			-.012
.714		-.029		.000	.003
.765				-.004	.011
.897		.032		-.040	.006
.957	-.059				-.024
.992	-.074			.076	-.008
		-.044		.018	-.020
$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.350		.134		.031
.059	.191		.046		-.010
.099	.191				-.003
.237		.092		-.011	
.296	.104	.033		-.042	-.029
.434	.085	.002			-.028
.519		.064		-.038	
.592		-.002			-.023
.714		-.014		-.040	-.011
.765				-.047	-.001
.897		-.037		-.058	-.012
.957	-.068			-.031	-.031
.992	-.068			-.030	-.036
		-.068			.008
$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.262		.139		.046
.059	.135		.049		-.005
.099	.134				.003
.237		.079		.005	
.296	.119	.031		-.031	-.024
.434	.076	.012			-.025
.519		.053		-.027	
.592		-.016			-.022
.714		-.025		-.036	-.007
.765				-.038	.002
.897		-.023		-.063	.000
.957	-.074				-.025
.992	-.072			.005	.000
		-.070		-.015	-.007
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.183		.138		.091
.059	.079		.050		.018
.099	.081				.025
.237		.063		.033	
.296	.145	.018		-.014	-.015
.434	.062	.044			-.016
.519		.042		.003	
.592		-.014			-.008
.714		-.016		-.013	-.000
.765				-.017	.011
.897		-.024		-.040	.007
.957	-.055				-.018
.992	-.065			.049	-.007
		-.049		.027	-.018
$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.121		.132		.151
.059	.039		.047		.054
.099	.043				.058
.237		.046		.054	
.296	.179	-.001		-.003	-.000
.434	.067	.032			.001
.519		.029		.043	
.592		.004			-.004
.714		.011		.006	.017
.765				.010	.016
.897		.010		-.019	.021
.957	-.021				-.002
.992	-.041			.081	.017
		-.022		.043	.009

TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x_B}{L_B}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 3.22; \alpha = -5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.298		.042		.040
.059	.151		-.007		-.010
.099	.132	.002		-.006	-.014
.237		-.033		-.028	-.040
.296	.107	-.038		-.041	
.434	.080	-.037		-.023	-.047
.519		-.054		-.035	-.043
.592		-.058		-.044	-.043
.714				-.052	-.046
.765		-.050		-.022	-.048
.897				-.038	-.052
.957	-.066				
.992	-.069	-.052			
$M = 3.22; \alpha = -5^\circ; \beta = -4^\circ; \delta_e = 0^\circ$					
.043	.298		.070		.033
.059	.156		.001		-.015
.099	.146	.029		-.010	-.012
.237		-.016		-.034	-.034
.296	.095	-.044		-.035	
.434	.083	-.019		-.025	-.040
.519		-.057		-.026	-.036
.592		-.076		-.033	-.048
.714				-.051	-.057
.765		-.069		-.062	-.050
.897				-.027	-.058
.957	-.080			-.040	
.992	-.076	-.067			
$M = 3.22; \alpha = -5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$					
.043	.284		.119		.038
.059	.147		.034		-.008
.099	.144	.066		-.004	-.001
.237		.016		-.037	-.029
.296	.109	-.011		-.026	-.029
.434	.076	-.038		-.025	
.519		-.024		-.029	-.025
.592		-.043		-.049	-.002
.714				-.069	-.004
.765		-.063		-.031	-.002
.897				-.011	-.003
.957	-.081			-.002	-.015
.992	-.077	-.073		-.026	
$M = 3.22; \alpha = -5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$					
.043	.275		.150		.041
.059	.141		.055		-.009
.099	.141	.089		.001	-.002
.237		.043		-.033	-.027
.296	.127	.016		-.036	-.027
.434	.080	.062		-.051	-.025
.519		-.011		-.045	-.012
.592		-.018		-.070	-.007
.714				-.033	-.011
.765		.016		-.010	-.008
.897				.003	-.024
.957	-.082				
.992	-.079	-.072			
$M = 3.22; \alpha = -5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$					
.043	.269		.208		.042
.059	.137		.097		-.012
.099	.143	.134		.021	-.008
.237		.088		-.024	-.031
.296	.133	.053		-.032	-.032
.434	.095	.084		-.060	-.035
.519		.021		-.061	-.039
.592		.018		-.074	-.058
.714				.010	-.042
.765		.093		.093	-.060
.897					
.957	-.085				
.992	-.077	-.059			

Body station $\frac{x_B}{L_B}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 3.22; \alpha = -5^\circ; \beta = -6^\circ; \delta_e = 0^\circ$					
.043	.299		.048		.031
.059	.156		-.010		-.016
.099	.142	.010		-.012	-.017
.237		-.029		-.032	-.040
.296	.094	-.048			-.040
.434	.083	-.040		-.030	
.519		-.065			-.047
.592		-.072		-.023	-.047
.714				-.048	-.057
.765		-.066		-.048	-.061
.897				-.025	-.054
.957	-.077			-.040	-.054
.992	-.074	-.065			-.063
$M = 3.22; \alpha = -5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$					
.043	.291		.103		.038
.059	.152		.022		-.010
.099	.145	.054		-.006	-.004
.237		.004		-.036	-.029
.296	.103	-.025			-.029
.434	.074	.019		-.024	-.030
.519		-.035		-.026	-.020
.592		-.053		-.051	-.019
.714				-.068	-.022
.765		-.073		-.043	-.024
.897				-.010	-.031
.957	-.083			-.043	
.992	-.077	-.072			
$M = 3.22; \alpha = -5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.281		.134		.039
.059	.144		.044		-.009
.099	.143	.078		-.002	-.001
.237		.028		-.036	-.027
.296	.114	.002			-.029
.434	.079	.051		-.032	
.519		-.019		-.038	-.024
.592		-.030		-.046	-.008
.714				-.069	-.000
.765		-.034		-.014	-.002
.897				-.011	-.003
.957	-.081				-.015
.992	-.078	-.075			
$M = 3.22; \alpha = -5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$					
.043	.273		.166		.040
.059	.140		.067		-.011
.099	.141	.102		.006	-.005
.237		.059		-.031	-.028
.296	.132	.029			-.028
.434	.084	.025		-.038	
.519		-.002		-.060	-.028
.592		-.008		-.050	-.018
.714				-.072	-.025
.765		.051			-.045
.897				.005	-.021
.957	-.084			.041	-.038
.992	-.078	-.070			
$M = 3.22; \alpha = -5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$					
.043	.261		.253		.039
.059	.132		.128		-.017
.099	.140	.164		.037	-.016
.237		.117		-.015	-.039
.296	.119	.080			-.040
.434	.098	.105		-.023	
.519		.049		-.053	-.043
.592		.042		-.065	-.043
.714				-.076	-.058
.765		.134			-.060
.897				.067	-.057
.957	-.085			.115	-.067
.992	-.077	-.047			

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TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x_B}{l_B}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 3.22; \alpha = -5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$					
.043	.254		.303		.033
.059	.127		.167		-.023
.099	.138	.199		-.058	-.023
.237		.152		-.002	-.050
.296	.110	.116			-.051
.434	.096	.143		-.010	
.519		.076			-.056
.592		.063		-.042	-.055
.714				-.063	-.059
.765		.183		-.079	-.065
.897					-.065
.957	-.086			.154	-.066
.992	-.078	-.034		.147	-.072
$M = 3.22; \alpha = 0^\circ; \beta = 6^\circ; \delta_e = 0^\circ$					
.043	.164		.248		.085
.059	.064		.129		.010
.099	.073	.132		.079	.004
.237		.059		.016	-.034
.296	.128	.134			-.037
.434	.078	.089		.014	
.519		.046			-.036
.592		.056		-.019	-.027
.714				-.035	-.023
.765		.149		-.065	-.029
.897					-.037
.957	-.083			.122	-.044
.992	-.077	-.004		.114	-.057
$M = 3.22; \alpha = 0^\circ; \beta = 2^\circ; \delta_e = 0^\circ$					
.043	.180		.172		.091
.059	.074		.076		.016
.099	.080	.086		.048	.019
.237		.032		-.004	-.022
.296	.148	.074			-.021
.434	.060	.041		.003	
.519		.002		-.015	-.015
.592		.004		-.023	-.005
.714				-.028	.001
.765		.043		-.054	-.005
.897				-.028	-.028
.957	-.075			.075	-.024
.992	-.077	-.032		.028	-.038
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.188		.139		.089
.059	.080		.053		.018
.099	.082	.066		.035	.023
.237		.021		-.011	-.017
.296	.144	.044			-.017
.434	.059	.037		.000	
.519		-.017		-.016	-.011
.592		-.020		-.001	-.001
.714				-.017	.001
.765		-.023		-.041	-.001
.897					-.021
.957	-.060			.042	-.012
.992	-.068	-.048		.024	-.024
$M = 3.22; \alpha = 0^\circ; \beta = -2^\circ; \delta_e = 0^\circ$					
.043	.190		.108		.096
.059	.082		.032		.015
.099	.080	.046		.023	.021
.237		.011		-.017	-.019
.296	.129	.015			-.019
.434	.051	.021		.000	
.519		-.031			-.012
.592		-.044		-.009	-.003
.714				-.013	.000
.765		-.027		-.040	-.002
.897					-.025
.957	-.063			.020	-.019
.992	-.074	-.051		-.001	-.027

Body station $\frac{x_B}{l_B}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 3.22; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$					
.043	.161		.290		.080
.059	.065		.157		.005
.099	.074	.157		.094	-.004
.237		.079		.026	-.041
.296	.126	.162			-.046
.434	.087	.121		.023	
.519		.072			-.046
.592		.078		-.012	-.036
.714				-.028	-.039
.765		.209		-.062	-.044
.897					-.042
.957	-.081			.106	-.059
.992	-.077	.009		.156	-.061
$M = 3.22; \alpha = 0^\circ; \beta = 4^\circ; \delta_e = 0^\circ$					
.043	.173		.208		.090
.059	.070		.101		.014
.099	.077	.109		.063	.012
.237		.042		.004	-.027
.296	.138	.101			-.028
.434	.073	.058		.007	
.519		.023			-.024
.592		.029		-.024	-.014
.714				-.038	-.008
.765		.096		-.064	-.014
.897					-.033
.957	-.082			.094	-.032
.992	-.079	-.019		.033	-.048
$M = 3.22; \alpha = 0^\circ; \beta = 1^\circ; \delta_e = 0^\circ$					
.043	.184		.154		.090
.059	.078		.063		.017
.099	.082	.075		.041	.022
.237		.026		-.009	-.019
.296	.142	.055			-.019
.434	.054	.040		.001	
.519		-.009			-.011
.592		-.007		-.019	-.002
.714				-.021	.004
.765		.011		-.047	-.001
.897					-.024
.957	-.066			.056	-.017
.992	-.074	-.040		.028	-.030
$M = 3.22; \alpha = 0^\circ; \beta = -1^\circ; \delta_e = 0^\circ$					
.043	.188		.122		.086
.059	.081		.041		.016
.099	.081	.055		.028	.023
.237		.014		-.014	-.017
.296	.141	.029			-.018
.434	.057	.028		.001	
.519		-.025			-.010
.592		-.034		-.012	-.001
.714				-.014	.011
.765		-.027		-.040	.001
.897					-.022
.957	-.059			.031	-.013
.992	-.071	-.044		.011	
$M = 3.22; \alpha = 0^\circ; \beta = -4^\circ; \delta_e = 0^\circ$					
.043	.193		.086		.084
.059	.082		.019		.011
.099	.078	.031		.014	.017
.237		.006		-.020	-.021
.296	.112	-.003			-.023
.434	.056	.004		-.001	
.519		-.046			-.016
.592		-.062		-.006	-.008
.714				-.017	.004
.765		-.029		-.039	-.009
.897					-.029
.957	-.075			.001	-.027
.992	-.076	-.059		-.020	-.037

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TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Continued

Body station $\frac{x}{L}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 3.22; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.195		.063		.078
.059	.083		.005		.006
.099	.073	.014		.007	.010
.237		-.005		-.022	-.027
.296	.108	-.020			-.031
.434	.062	-.020		-.004	
.519		-.059			-.026
.592		-.073		-.013	-.019
.714				-.020	-.010
.765		-.045		-.025	-.033
.897					-.037
.957	-.078			.021	.037
.992	-.078	-.059		-.036	-.047
$M = 3.22; \alpha = 5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.111		.045		.148
.059	.030		-.002		.047
.099	.020	.003		.002	.059
.237		-.023		-.032	-.004
.296	.128	.002			-.009
.434	.079	.001		.010	
.519		-.046		-.027	
.592		-.054		-.009	-.024
.714				-.004	-.026
.765		-.044		-.024	-.025
.897				-.027	
.957	-.062			-.031	-.032
.992	-.063	-.040		-.038	-.022
$M = 3.22; \alpha = 5^\circ; \beta = -4^\circ; \delta_e = 0^\circ$					
.043	.106		.072		.161
.059	.025		.010		.059
.099	.023	.016		.026	.068
.237		-.016		-.016	.009
.296	.144	.015			.009
.434	.059	.029		.030	
.519		-.023			-.011
.592		-.013		.000	.005
.714				.009	.009
.765		-.034		-.014	.011
.897					-.008
.957	-.044			-.000	.007
.992	-.052	-.024		-.021	.011
$M = 3.22; \alpha = 5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$					
.043	.102		.110		.168
.059	.025		.031		.069
.099	.028	.029		.047	.069
.237		-.013		-.004	.009
.296	.166	.040			.011
.434	.069	.040		.047	
.519		-.003			-.002
.592		.009		.009	.016
.714				.016	.017
.765		-.008		-.012	.024
.897					.004
.957	-.006			.056	.026
.992	-.027	.003		.021	.022
$M = 3.22; \alpha = 5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$					
.043	.101		.139		.173
.059	.025		.050		.070
.099	.030	.042		.065	.070
.237		-.011		.005	.009
.296	.165	.041			.010
.434	.068	.038		.056	
.519		.013			-.003
.592		.029		.013	.018
.714				.018	.017
.765		.037		-.011	.024
.897					.004
.957	-.012			.106	.026
.992	-.030	-.008		.060	.022

Body station $\frac{x}{L}$	C_p at $\theta = -$				
	0°	60°	90°	120°	180°
$M = 3.22; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.197		.049		.072
.059	.084		-.002		.001
.099	.070	.003		.001	.003
.237		-.013		-.025	-.033
.296	.107	-.051			-.039
.434	.069	-.036		-.007	
.519		-.065			-.035
.592		-.072		-.016	-.029
.714				-.020	-.024
.765		-.049		-.024	-.033
.897					-.035
.957	-.079			-.036	-.049
.992	-.077	-.054		-.047	-.054
$M = 3.22; \alpha = 5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$					
.043	.105		.052		.158
.059	.024		.001		.057
.099	.016	.007		.011	.068
.237		-.021		-.027	.003
.296	.127	.007			.001
.434	.071	.013		.020	
.519		-.037			-.021
.592		-.046		-.007	-.013
.714				.000	-.007
.765		-.038		-.021	-.006
.897					-.021
.957	-.059			-.030	-.017
.992	-.062	-.033		-.037	-.007
$M = 3.22; \alpha = 5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$					
.043	.104		.093		.166
.059	.026		.021		.063
.099	.027	.024		.037	.068
.237		-.013		-.009	.011
.296	.163	.030			.013
.434	.061	.039		.039	
.519		-.010			-.004
.592		-.001		.006	.014
.714				.014	.017
.765		-.028		-.010	.018
.897					.000
.957	-.023			.035	.021
.992	-.037	-.003		.004	.019
$M = 3.22; \alpha = 5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$					
.043	.101		.125		.169
.059	.026		.041		.068
.099	.030	.037		.057	.070
.237		-.013		.000	.009
.296	.167	.041			.010
.434	.070	.037		.052	
.519		.006			-.002
.592		.018		.010	.018
.714				.016	.019
.765		.013		-.011	.025
.897					.005
.957	-.006			.080	.026
.992	-.027	-.008		.041	.023
$M = 3.22; \alpha = 5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$					
.043	.102		.158		.177
.059	.026		.061		.072
.099	.031	.051		.076	.071
.237		-.005		.012	.009
.296	.161	.044			.010
.434	.067	.045		.049	
.519		.023			-.003
.592		.043		.020	.019
.714				.022	.021
.765		.065		-.009	.004
.897					.023
.957	-.019			.134	.023
.992	-.033	.002		.083	.022

TABLE III. - PRESSURE COEFFICIENTS ON BOOSTER BODY MEASURED IN PRESENCE OF MISSILE - Concluded

Body station $\frac{x_B}{l_B}$	C_p at $\theta =$ —				
	0°	60°	90°	120°	180°
$M = 3.22; \alpha = 5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$					
.043	.095		.185		.174
.059	.020		.078		.070
.099	.026	.063		.092	.064
.237		.002		.022	.005
.296	.149	.040		.045	.006
.434	.074	.058			
.519		.035			-.013
.592		.060		.019	.006
.714				.020	.009
.765		.111		-.009	.009
.897					-.006
.957	-.036			.184	.010
.992	-.046	.012		.112	.010
$M = 3.22; \alpha = 5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$					
.043	.079		.263		.169
.059	.007		.132		.068
.099	.012	.101		.136	.052
.237		.026		.056	-.006
.296	.142	.061			-.008
.434	.098	.110		.066	
.519		.075			-.032
.592		.115		.037	-.024
.714				.035	-.018
.765		.227		-.002	-.022
.897				-.026	-.002
.957	-.053			.301	-.028
.992	-.058	.062		.152	-.013
$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.309		.135		.032
.059	.167		.047		-.011
.099	.165	.087		-.008	-.003
.237		.032		-.041	-.030
.296	.081	-.009			-.029
.434	.086	.064		-.036	
.519		-.009			-.028
.592		-.014		-.047	-.009
.714				-.044	-.002
.765		-.024		-.065	-.004
.897				-.031	-.031
.957	-.077			-.032	-.016
.992	-.076	-.068		-.018	.002
$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.220		.137		.065
.059	.105		.049		.005
.099	.106	.070		.018	.011
.237		.027		-.023	-.021
.296	.095	.010			-.023
.434	.063	.050		-.017	
.519		-.015		-.025	-.021
.592		-.019		-.006	-.006
.714				-.027	.005
.765		-.022		-.055	.003
.897					-.023
.957	-.068			.031	-.005
.992	-.074	-.059		.015	-.019
$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.154		.137		.118
.059	.061		.050		.032
.099	.063	.057		.044	.038
.237		.007		-.010	-.010
.296	.123	.038			
.434	.053	.039		.016	
.519		-.006			-.007
.592		.000		-.003	.003
.714				-.005	.016
.765		-.003		-.033	.015
.897					-.011
.957	-.045			.070	.003
.992	-.057	-.038		.040	-.005

Body station $\frac{x_B}{l_B}$	C_p at $\theta =$ —				
	0°	60°	90°	120°	180°
$M = 3.22; \alpha = 5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$					
.043	.088		.225		.171
.059	.015		.105		.068
.099	.020	.081		.113	.058
.237		.013		.038	.000
.296	.131	.046			-.001
.434	.085	.081		.055	
.519		.055			-.023
.592		.086		.026	-.008
.714				.026	-.003
.765		.168		-.007	-.004
.897					-.016
.957	-.045			.261	-.008
.992	-.052	.032		.154	.003
$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.353		.134		.031
.059	.195		.048		-.010
.099	.194	.093		-.011	-.004
.237		.026		-.042	-.033
.296	.078	-.012			-.032
.434	.093	.068		-.038	
.519		-.005			-.027
.592		-.010		-.042	-.013
.714				-.046	-.005
.765		-.020		-.060	-.010
.897					-.037
.957	-.073			-.027	-.040
.992	-.073	-.061		-.029	.023
$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.262		.137		.045
.059	.134		.048		-.005
.099	.133	.078		.004	.001
.237		.032		-.032	-.025
.296	.089	-.003			-.026
.434	.073	.055		-.026	
.519		-.012			-.026
.592		-.018		-.037	-.007
.714				-.035	.003
.765		-.020		-.062	.001
.897					-.025
.957	-.076			.000	.009
.992	-.075	-.067		-.016	-.005
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.186		.138		.089
.059	.082		.051		.017
.099	.084	.064		.031	.024
.237		.018		-.016	-.016
.296	.107	.025			-.017
.434	.059	.049		-.003	
.519		-.012			-.014
.592		-.011		-.014	-.003
.714				-.017	.011
.765		-.015		-.042	.008
.897					-.019
.957	-.059			.051	-.007
.992	-.067	-.050		.026	-.018
$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$					
.043	.128		.133		.143
.059	.045		.047		.047
.099	.047	.048		.051	.033
.237		.000		-.006	-.004
.296	.147	.026			-.003
.434	.054	.033		.027	
.519		.002			-.008
.592		.012		.005	.009
.714				.007	.016
.765		.011		-.025	.020
.897					-.005
.957	-.031			.082	.014
.992	-.048	-.028			.007

TABLE IV.- PRESSURE COEFFICIENTS ON MISSILE BODY

Body station $\frac{x_M}{L_M}$	C_p at $\theta = -$						Body station $\frac{x_M}{L_M}$
	0°		60°	90°	120°	180°	
	Body center line	Nacelle center line				Body center line	Nacelle center line
$M = 2.29; \alpha = -10^\circ; \beta = 0^\circ; \delta_e = -5^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.244 .355 .334 .260			-.001 -.013 -.039 -.046		-.010 -.073 -.084 -.091 .019 -.004 -.027	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
		.360 .223 .155 .132 .132 .097 .170	.153 -.009 -.046 .144 -.142		-.106 -.143 -.129 .156 -.140 -.123	-.058 -.130 -.087 -.069 -.094	
		.154 .125 .032	.089	-.075	-.057	-.121 -.114	
$M = 2.29; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.203 .293 .277 .210		.002 .003 -.020 -.028		-.005 -.058 -.072 -.077 .014 .003 -.018	-.037 -.089 -.080 -.057 -.085	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
	.125 .104 .070 .103 .133	.277 .175 .110 .107 .112	.130 -.020 -.048 .114 -.116		-.068 -.104 -.122 .120 -.122 -.112	-.066 -.095	-.089 -.080 -.057 -.085 -.114 -.099
		.117 .082 .006	.057 -.079		-.053		
$M = 2.29; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.163 .230 .226 .162		.002 .018 -.004 -.018		.001 -.038 -.052 -.060 .013 .007 -.013	-.026 -.075 -.050 -.056 -.065	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
	.107 .073 .044 .071 .100	.199 .114 .069 .073 .080	.063 -.048 -.059 .085 -.094		-.005 -.095 -.057 .088 -.102 -.102	-.059 -.066	-.097 -.098
		.084 .057 -.019	.025 -.068		-.051		
$M = 2.29; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.123 .179 .180 .119		.011 .025 .006 -.011		.009 -.019 -.033 -.042 .005 .011 -.006	-.019 -.071 -.034 -.049 -.037	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
	.084 .046 .027 .044 .067	.110 .063 .028 .045 .050	.106 -.039 -.031 .068 -.077		.062 -.063 -.060 .069 -.091 -.093	-.058 -.049	-.076 -.091
		.052 .028 -.038	.000 -.053		-.055		

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —							Body station $\frac{x_M}{L_M}$
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line	Nacelle center line	
$M = 2.29; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095	.092					.025		.095
.124	.129					.003		.124
.153	.132					-.010		.153
.191	.078			.011		-.020		.191
.267				.026		-.002		.267
.344				.011		.017		.344
.439				-.005		-.004		.439
.538		.052	.145		.046		-.005	.538
.556		.003	-.027		-.045		-.069	.556
.592	.067	-.001	-.028		-.046	-.047	-.028	.592
.668	.030	.025	.031			-.044	-.035	.668
.744	.009	.024	-.041		.040		-.018	.744
.821	.017				-.064			.821
.897	.036	.022	-.025		-.072		-.054	.897
.945		-.001						.945
.989		-.061	-.044		-.049		-.077	.989
$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095	.068					.044		.095
.124	.091					.027		.124
.153	.090					.014		.153
.191	.039			.015		.001		.191
.267				.023		.000		.267
.344				.009		.022		.344
.439				-.006		.001		.439
.538		.048	.156		.049		.011	.538
.556		-.020	-.006		-.038		-.065	.556
.592	.055	-.019	-.038		-.037	-.036	-.023	.592
.668	.017	.008	.008			-.042	-.031	.668
.744	-.001	.004	-.004		.006		-.003	.744
.821	-.004				-.040			.821
.897	.005	-.006	-.048		-.054		-.032	.897
.945		-.025						.945
.989		-.084	-.044		-.047		-.061	.989
$M = 2.29; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095	.048					.062		.095
.124	.058					.053		.124
.153	.056					.041		.153
.191	.010			.015		.027		.191
.267				.019		.011		.267
.344				.007		.025		.344
.439				-.009		.011		.439
.538		.039	.115		.056		.036	.538
.556		-.032	-.018		-.030		-.045	.556
.592	.047	-.033	-.041		-.031	-.018	-.015	.592
.668	.005	-.004	-.017			-.033	-.024	.668
.744	-.011	-.014	.033		-.021		.014	.744
.821	-.021				-.014			.821
.897	-.017	-.028	-.068		-.033		-.010	.897
.945		-.047						.945
.989		-.100	-.045		-.042		-.047	.989
$M = 2.29; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095	.030					.084		.095
.124	.033					.082		.124
.153	.019					.071		.153
.191	-.016			-.002		.056		.191
.267				.013		.032		.267
.344				-.003		.033		.344
.439				-.017		.020		.439
.538		.011	.078		.062		.056	.538
.556		-.050	-.032		-.021		-.020	.556
.592	.031	-.050	-.050		-.028	.021	-.002	.592
.668	-.007	-.017	-.047			-.018	-.015	.668
.744	-.020	-.030	.074		-.047		.035	.744
.821	-.039				.017			.821
.897	-.038	-.050	-.085		-.010		.015	.897
.945		-.060						.945
.989		-.116	-.047		-.040		-.030	.989

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x}{L}$	C_p at $\theta = -$							Body station $\frac{x}{L}$
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line	Nacelle center line	
$M = 2.29; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.016 .015 -.012 -.042					.106 .111 .102 .086 .059 .039 .031		.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
				-.016 .000 -.019 -.030				

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{l_M}$ $\frac{z_M}{l_M}$	C_p at $\theta = \text{---}$						Body station $\frac{x_M}{l_M}$ $\frac{z_M}{l_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.29; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	-.004 -.048 -.077 -.096 -.002 -.035 -.045 -.072 -.074 -.095 -.111 -.132	 -.059 -.101 -.069 -.038 -.064 -.123 -.111 -.132	 .014 -.076 -.078 -.110 .152 -.123 -.074	 -.013 -.027 -.042 -.051 -.070 -.014 -.035 -.097 .086 .049 -.039	 .134 .164 .172 .150 .096 .053 .034 .052 .014 .077 .004	 		

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta = \text{---}$						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.29; \alpha = 0^0; \beta = 0^0; \delta_e = 0^0$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.051 .051 .032 .009 .068 .024 .003 -.001 .008 <							

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.29; \alpha = -8^0; \beta = 0^0; \delta_e = 0^0$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.145 .197 .212 .158 .126 .108 .074 .105 .132 	 .325 .193 .125 .108 .117 .116 .089 .006	 .165 -.003 -.040 .116 -.116 .053 -.079	 .014 -.005 -.024 -.041 	 -.042 -.094 -.102 .120 -.117 -.105 -.056	 -.070 -.111 -.058 -.066 -.079 -.108 -.099	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 2.29; \alpha = -10^0; \beta = 0^0; \delta_e = 0^0$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.185 .240 .260 .205 .137 .136 .099 .137 .169 	 .390 .244 .164 .138 .146 .153 .123 .032	 .172 .002 -.040 .149 -.140 .087 -.079	 .007 -.029 -.047 -.062 	 -.073 -.126 -.125 .156 -.131 -.110 -.061	 -.094 -.125 -.064 -.077 -.092 -.125 -.106	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 2.29; \alpha = 0^0; \beta = -6^0; \delta_e = 0^0$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.014 .014 .005 -.010 .042 .011 -.002 -.010 .000 	 .093 .043 .017 -.005 -.029 -.027 -.048 -.142	 .062 -.067 -.072 .023 -.003 -.050 -.060	 -.007 -.009 -.011 -.024 	 -.040 -.083 -.052 .001 -.030 -.046 -.049	 .004 -.037 -.072 -.079 -.037 -.041 -.084	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 2.29; \alpha = 0^0; \beta = -6^0; \delta_e = 0^0$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.028 .029 .017 -.003 .050 .015 .002 -.003 .007 	 .093 .035 .015 .009 -.016 -.037 -.141	 .076 -.064 -.064 .020 -.005 -.052 -.054	 -.002 -.002 -.006 -.017 	 -.017 -.071 -.047 .003 -.033 -.047 -.045	 .000 -.051 -.053 -.069 -.029 -.038 -.076	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta = -$						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.29; \alpha = 0^\circ; \beta = -4^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.042 .038 .034 .005 .062 .020 .007 .003 .012 	 .090 .035 .010 .020 .004 -.004 -.026 -.113	 .096 -.048 -.052 .021 -.011 -.050 -.052	 .008 .005 -.002 -.013 -.062 -.046 .005 -.040 -.053 -.042	 <			

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{l_M}$	C_p at $\theta = \text{---}$						Body station $\frac{x_M}{l_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.29; \alpha = 0^\circ; \beta = 1^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.051 .051 .034 .010			.029 .026 .013 -.002	.037 -.050 -.043 -.053 .010 -.050 -.065 -.051	.044 .042 .033 .020 .007 .008 -.009 -.006 -.069 -.033 -.034 -.019 -.046 -.068	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 2.29; \alpha = 0^\circ; \beta = 2^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.048 .048 .037 .009			.036 .033 .018 .002	.047 -.045 -.041 -.009 -.050 -.066 -.054	.042 .040 .032 .018 .004 .005 -.012 -.018 -.070 -.036 -.031 -.019 -.049 -.068	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 2.29; \alpha = 0^\circ; \beta = 4^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.040 .040 .031 .006			.051 .046 .032 .012	.064 -.033 -.035 -.011 -.048 -.066 -.059	.036 .032 .023 .006 -.007 -.008 -.022 -.037 -.073 -.043 -.025 -.022 -.062 -.073	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 2.29; \alpha = 0^\circ; \beta = 6^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.027 .027 .017 -.001			.068 .062 .047 .023	.085 -.020 -.028 -.012 -.050 -.061 -.060	.023 .020 .006 -.008 -.025 -.030 -.040 -.051 -.084 -.062 -.044 -.019 -.079 -.084	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —							Body station $\frac{x_M}{L_M}$
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line	Nacelle center line	
$M = 2.29; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.009 .006 .000 -.011 .043 .004 -.005 -.013 -.006	 .100 .030 .001 -.039 -.009 -.018 -.014 -.022	 .264 .061 .019 .057 -.048 -.020 -.054	 .090 .079 .065 .039 <				

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{l_M}$		C_p at $\theta = -$						Body station $\frac{x_M}{l_M}$	
$\frac{x_M}{l_M}$		0°		60°	90°	120°	180°		
$\frac{x_M}{l_M}$		Body center line	Nacelle center line				Body center line	Nacelle center line	
$M = 2.29; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$									
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.079 .088 .072 .045						.023 .020 .016 -.003 -.001 -.002 -.013	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
		.244 .118 .061 .062 .060 .053 .028 -.046	.183 .004 -.030 .081 -.076 -.004 -.068			-.011 -.083 -.075 .070 -.086 -.087 -.054	-.040 -.101 -.044 -.056 -.045 -.078 -.083		
$M = 2.29; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$									
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.059 .056 .037 .012			.019 .018 .007 -.009		.041 .043 .034 .024 .007 .002 -.010	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989		
	.095 .051 .022 .027 .038	.186 .084 .039 .042 .039 .027 .001 -.065	.180 -.001 -.035 .060 -.039 -.028 -.058		.001 -.068 -.061 .039 -.062 -.075 -.051	-.021 -.082 -.044 -.042 -.029 -.056 -.071			
$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 5^\circ$									
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.040 .025 .005 -.018			.016 .020 .008 -.006		.053 .069 .062 .050 .014 .003 -.008	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989		
	.082 .031 .004 .003 .010	.146 .055 .015 .022 .017 .001 -.023 -.083	.124 -.031 -.047 .030 -.008 -.055 -.055		.029 -.056 -.049 .005 -.041 -.059 -.045	.005 -.071 -.037 -.040 -.014 -.035 -.061			
$M = 2.29; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$									
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.024 .000 -.021 -.041			.021 .018 .006 -.007		.065 .093 .094 .079 .025 .007 -.004	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989		
	.063 .017 -.007 -.015 -.011	-.005 .006 -.001 -.021 -.039 -.099	.100 -.038 -.046 -.002 .028 -.076 -.052		.037 -.043 -.039 -.024 -.016 -.038 -.042	.029 -.057 -.027 -.031 .005 -.015 -.049			

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 2.29; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$

.095	.007					.085	.095
.124	-.028					.123	.124
.153	-.049			.021		.135	.153
.191	-.067			.011		.112	.191
.267				-.002		.043	.267
.344				-.013		.015	.344
.439					.048	.003	.439
.538		.051	.087		-.032	.058	.538
.556		-.012	-.053		-.032	-.035	.556
.592	.043	-.026	-.046		-.032	-.001	.592
.668	.002	-.014	-.040		-.024	-.019	.668
.744	-.022	-.022	.069		-.045	.028	.744
.821	-.035				.015		.821
.897	-.034	-.046	-.095		-.011	.013	.897
.945		-.058					.945
.989		-.117	-.055		-.041	-.030	.989

$M = 2.29; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = 5^\circ$

.095	-.003					.107	.095
.124	-.048					.155	.124
.153	-.071			.017		.176	.153
.191	-.089			.000		.147	.191
.267				-.014		.065	.267
.344				-.024		.029	.344
.439					.058	.013	.439
.538		-.006	.044		-.024	.086	.538
.556		-.051	-.053		-.034	-.014	.556
.592	.023	-.046	-.058			.021	.592
.668	-.015	-.022	-.064		-.008	-.008	.668
.744	-.039	-.042	.108		-.068	.049	.744
.821	.055				.046		.821
.897	-.057	-.068	-.106		.014	.040	.897
.945		-.080					.945
.989		-.126	-.068		-.040	-.016	.989

$M = 2.29; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = 5^\circ$

.095	-.011					.136	.095
.124	-.072					.192	.124
.153	-.095			.009		.219	.153
.191	-.111			-.016		.187	.191
.267				-.038		.093	.267
.344				-.042		.049	.344
.439					.065	.026	.439
.538		-.052	.013		-.018	.119	.538
.556		-.087	-.076		-.038	.011	.556
.592	-.001	-.064	-.073			.050	.592
.668	-.034	-.037	-.103		.012	.006	.668
.744	-.047	-.065	.151		-.094	.077	.744
.821	-.075				.084		.821
.897	-.078	-.097	-.122		.048	.076	.897
.945		-.109					.945
.989		-.129	-.081		-.039	.006	.989

$M = 2.29; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = 5^\circ$

.095	-.021					.165	.095
.124	-.089					.227	.124
.153	-.114			-.012		.257	.153
.191	-.128			-.031		.226	.191
.267				-.065		.123	.267
.344				-.071		.070	.344
.439					.069	.045	.439
.538		-.091	-.017		-.014	.145	.538
.556		-.122	-.101		-.042	.031	.556
.592	-.016	-.074	-.093			.076	.592
.668	-.047	-.043	-.130		.035	.022	.668
.744	-.055	-.085	.180		-.114	.099	.744
.821	-.090				.124		.821
.897	-.095	-.113	-.135		.079	.105	.897
.945		-.122					.945
.989		-.126	-.076		-.040	.023	.989

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x}{M}$ $\frac{z}{M}$	C_p at $\theta =$ —						Body station $\frac{x}{M}$ $\frac{z}{M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 2.75; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.095	.152					-.001		.095
.124	.217					-.035		.124
.153	.257					-.052		.153
.191	.204			.006		-.057		.191
.267				.007		-.003		.267
.344				-.010		.002		.344
.439				-.030		-.011		.439
.538		.289	.165		-.054		-.026	.538
.556		.183	.006		-.086		-.063	.556
.592	.102	.133	-.029		-.092	-.056	-.063	.592
.668	.100	.108	.096			-.066	-.048	.668
.744	.077	.083	-.113		.105		-.063	.744
.821	.071				-.110			.821
.897	.106	.095	.055		-.095		-.088	.897
.945	.077	.077						.945
.989	.019	.019	-.048		-.047		-.083	.989

$M = 2.75; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.095	.119					.005		.095
.124	.177					-.020		.124
.153	.208					-.035		.153
.191	.162			.008		-.042		.191
.267				.019		-.013		.267
.344				.000		.004		.344
.439				-.015		-.005		.439
.538		.212	.133		-.001		-.021	.538
.556		.120	-.008		-.070		-.054	.556
.592	.093	.093	-.039		-.049	-.042	-.040	.592
.668	.075	.082	.070			-.051	-.044	.668
.744	.053	.057	-.086		.071		-.046	.744
.821	.045				-.089			.821
.897	.076	.067	.028		-.085		-.076	.897
.945		.049						.945
.989		-.003	-.058		-.041		-.080	.989

$M = 2.75; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.095	.089					.011		.095
.124	.139					-.005		.124
.153	.165					-.018		.153
.191	.123			.014		-.026		.191
.267				.025		-.018		.267
.344				.008		.006		.344
.439				-.007		-.003		.439
.538		.107	.119		.069		-.015	.538
.556		.070	-.003		-.037		-.049	.556
.592	.076	.048	-.028		-.041	-.034	-.031	.592
.668	.049	.053	.057			-.042	-.032	.668
.744	.032	.039	-.067		.053		-.031	.744
.821	.027				-.074			.821
.897	.052	.042	.006		-.074		-.054	.897
.945		.025						.945
.989		-.023	-.055		-.048		-.080	.989

$M = 2.75; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$

.095	.074					.023		.095
.124	.101					.011		.124
.153	.126					-.001		.153
.191	.084			.013		-.010		.191
.267				.024		-.014		.267
.344				.014		.012		.344
.439				-.001		-.001		.439
.538		.063	.153		.058		.002	.538
.556		.013	.095		-.020		-.044	.556
.592	.062	.020	-.012		-.030	-.027	-.023	.592
.668	.032	.030	.028			-.039	-.023	.668
.744	.016	.020	-.038		.033		-.017	.744
.821	.010				-.055			.821
.897	.028	.018	-.014		-.057		-.039	.897
.945		.001						.945
.989		-.040	-.045		-.042		-.065	.989

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{l_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{l_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.055 .071 .089 .051 .052 .020 .005 .000 .007	 						

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TABLE IV.- PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station x_M z_M		C_p at $\theta =$ —						Body station x_M z_M		
		0°		60°	90°	120°	180°			
		Body center line	Nacelle center line				Body center line			Nacelle center line
$M = 2.75; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.004 .003 -.023 -.047 .002 -.033 -.028 -.048 -.052	 -.038 -.079 -.074 -.031 -.049 -.073 -.080 -.099	 .032 -.053 -.057 -.089 .158 -.087 -.068	 -.015 -.023 -.032 -.047	 .089 .010 -.011 -.083 .084 .060 -.038	 .084 .037 .054 .017	.128 .130 .128 .119 .096 .075 .047 .132 .049 .039 .027 .067 .017	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989		
$M = 2.75; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = -5^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	-.005 -.013 -.042 -.064 -.007 -.051 -.031 -.056 -.062	 -.093 -.103 -.083 -.041 -.071 -.094 -.099 -.098	 -.001 -.076 -.081 -.110 .166 -.097 -.068	 -.003 -.059 -.070 -.069	 .097 .014 -.012 -.097 .124 .094 -.033	 .109 .058 .073 .039	.161 .163 .164 .153 .127 .106 .073 .176 .070 .064 .042 .085 .073 .039	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989		
$M = 2.75; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = 0^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	-.008 -.035 -.061 -.079 -.009 -.044 -.030 -.058 -.066	 -.077 -.097 -.078 -.040 -.070 -.093 -.097 -.096	 -.006 -.076 -.075 -.105 .145 -.095 -.070	 .184 -.034 -.059 -.070	 .091 .012 -.014 -.094 .124 .094 -.031	 .095 .051 .174 .062 .055 .040 .083 .071 .039	.161 .168 .182 .178 .141 .103 .064 .062 .055 .040 .083 .071 .039	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989		
$M = 2.75; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	-.001 -.023 -.048 -.068 -.001 -.030 -.026 -.049 -.056	 -.041 -.084 -.066 -.032 -.050 -.074 -.083 -.100	 .029 -.055 -.058 -.093 .155 -.090 -.068	 .198 -.020 -.042 -.049	 .084 .006 -.013 -.083 .083 .061 -.037	 .067 .031 .136 .040 .034 .025 .064 .053 .016	.129 .137 .146 .142 .108 .073 .040 .040 .040 .034 .025 .064 .016	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989		

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta = -$						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 2.75; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.006					.102	.095
.124	-.008					.111	.124
.153	-.030					.116	.153
.191	-.051			.217		.109	.191
.267				-.005		.078	.267
.344				-.015		.049	.344
.439				-.025		.023	.439
.538		-.002	.061		.077		.538
.556		-.058	-.034		.001	.100	.556
.592	.013	-.046	-.043		-.013	.043	.592
.668	-.014	-.021	-.066		.011	.010	.668
.744	-.021	-.034	.115		-.063	.043	.744
.821	-.039				.050		.821
.897	-.042	-.054	-.081		.032	.029	.897
.945		-.065					.945
.989		-.091	-.054		-.039	.000	.989

$M = 2.75; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.013					.079	.095
.124	.008					.088	.124
.153	-.010					.088	.153
.191	-.032			.221		.082	.191
.267				.007		.053	.267
.344				-.002		.032	.344
.439				-.011		.013	.439
.538		.029	.089		.069	.070	.538
.556		-.031	-.014		-.005	.000	.556
.592	.030	-.023	-.034		-.012	.001	.592
.668	-.000	-.010	-.040		-.023	-.001	.668
.744	-.013	-.021	.072		-.042	.024	.744
.821	-.026				.023		.821
.897	-.029	-.038	-.070		.005	.008	.897
.945		-.048					.945
.989		-.085	-.046		-.039	-.019	.989

$M = 2.75; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.028					.058	.095
.124	.022					.063	.124
.153	.016					.059	.153
.191	-.007			.231		.052	.191
.267				.017		.028	.267
.344				.007		.017	.344
.439				-.002		.003	.439
.538		.059	.120		.059	.040	.538
.556		-.004	.003		-.014	-.019	.556
.592	.048	.000	-.027		-.016	-.013	.592
.668	.014	.003	-.011		-.022	-.012	.668
.744	-.003	-.005	.030		-.020	.005	.744
.821	-.013				-.005		.821
.897	-.010	-.019	-.053		-.017	-.008	.897
.945		-.033					.945
.989		-.069	-.041		-.037	-.036	.989

$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.045					.043	.095
.124	.047					.043	.124
.153	.035					.037	.153
.191	.023			.237		.028	.191
.267				.024		.014	.267
.344				.011		.010	.344
.439				.001		-.001	.439
.538		.081	.157		.051	.019	.538
.556		.033	.010		-.020	-.034	.556
.592	.063	.015	-.018		-.021	-.020	.592
.668	.026	.019	.016		-.034	-.020	.668
.744	.008	.010	-.004		.005	-.010	.744
.821	.001				-.028		.821
.897	.008	.000	-.036		-.036	-.022	.897
.945		-.017					.945
.989		-.056	-.043		-.039	-.051	.989

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x}{L}$ $\frac{z}{L}$	C_p at $\theta = \text{---}$						Body station $\frac{x}{L}$ $\frac{z}{L}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.75; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.059 .073 .069 .050 .076 .040 .020 .014 .030	 .114 .053 .038 .039 .026 .021 .002 -.039	 .168 .020 -.015 .039 -.035 -.019 -.051	 .240 .024 .010 -.002 				

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x}{L}$		C_p at $\theta =$ —						Body station $\frac{x}{L}$
$\frac{x}{L}$	Body center line	0°		60°	90°	120°	180°	
		Body center line	Nacelle center line				Body center line	Nacelle center line
$M = 2.75; \alpha = -10^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095	.156						-.006	.095
.124	.212						-.029	.124
.153	.245						-.048	.153
.191	.202				.243		-.055	.191
.267					-.017		-.004	.267
.344					-.035		-.008	.344
.439					-.065		-.017	.439
.538		.370	.174			-.048		.538
.556		.248	.016			-.082		.556
.592	.102	.183	-.026			-.099	-.071	.592
.668	.131	.134	.123				-.081	.668
.744	.107	.113	-.115			.139		.744
.821	.101					-.115		.821
.897	.138	.125	.082			-.090		.897
.945		.108					-.094	.945
.989		.044	-.034			-.055	-.085	.989
$M = 2.75; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$								
.095	.012						.008	.095
.124	.014						.006	.124
.153	.003						-.004	.153
.191	-.002				.241		-.014	.191
.267					-.010		-.029	.267
.344					-.011		-.043	.344
.439					-.016		-.053	.439
.538		.108	.072			-.022		.538
.556		.051	-.031			-.061		.556
.592	.028	.024	-.052			-.046	-.040	.592
.668	.013	-.004	.015				-.040	.668
.744	.000	-.031	.000			.004		.744
.821	-.010					-.022		.821
.897	-.004	-.020	-.040			-.032		.897
.945		-.033					-.028	.945
.989		-.099	-.048			-.043	-.067	.989
$M = 2.75; \alpha = 0^\circ; \beta = -6^\circ; \delta_e = 0^\circ$								
.095	.024						.021	.095
.124	.026						.018	.124
.153	.019						.012	.153
.191	.009				.247		.002	.191
.267					-.004		-.014	.267
.344					-.008		-.027	.344
.439					-.013		-.031	.439
.538		.065	.084			-.001		.538
.556		.036	-.029			-.050		.556
.592	.043	.023	-.048			-.038	-.033	.592
.668	.018	.014	.014				-.039	.668
.744	.004	-.017	.000			.005		.744
.821	-.006					-.023		.821
.897	.002	-.012	-.039			-.034		.897
.945		-.026					-.023	.945
.989		-.094	-.043			-.039	-.060	.989
$M = 2.75; \alpha = 0^\circ; \beta = -4^\circ; \delta_e = 0^\circ$								
.095	.036						.030	.095
.124	.034						.030	.124
.153	.039						.026	.153
.191	.018				.255		.016	.191
.267					.003		-.001	.267
.344					-.004		-.007	.344
.439					-.011		-.014	.439
.538		.086	.121			.015		.538
.556		.034	-.018			-.042	-.043	.556
.592	.057	.023	-.035			-.031	-.026	.592
.668	.022	.028	.013				-.041	.668
.744	.009	.000	-.001			.010		.744
.821	-.001					-.028		.821
.897	.008	-.001	-.037			-.038	-.022	.897
.945		-.012						.945
.989		-.075	-.043			-.037	-.056	.989

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TABLE IV.- PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x}{L}$ $\frac{z}{L}$		C_p at $\theta =$ —						Body station $\frac{x}{L}$ $\frac{z}{L}$		
		0°		60°	90°	120°	180°			
		Body center line	Nacelle center line				Body center line			Nacelle center line
$M = 2.75; \alpha = 0^\circ; \beta = -2^\circ; \delta_e = 0^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.043 .043 .043 .022				.259 .013 .001 -.008		.038 .037 .031 .025 .005 .003 -.005	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989		
		.093 .033 .022 .030 .012	.145 -.005 -.025 .017 -.005			.030 -.035 -.028		.033 -.043 -.024 -.037 -.018		
	.063 .028 .010 .005 .013					.012 -.030 -.042				
		.002 -.007 -.067	-.036 -.049					-.025 -.057		
$M = 2.75; \alpha = 0^\circ; \beta = -1^\circ; \delta_e = 0^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.045 .050 .038 .025				.264 .019 .006 -.006		.040 .040 .033 .023 .009 .007 -.002	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989		
		.097 .037 .021 .029 .013	.153 .002 -.021 .018 -.007			.038 -.030 -.027		.025 -.041 -.024 -.030 -.017		
	.065 .030 .011 .005 .013					.013 -.031 -.045				
		.004 -.009 -.061	-.034 -.050					-.027 -.057		
$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.046 .050 .038 .026				.268 .025 .011 -.003		.040 .040 .033 .023 .008 .008 -.001	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989		
		.092 .041 .020 .026 .014	.164 .009 -.018 .023 -.012			.046 -.025 -.024		.015 -.039 -.023 -.024 -.016		
	.065 .030 .012 .005 .014					.012 -.031 -.045				
		.006 -.011 -.052	-.031 -.049					-.028 -.058		
$M = 2.75; \alpha = 0^\circ; \beta = 1^\circ; \delta_e = 0^\circ$										
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.046 .050 .039 .025				.274 .031 .017 .000		.040 .040 .033 .026 .008 .007 -.002	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989		
		.093 .041 .019 .020 .014	.172 .019 -.008 .026 -.017			.054 -.020 -.022		.006 -.038 -.022 -.021 -.014		
	.065 .029 .012 .005 .014					.011 -.032 -.047				
		.005 -.013 -.043	-.030 -.050					-.033 -.059		

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{z_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{z_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.75; \alpha = 0^\circ; \beta = 2^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.044 .044 .042 .022 .063 .027 .010 .004 .013 	 .100 .040 .019 .013 .013 .004 -.014 -.035	 .183 .026 -.001 .030 -.021 -.029 -.049	 .277 .035 .021 .004 	 .062 -.016 -.019 .010 -.034 -.047 -.049	 -.004 -.041 -.023 -.020 -.012 -.039 -.061	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 2.75; \alpha = 0^\circ; \beta = 4^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.036 .034 .036 .019 .056 .021 .009 .000 .007	 .122 .039 .025 -.004 .008 -.001 -.017 -.027	 .212 .047 .012 .040 -.028 -.026 -.048	 .287 .047 .035 .016 	 .081 -.003 -.010 .012 -.036 -.047 -.051	 -.034 -.040 -.018 -.047 -.029 -.022 -.010 -.050 -.066	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 2.75; \alpha = 0^\circ; \beta = 6^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.024 .027 .018 .011 .044 .015 .004 -.007 .002	 .126 .042 .025 -.014 -.002 -.008 -.018 -.021	 .249 .072 .025 .052 -.033 -.021 -.051	 .299 .062 .050 .032 	 .106 .011 .002 .018 -.042 -.046 -.045	 -.020 -.017 .009 .000 -.015 -.027 -.031 -.028 -.056 -.045 -.037 -.013 -.058 -.081	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 2.75; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	 .010 .000 -.013 -.007	 .033 -.022 -.031 -.016 -.023 -.018	 .046 .070 -.026 -.018 -.056	 .016 .031 -.037 -.051 -.045	 -.046 -.036 	 -.067 -.059 -.027 -.062 -.099	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —							Body station $\frac{x_M}{L_M}$
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line	Nacelle center line	
$M = 2.75; \alpha = -10^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.150					-.002		.095
.124	.179					-.010		.124
.153	.185					-.028		.153
.191	.161			-.035		-.039		.191
.267				-.029		-.011		.267
.344				-.042		-.013		.344
.439				-.063		-.020		.439
.538		.373	.184		-.040		-.089	.538
.556		.252	.020		-.077		-.086	.556
.592	.106	.188	-.023		-.092	-.072	-.066	.592
.668	.134	.137	.126			-.069	-.062	.668
.744	.109	.116	-.098		.141		-.067	.744
.821	.103				-.097			.821
.897	.139	.127	.078		-.085		-.085	.897
.945		.106						.945
.989		.042	-.035		-.061		-.080	.989
$M = 2.75; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.123					.003		.095
.124	.146					-.001		.124
.153	.148					-.019		.153
.191	.126			-.012		-.031		.191
.267				-.013		-.007		.267
.344				-.029		-.007		.344
.439				-.046		-.014		.439
.538		.324	.192		-.023		-.083	.538
.556		.211	.025		-.071		-.084	.556
.592	.113	.152	-.019		-.085	-.071	-.066	.592
.668	.111	.120	.107			-.065	-.057	.668
.744	.086	.095	-.105		.110		-.059	.744
.821	.080				-.100			.821
.897	.110	.101	.056		-.079		-.085	.897
.945		.078						.945
.989		.021	-.047		-.053		-.080	.989
$M = 2.75; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.096					.008		.095
.124	.112					.011		.124
.153	.108					.000		.153
.191	.089			.007		-.015		.191
.267				.001		-.005		.267
.344				-.013		-.004		.344
.439				-.026		-.011		.439
.538		.275	.207		.002		-.072	.538
.556		.161	.031		-.058		-.082	.556
.592	.088	.114	-.011		-.069	-.063	-.056	.592
.668	.088	.089	.089			-.060	-.049	.668
.744	.064	.071	-.092		.079		-.045	.744
.821	.056				-.089			.821
.897	.081	.071	.030		-.073		-.071	.897
.945		.051						.945
.989		.000	-.059		-.044		-.078	.989
$M = 2.75; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.074					.016		.095
.124	.084					.020		.124
.153	.077					.019		.153
.191	.057			.014		.004		.191
.267				.015		.000		.267
.344				-.001		-.002		.344
.439				-.013		-.009		.439
.538		.228	.214		.011		-.018	.538
.556		.122	.033		-.050		-.075	.556
.592	.100	.083	-.009		-.053	-.050	-.043	.592
.668	.068	.066	.072			-.056	-.039	.668
.744	.043	.051	-.069		.058		-.040	.744
.821	.037				-.074			.821
.897	.057	.046	.007		-.070		-.058	.897
.945		.026						.945
.989		-.019	-.061		-.043		-.070	.989

TABLE IV.- PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{z_M}$	C_p at $\theta = \text{---}$						Body station $\frac{x_M}{z_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.75; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.055					.031	.095	
.124	.057					.033	.124	
.153	.045					.036	.153	
.191	.025			.017		.027	.191	
.267				.019		.009	.267	
.344				.007		.001	.344	
.439				-.004		-.008	.439	
.538		.178	.201		.012		.538	
.556		.083	.031		-.043	-.007	.556	
.592	.086	.055	-.014		-.039	-.035	.592	
.668	.050	.045	.054			-.048	.668	
.744	.027	.034	-.038		.032	-.028	.744	
.821	.021				-.054		.821	
.897	.033	.023	-.015		-.056	-.042	.897	
.945		.005					.945	
.989		-.036	-.055		-.041	-.060	.989	
$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.039					.044	.095	
.124	.031					.055	.124	
.153	.017					.055	.153	
.191	-.002			.014		.054	.191	
.267				.022		.019	.267	
.344				.010		.005	.344	
.439				-.001		-.006	.439	
.538		.140	.139		.039		.538	
.556		.054	-.002		-.029	-.041	.556	
.592	.074	.033	-.035		-.027	-.023	.592	
.668	.035	.025	.028			-.040	.668	
.744	.011	.016	-.007		.003	-.015	.744	
.821	.004				-.033		.821	
.897	.012	.002	-.039		-.041	-.026	.897	
.945		-.015					.945	
.989		-.052	-.050		-.033	-.052	.989	
$M = 2.75; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.022					.058	.095	
.124	.009					.077	.124	
.153	-.007					.076	.153	
.191	-.025			.017		.079	.191	
.267				.022		.033	.267	
.344				.009		.014	.344	
.439				-.002		-.000	.439	
.538		.097	.121		.051		.538	
.556		.028	-.011		-.020	.036	.556	
.592	.056	.009	-.028		-.020	-.023	.592	
.668	.020	.006	-.004			-.004	.668	
.744	-.002	-.001	.028		-.020	-.025	.744	
.821	-.011				-.007	.001	.821	
.897	-.007	-.016	-.057		-.020	-.012	.897	
.945		-.030					.945	
.989		-.068	-.045		-.038	-.038	.989	
$M = 2.75; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.009					.077	.095	
.124	-.012					.101	.124	
.153	-.030					.099	.153	
.191	-.048			.017		.108	.191	
.267				.016		.053	.267	
.344				.001		.028	.344	
.439				-.009		.005	.439	
.538		.024	.083		.062		.538	
.556		-.010	-.017		-.010	.064	.556	
.592	.035	-.015	-.034		-.018	-.005	.592	
.668	.003	-.005	-.034			-.009	.668	
.744	-.015	-.017	.067		-.039	.020	.744	
.821	-.026				.022		.821	
.897	-.027	-.037	-.069		.002	.005	.897	
.945		-.047					.945	
.989		-.083	-.048		-.039	-.021	.989	

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095		.002					.095	
.124		-.028					.124	
.153		-.047					.153	
.191		-.065			.012		.191	
.267					.008		.267	
.344					-.012		.344	
.439					-.022		.439	
.538			.054				.538	
.556		-.050	-.035		.073		.556	
.592		-.034	-.044		-.002		.592	
.668	.014	-.018	-.062		-.016	.040	.668	
.744	-.023	-.034	.114			.009	.744	
.821	-.040				-.061	.040	.821	
.897	-.042	-.056	-.080		.051		.897	
.945		-.065			.030		.945	
.989		-.091	-.056		-.039	-.001	.989	
$M = 2.75; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	-.014					.163	.095	
.124	-.055					.193	.124	
.153	-.077					.195	.153	
.191	-.096			-.015		.213	.191	
.267				-.021		.142	.267	
.344				-.066		.100	.344	
.439				-.077		.059	.439	
.538		-.070	-.013		.087		.538	
.556		-.096	-.078		.008	.172	.556	
.592	-.013	-.075	-.073		-.018	.060	.592	
.668	-.044	-.043	-.109			.093	.668	
.744	-.032	-.072	.131			.050	.744	
.821	-.060				-.098	.081	.821	
.897	-.068	-.092	-.100		.125		.897	
.945		-.094			.095	.068	.945	
.989		-.099	-.074		-.033	.031	.989	
$M = 3.22; \alpha = -10^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095	.143					-.004	.095	
.124	.218					-.018	.124	
.153	.253					-.029	.153	
.191	.233			.009		-.039	.191	
.267				-.005		-.019	.267	
.344				-.023		-.004	.344	
.439				-.037		-.011	.439	
.538		.340	.177		-.019		.538	
.556		.222	.031		-.052	-.047	.556	
.592	.066	.174	-.011		-.064	-.059	.592	
.668	.105	.110	.106			-.049	.668	
.744	.094	.090	-.061			-.054	.744	
.821	.083			.118			.821	
.897	.113	.113	.071	-.063			.897	
.945		.091		-.054		-.052	.945	
.989		.045	-.016	-.055		-.064	.989	
$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095	.120					-.006	.095	
.124	.183					-.020	.124	
.153	.213					-.034	.153	
.191	.193			.012		-.044	.191	
.267				.003		-.025	.267	
.344				-.008		-.002	.344	
.439				-.030		-.006	.439	
.538		.291	.171		-.034		.538	
.556		.184	.027		-.063	-.017	.556	
.592	.074	.145	-.014		-.069	-.059	.592	
.668	.092	.097	.091			-.054	.668	
.744	.079	.077	-.073		.095		.744	
.821	.064			.095			.821	
.897	.089	.087	.060	-.075			.897	
.945		.072		-.062		-.058	.945	
.989		.032	-.023	-.049		-.064	.989	

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$		C_p at $\theta = -$						Body station $\frac{x_M}{L_M}$	
$\frac{z_M}{L_M}$		0°		60°	90°	120°	180°		
		Body center line	Nacelle center line				Body center line	Nacelle center line	
$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$									
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.095 .144 .170 .150						-.002 -.011 -.025 -.036 -.025 .000 -.002	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
				.014 .015 .004 -.013					
		.212 .124 .102 .077 .057	.152 .015 -.017 .069 -.072			-.010 -.051 -.059		-.023 -.047 -.042 -.032 -.044	
	.076 .069 .059	.077 .057				.070 -.073 -.062		-.032 -.032 -.038	
	.065 .065	.063 .046 .013	.037					-.057	
			-.034			-.042		-.066	
$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$									
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.074 .112 .133 .114			.012 .022 .012 -.002			.004 -.004 -.014 -.024 -.023 .002 .001	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
		.122 .077 .058 .049 .041 .028 .045	.134 .014 -.017 .054 -.062			.058 -.028 -.039 .050 -.065 -.058		-.018 -.041 -.033 -.026 -.027 -.047	
	.065 .049 .041 .028 .045	.050 .038						-.063	
		.040 .027 -.003	.019			-.036			
			-.043						
$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$									
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989								.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
		.033 .033 .028	-.005 .035 -.043			-.024 .037 -.048 -.051	-.016 -.033	-.024 -.017 -.012 -.038 -.057	
	.036 .028 .019 .031	.025 .008 -.018							
			-.043			-.039			
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = -5^\circ$									
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.046 .058 .074 .052			.009 .022 .016 .006			.030 .024 .017 .007 -.002 .004 .010	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x}{L}$ $\frac{z}{L}$	C_p at $\theta =$ —						Body station $\frac{x}{L}$ $\frac{z}{L}$
	0°		60°	90°	120°	180°	
	Body center line	Nacelle center line				Body center line	Nacelle center line
$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.032 .036 .048 .027			.003 .017 .013 .003		.044 .041 .034 .024 .014 .011 .013	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
		.074 -.003 -.008 .012 .006 -.003 .001	.131 .021 -.009 -.001 .016 -.025 -.019 -.041		.068 -.002 -.009 -.008 -.004 -.019 -.036	.034 -.007 -.012 -.002 .004 -.013 -.036	
$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.017 .022 .030 .007			-.007 .010 .007 -.004		.065 .063 .058 .048 .035 .026 .021	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
	.027 .002 -.004 -.013 -.013	.045 -.022 -.026 -.010 -.013	.103 .003 -.017 -.021 .057		.078 .007 -.004 -.031 .023 .004 -.037	.060 .011 .001 .006 .018 -.002 -.023	
$M = 3.22; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.006 .013 .011 -.013			-.010 -.004 -.003 -.016		.087 .086 .081 .073 .058 .046 .034	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
	.017 -.010 -.012 -.021 -.023	.013 -.041 -.045 -.019 -.022	.076 -.015 -.026 -.040 .098		.085 .013 -.002 -.047 .052 .028 -.039	.091 .032 .015 .016 .036 .014 -.006	
$M = 3.22; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$							
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.000 .003 -.005 -.027			-.006 -.023 -.024 -.028		.114 .112 .109 .101 .085 .071 .055	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989
	.007 -.012 -.019 -.029 -.032	-.021 -.057 -.057 -.026 -.034	.052 -.031 -.039 -.056 .125		.094 .020 .002 -.062 .084 .055 -.038	.126 .056 .032 .031 .052 .035 .012	

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 3.22; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = -5^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	-.007 -.009 -.018 -.043 -.006 -.034 -.026 -.037 -.040	 -.052 -.074 -.069 -.038 -.048 -.064 -.066 -.069	 .025 .047 .052 .071 .139 .059 .055	 .005 -.035 -.046 -.053	 .104 .026 .005 -.073 .119 .084 -.036	.146 .141 .140 .132 .115 .100 .082 .160 .080 .053 .049 .076 .054 .032	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 3.22; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	-.010 -.020 -.035 -.055 -.005 -.032 -.028 -.037 -.044	 -.055 -.072 -.064 -.039 -.049 -.072 -.069 -.066	 .018 .050 .054 .071 .141 .057 .055	 .425 -.042 -.055 -.061	 .100 .026 .005 -.070 .123 .091 -.032	.152 .146 .153 .154 .133 .109 .083 .161 .076 .053 .050 .079 .060 .035	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 3.22; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	-.006 -.015 -.031 -.051 .001 -.026 -.022 -.032 -.038	 -.039 -.070 -.060 -.032 -.040 -.058 -.063 -.068	 .039 .039 .048 .070 .127 .059 .053	 .432 -.021 -.040 -.039	 .092 .019 .002 -.070 .093 .066 -.037	.122 .119 .123 .123 .104 .081 .058 .131 .057 .036 .034 .056 .042 .016	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	
$M = 3.22; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.000 -.006 -.019 -.040 .011 -.014 -.017 -.026 -.030	 -.002 -.054 -.047 -.022 -.029 -.042 -.049 -.063	 .065 .024 .036 .058 .103 .055 .046	 .439 -.006 -.018 -.022	 .083 .013 -.002 -.060 .061 .040 -.037	.094 .093 .097 .093 .076 .056 .036 .097 .036 .018 .019 .038 .022 -.001	.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x}{L}$ $\frac{z}{L}$	C_p at $\theta = -$						Body station $\frac{x}{L}$ $\frac{z}{L}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.008 .004 -.002 -.024 .025 .000 -.006 -.016 -.017 .039 -.028 -.021 -.009 -.016 -.025 -.036 -.057	 .093 -.004 -.022 -.032 .064 -.048 -.041	 <					

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x}{L_M}$	C_p at $\theta =$ —						Body station $\frac{x}{L_M}$
	0°		60°	90°	120°	180°	
	Body center line	Nacelle center line				Body center line	Nacelle center line

$M = 3.22; \alpha = -40; \beta = 00; \delta_e = 00$

.095	.064					.013	.095
.124	.079					.010	.124
.153	.082					.005	.153
.191	.076			.014		-.004	.191
.267				.015		-.010	.267
.344				.010		.001	.344
.439				.000		-.001	.439
.538		.178	.167		.040		.538
.556		.093	.029		-.024	-.017	.556
.592	.074	.076	-.008		-.029	-.038	.592
.668	.050	.052	.054		-.024	-.029	.668
.744	.040	.033	-.046		-.038	-.022	.744
.821	.027				.048	-.021	.821
.897	.044	.040	.010		-.048		.897
.945		.023			-.045	-.037	.945
.989		-.007	-.044		-.035	-.055	.989

$M = 3.22; \alpha = -60; \beta = 00; \delta_e = 00$

.095	.085					.004	.095
.124	.107					.000	.124
.153	.110					-.006	.153
.191	.105			.011		-.017	.191
.267				.009		-.014	.267
.344				.003		-.002	.344
.439				-.010		-.003	.439
.538		.242	.186		.006		.538
.556		.141	.033		-.038	-.033	.556
.592	.078	.110	-.008		-.042	-.051	.592
.668	.069	.074	.071			-.035	.668
.744	.057	.052	-.060		.066	-.032	.744
.821	.043				-.061	-.031	.821
.897	.063	.061	.028		-.053		.897
.945		.043				-.047	.945
.989		.007	-.040		-.037	-.060	.989

$M = 3.22; \alpha = -80; \beta = 00; \delta_e = 00$

.095	.113					-.002	.095
.124	.141					-.007	.124
.153	.144					-.017	.153
.191	.144			.002		-.029	.191
.267				-.002		-.017	.267
.344				-.009		-.005	.344
.439				-.025		-.005	.439
.538		.296	.194		-.005		.538
.556		.189	.039		-.044	-.051	.556
.592	.080	.150	-.009		-.055	-.060	.592
.668	.091	.098	.091			-.044	.668
.744	.080	.077	-.074		.093	-.039	.744
.821	.065				-.072	-.040	.821
.897	.089	.089	.052		-.060		.897
.945		.066				-.056	.945
.989		.028	-.029		-.041	-.064	.989

$M = 3.22; \alpha = -100; \beta = 00; \delta_e = 00$

.095	.144					-.010	.095
.124	.180					-.018	.124
.153	.181					-.030	.153
.191	.186			-.013		-.042	.191
.267				-.021		-.021	.267
.344				-.026		-.009	.344
.439				-.042		-.013	.439
.538		.254	.188		-.021		.538
.556		.233	.036		-.054	-.068	.556
.592	.073	.184	-.008		-.070	-.068	.592
.668	.118	.120	.113			-.052	.668
.744	.105	.100	-.084		.123	-.059	.744
.821	.089				-.083		.821
.897	.118	.116	.080		-.067	-.067	.897
.945		.098					.945
.989		.050	-.014		-.054	-.066	.989

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x}{L}$ $\frac{z}{L}$	C_p at $\theta =$						Body station $\frac{x}{L}$ $\frac{z}{L}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 3.22; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$

.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.021 .015 .015 .003 .029 .012 .008 .000 .004 	 .080 .041 .027 .013 -.009 -.008 -.019 -.062	 .090 -.011 -.035 .008 .002 -.025 -.037	 <
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TABLE IV.- PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.037 .039 .037 .022 .059 .028 .017 .008 .016 	 .099 .046 .027 .024 .014 -0.003 -0.035	 .153 .022 -0.012 .024 -0.015 -0.019 -0.040	 .014 .014 .008 .000 -0.017 -0.019 -0.025 -0.032 -0.033	 			

TABLE IV.- PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{l_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{l_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 3.22; \alpha = 0^\circ; \beta = 6^\circ; \delta_e = 0^\circ$

.095	.023					.019	.095
.124	.018					.013	.124
.153	.020					.011	.153
.191	.008			.063		.001	.191
.267				.059		-.007	.267
.344				.045		-.017	.344
.439				.035		-.024	.439
.538		.143	.250		.114		-.013
.556		.057	.087		.022		-.038
.592	.035	.049	.033		.008	-.028	-.035
.668	.014	-.003	.057			-.031	-.029
.744	.010	-.008	-.020		.031		-.017
.821	.002				-.032		
.897	.006	-.002	-.010		-.035		
.945		-.009					-.038
.989		-.012	-.042		-.037		-.062

$M = 3.22; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.095	.013					.008	.095
.124	.009					.003	.124
.153	.003					-.002	.153
.191	-.008			.090		-.016	.191
.267				.078		-.024	.267
.344				.065		-.031	.344
.439				.053		-.041	.439
.538		.158	.291		.143		-.016
.556		.041	.119		.039		-.041
.592	.013	.054	.055		.022	-.040	-.045
.668	.005	-.006	.079			-.033	-.044
.744	.000	-.022	-.012		.045		-.040
.821	-.008				-.024		
.897	-.004				-.031		
.945		-.018	-.001				-.045
.989		-.012	-.043		-.033		-.070

$M = 3.22; \alpha = 5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.095	-.006					.060	.095
.124	-.008					.054	.124
.153	-.011					.054	.153
.191	-.021			.057		.038	.191
.267				.052		.019	.267
.344				.047		.010	.344
.439				.042		.002	.439
.538		.088	.229		.188		.059
.556		.001	.076		.076		.014
.592	-.010	-.019	.034		.053	.003	.002
.668	-.022	-.036	.013			.006	-.003
.744	-.016	-.037	.070		-.006		.043
.821	-.026				.035		
.897	-.022				.013		
.945		-.044	-.019				.000
.989		-.038	-.046		-.022		-.047

$M = 3.22; \alpha = 5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$

.095	-.012					.067	.095
.124	-.019					.063	.124
.153	-.024					.060	.153
.191	-.035			.043		.046	.191
.267				.036		.033	.267
.344				.035		.019	.344
.439				.028		.009	.439
.538		.062	.185		.155		.059
.556		-.021	.051		.058		.016
.592	.009	-.033	.020		.043	.016	.007
.668	-.022	-.055	-.005			.006	.010
.744	-.025	-.049	.069		-.027		.050
.821	-.033				.034		
.897	-.021				.020		
.945		-.040	-.035				.003
.989		-.047	-.050		-.023		-.047

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{l_M}$	C_p at $\theta =$ —						Body station $\frac{x_M}{l_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 3.22; \alpha = 5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$

.095	-.006					.074	.095
.124	-.012					.074	.124
.153	-.020					.067	.153
.191	-.032			.026		.065	.191
.267				.021		.046	.267
.344				.022		.030	.344
.439				.014		.016	.439
.538		.044	.147		.129	.065	.538
.556		-.030	.029		.042	.020	.556
.592	.015	-.040	.004		.029	.026	.592
.668	-.012	-.049	-.019			.004	.668
.744	-.018	-.034	.069		-.037	.045	.744
.821	-.026				.035		.821
.897	-.025	-.033	-.042		.021	.003	.897
.945		-.041	-.047		-.031	-.045	.945
.989		-.051					.989

$M = 3.22; \alpha = 5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$

.095	-.002					.076	.095
.124	-.006					.079	.124
.153	-.014					.075	.153
.191	-.030			.014		.074	.191
.267				.015		.055	.267
.344				.014		.037	.344
.439				.004		.019	.439
.538		.033	.118		.107	.071	.538
.556		-.034	.013		.028	.022	.556
.592	.018	-.039	-.008		.018	.032	.592
.668	-.008	-.035	-.027			.004	.668
.744	-.015	-.027	.072		-.041	.038	.744
.821	-.015				.038		.821
.897	-.024	-.033	-.043		.022	.004	.897
.945	-.025	-.041	-.043		-.033	-.035	.945
.989		-.054					.989

$M = 3.22; \alpha = 5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$

.095	-.001					.078	.095
.124	-.002					.080	.124
.153	-.012					.080	.153
.191	-.031			.003		.076	.191
.267				.008		.060	.267
.344				.004		.039	.344
.439				-.006		.022	.439
.538		.023	.096		.090	.075	.538
.556		-.038	-.002		.017	.022	.556
.592	.017	-.039	-.018		.007	.036	.592
.668	-.007	-.024	-.035			.006	.668
.744	-.014	-.023	.079		-.045	.033	.744
.821	-.023				.041		.821
.897	-.025	-.034	-.046		.022	.002	.897
.945		-.040	-.042		-.035	-.023	.945
.989		-.057					.989

$M = 3.22; \alpha = 5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.095	.003					.079	.095
.124	.001					.081	.124
.153	-.010					.082	.153
.191	-.028			-.002		.078	.191
.267				.005		.061	.267
.344				-.001		.041	.344
.439				-.011		.025	.439
.538		.023	.083		.079	.079	.538
.556		-.037	-.010		.010	.022	.556
.592	.019	-.034	-.022		-.001	.039	.592
.668	-.005	-.015	-.037			.006	.668
.744	-.011	-.020	.084		-.044	.029	.744
.821	-.021				.042		.821
.897	-.023	-.033	-.048		.022	.007	.897
.945		-.038	-.042		-.035	-.015	.945
.989		-.058					.989

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta = \text{---}$						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 3.22; \alpha = 5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$								
.095 .124 .153 .191 .267 .344 .439 .538 .556 .592 .668 .744 .821 .897 .945 .989	.002 .001 -.009 -.029 .018 -.005 -.011 -.021 -.023 <							

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TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta = \text{---}$						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 3.22; \alpha = 5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$								
.095	-.018					.066		.095
.124	-.029					.060		.124
.153	-.034					.059		.153
.191	-.047				-.012	.046		.191
.267					-.018	.027		.267
.344					-.023	.021		.344
.439					-.030	.014		.439
.538		.029	.000				.084	.538
.556		-.014	-.052		-.005		.002	.556
.592	-.001	-.017	-.044		-.042		-.006	.592
.668	-.038	-.016	-.037		-.046	.021	-.031	.668
.744	-.030	-.038	.003			.014	-.024	.744
.821	-.035				-.045			.821
.897	-.033	-.045	-.056		.004			.897
.945		-.053			.001		.023	.945
.989		-.069	-.046		-.025		-.015	.989
$M = 3.22; \alpha = -10^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.147					-.003		.095
.124	.155					-.006		.124
.153	.164					-.012		.153
.191	.154				-.033	-.027		.191
.267					-.033	-.013		.267
.344					-.024	-.009		.344
.439					-.038	-.014		.439
.538		.397	.201				-.060	.538
.556		.247	.048		-.013		-.061	.556
.592	.085	.189	-.003		-.050		-.046	.592
.668	.125	.127	.117		-.062	-.049	-.046	.668
.744	.114	.102	-.069			-.052	-.044	.744
.821	.093				.127		-.047	.821
.897	.119	.120	.083		-.069			.897
.945		.101			-.051		-.057	.945
.989		.054	-.009		-.049		-.059	.989
$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.118					-.002		.095
.124	.126					-.002		.124
.153	.131					-.007		.153
.191	.118				-.023	-.024		.191
.267					-.016	-.012		.267
.344					-.018	-.008		.344
.439					-.032	-.010		.439
.538		.330	.209				-.063	.538
.556		.205	.049		-.001		-.061	.556
.592	.070	.154	-.002		-.046	-.053	-.049	.592
.668	.102	.109	-.098		-.065	-.053	-.045	.668
.744	.089	.082	-.080				-.044	.744
.821	.069				.099			.821
.897	.093	.093	.061		-.079		-.060	.897
.945		.073			-.054			.945
.989		.036	-.021		-.046		-.063	.989
$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.091					.004		.095
.124	.095					.006		.124
.153	.097					.005		.153
.191	.084				-.005	-.011		.191
.267					-.004	-.010		.267
.344					-.007	-.005		.344
.439					-.017	-.007		.439
.538		.272	.206				-.054	.538
.556		.159	.045		.013		-.061	.556
.592	.061	.119	.001		-.040	-.049	-.044	.592
.668	.080	.082	.081		-.058	-.051	-.040	.668
.744	.069	.058	-.078				-.039	.744
.821	.049				.072			.821
.897	.070	.066	.038		-.075		-.054	.897
.945		.049			-.053			.945
.989		.017	-.033		-.040		-.061	.989

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Continued

Body station $\frac{x_M}{L_M}$	C_p at $\theta = -$						Body station $\frac{x_M}{L_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line

$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$

.095	.069					.012		.095
.124	.070					.013		.124
.153	.068					.018		.153
.191	.056			.001		.004		.191
.267				.008		-.005		.267
.344				.004		-.002		.344
.439				-.006		-.004		.439
.538		.226	.196		.022		-.014	.538
.556		.117	.050		-.031		-.049	.556
.592	.051	.088	.002		-.044	-.036	-.037	.592
.668	.064	.053	.065			-.047	-.031	.668
.744	.054	.042	-.063		.053		-.030	.744
.821	.034				-.064			.821
.897	.050	.043	.019		-.052		-.046	.897
.945		.028						.945
.989		-.002	-.042		-.037		-.057	.989

$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$

.095	.048					.022		.095
.124	.048					.024		.124
.153	.044					.023		.153
.191	.030			.007		.021		.191
.267				.013		.003		.267
.344				.011		.000		.344
.439				.000		-.003		.439
.538		.179	.182		.024		-.008	.538
.556		.083	.042		-.029		-.039	.556
.592	.072	.062	-.005		-.032	-.027	-.032	.592
.668	.046	.035	.051			-.042	-.025	.668
.744	.034	.027	-.040		.031		-.022	.744
.821	.020				-.048			.821
.897	.031	.022	.000		-.045		-.036	.897
.945		.010						.945
.989		-.016	-.046		-.037		-.051	.989

$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 5^\circ$

.095	.033					.035		.095
.124	.029					.041		.124
.153	.022					.043		.153
.191	.007			.010		.037		.191
.267				.018		.017		.267
.344				.012		.008		.344
.439				.004		.000		.439
.538		.141	.121		.045		.011	.538
.556		.058	.016		-.019		-.025	.556
.592	.063	.040	-.015		-.020	-.012	-.024	.592
.668	.035	.019	.031			-.031	-.019	.668
.744	.021	.014	-.009		.009		-.011	.744
.821	.008				-.022			.821
.897	.017	.007	-.017		-.031		-.024	.897
.945		-.004						.945
.989		-.028	-.045		-.035		-.044	.989

$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$

.095	.020					.048		.095
.124	.011					.060		.124
.153	.003					.062		.153
.191	-.012			.009		.058		.191
.267				.020		.032		.267
.344				.011		.017		.344
.439				.002		.004		.439
.538		.086	.111		.056		.033	.538
.556		.035	.005		-.009		-.010	.556
.592	.049	.014	-.016		-.013	.004	-.016	.592
.668	.021	.009	.005			-.020	-.010	.668
.744	.008	.003	.020		-.010		-.001	.744
.821	-.003				-.002			.821
.897	.002	-.009	-.031		-.016		-.014	.897
.945		-.017						.945
.989		-.040	-.040		-.036		-.035	.989

TABLE IV. - PRESSURE COEFFICIENTS ON MISSILE BODY - Concluded

Body station $\frac{x_M}{l_M}$	C_p at $\theta = \text{---}$						Body station $\frac{x_M}{l_M}$	
	0°		60°	90°	120°	180°		
	Body center line	Nacelle center line				Body center line		Nacelle center line
$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	.009					.067	.095	
.124	-.001					.083	.124	
.153	-.012					.085	.153	
.191	-.029			.006		.088	.191	
.267				.016		.052	.267	
.344				.004		.030	.344	
.439				-.005		.013	.439	
.538		.041	.088		.067		.538	
.556		-.008	-.005		.001	.061	.556	
.592	.031	-.006	-.022		-.008	.023	.592	
.668	.008	-.003	-.018			-.005	.668	
.744	-.002	-.010	.058		-.026	.015	.744	
.821	-.014				.025		.821	
.897	-.014	-.026	-.039		.005		.897	
.945		-.030				-.001	.945	
.989		-.051	-.038		-.036	-.023	.989	
$M = 3.22; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	-.001					.091	.095	
.124	-.011					.105	.124	
.153	-.025					.110	.153	
.191	-.042			.000		.120	.191	
.267				.007		.077	.267	
.344				-.011		.052	.344	
.439				-.016		.028	.439	
.538		.009	.065		.078		.538	
.556		-.037	-.020		.008	.094	.556	
.592	.015	-.033	-.029		-.005	.029	.592	
.668	-.005	-.014	-.039			.043	.668	
.744	-.013	-.020	.097		-.043	.010	.744	
.821	-.025				.053	.013	.821	
.897	-.026	-.039	-.047		.030	.032	.897	
.945		-.042				.017	.945	
.989		-.057	-.041		-.036	-.007	.989	
$M = 3.22; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	-.010					.117	.095	
.124	-.024					.124	.124	
.153	-.038					.138	.153	
.191	-.056			-.016		.149	.191	
.267				-.009		.105	.267	
.344				-.026		.074	.344	
.439				-.034		.046	.439	
.538		-.025	.038		.085		.538	
.556		-.057	-.038		.013	.124	.556	
.592	.000	-.052	-.042		-.004	.048	.592	
.668	-.021	-.027	-.060			.065	.668	
.744	-.022	-.034	.120		-.060	.028	.744	
.821	-.034				.086	.025	.821	
.897	-.038	-.055	-.056		.055	.048	.897	
.945		-.056				.035	.945	
.989		-.067	-.052		-.038	.009	.989	
$M = 3.22; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = 5^\circ$								
.095	-.012					.151	.095	
.124	-.030					.158	.124	
.153	-.046					.169	.153	
.191	-.065			-.030		.186	.191	
.267				-.028		.141	.267	
.344				-.046		.106	.344	
.439				-.052		.074	.439	
.538		-.048	.019		.095		.538	
.556		-.072	-.049		.021	.158	.556	
.592	-.005	-.062	-.053		.002	.072	.592	
.668	-.031	-.036	-.073			.093	.668	
.744	-.028	-.047	.138		-.069	.051	.744	
.821	-.036				.120		.821	
.897	-.044	-.069	-.058		.087	.058	.897	
.945		-.066					.945	
.989		-.067	-.055		-.033	.032	.989	

TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$$M = 2.29; \alpha = -10^\circ; \beta = 0^\circ; \delta_e = -5^\circ$$

.037	.377	-.145	.048	.479	-.082	.069	.554	-.039	.100	.517	-.116
.200	.244	-.178	.200	.323	-.082	.200	.427	-.071	.200	.455	-.108
.450	.187	-.175	.450	.219	-.168	.450	.283	-.122	.450	.326	-.092
.750	.124	-.147	.750			.750	.168	-.160	.750	.208	-.127
.963	.105	-.145	.953	.119	-.159	.931	.143	-.157	.901	.174	-.130

$$M = 2.29; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$$

.037	.318	-.145	.048	.397	-.072	.069	.455	-.016	.100	.429	-.081
.200	.192	-.181	.200	.270	-.072	.200	.361	-.053	.200	.387	-.070
.450	.146	-.151	.450	.168	-.162	.450	.230	-.112	.450	.274	-.068
.750	.087	-.132	.750			.750	.121	-.155	.750	.164	-.116
.963	.071	-.126	.953	.082	-.157	.931	.105	-.154	.901	.129	-.118

$$M = 2.29; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$$

.037	.263	-.111	.048	.301	-.042	.069	.359	-.017	.100	.323	-.042
.200	.146	-.129	.200	.220	-.042	.200	.286	-.021	.200	.308	-.036
.450	.109	-.119	.450	.121	-.133	.450	.180	-.085	.450	.216	-.043
.750	.054	-.116	.750			.750	.080	-.135	.750	.122	-.095
.963	.037	-.113	.953	.047	-.136	.931	.057	-.138	.901	.092	-.101

$$M = 2.29; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$$

.037	.190	-.058	.048	.232	-.002	.069	.286	.051	.100	.228	-.003
.200	.102	-.084	.200	.168	-.002	.200	.220	.016	.200	.238	.009
.450	.070	-.095	.450	.079	-.096	.450	.132	-.054	.450	.160	-.012
.750	.022	-.102	.750			.750	.043	-.107	.750	.077	-.068
.963	.009	-.098	.953	.015	-.117	.931	.019	-.114	.901	.052	-.081

$$M = 2.29; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$$

.037	.117	-.036	.048	.166	.053	.069	.234	.118	.100	.147	.043
.200	.057	-.027	.200	.105	.053	.200	.166	.069	.200	.173	.056
.450	.035	-.060	.450	.035	-.056	.450	.077	-.015	.450	.116	.035
.750	-.009	-.077	.750			.750	.002	-.082	.750	.033	-.032
.963	-.018	-.079	.953	-.015	-.089	.931	-.018	-.096	.901	.010	-.052

$$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = -5^\circ$$

.037	.048	.040	.048	.107	.108	.069	.172	.180	.100	.088	.092
.200	.017	.005	.200	.040	.108	.200	.117	.102	.200	.102	.105
.450	-.010	-.021	.450	-.004	-.014	.450	.028	.022	.450	.073	.076
.750	-.036	-.051	.750			.750	-.043	-.047	.750	-.003	.002
.963	-.042	-.053	.953	-.048	-.059	.931	-.054	-.059	.901	-.027	-.020

$$M = 2.29; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$$

.037	.010	.103	.048	.051	.161	.069	.107	.234	.100	.043	.137
.200	-.024	.053	.200	-.002	.033	.200	.065	.169	.200	.047	.162
.450	-.039	.020	.450	-.049	.033	.450	-.010	.062	.450	.026	.115
.750	-.062	-.023	.750			.750	-.072	-.006	.750	-.035	.034
.963	-.065	-.029	.953	-.073	-.026	.931	-.085	-.021	.901	-.055	.009

$$M = 2.29; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$$

.037	-.035	.181	.048	.003	.221	.069	.051	.298	.100	.002	.208
.200	-.063	.105	.200	-.045	.083	.200	.016	.225	.200	-.001	.236
.450	-.071	.066	.450	-.086	.083	.450	-.046	.114	.450	-.017	.160
.750	-.083	.010	.750			.750	-.101	.040	.750	-.068	.075
.963	-.086	.000	.953	-.097	.007	.931	-.110	.021	.901	-.082	.051

TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
$M = 2.29; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	-.072	.259	.048	-.042	.285	.069	.006	.365	.100	-.034	.296
.200	-.102	.158	.200	-.080		.200	-.026	.281	.200	-.038	.303
.450	-.104	.114	.450	-.119	.131	.450	-.079	.170	.450	-.054	.205
.750	-.103	.042	.750			.750	-.125	.082	.750	-.096	.122
.963	-.105	.030	.953	-.121	.040	.931	-.129	.060	.901	-.104	.096
$M = 2.29; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	-.103	.336	.048	-.077	.369	.069	-.031	.443	.100	-.065	.392
.200	-.135	.215	.200	-.110		.200	-.059	.352	.200	-.078	.369
.450	-.133	.161	.450	-.143	.183	.450	-.106	.227	.450	-.084	.262
.750	-.122	.077	.750			.750	-.145	.125	.750	-.120	.171
.963	-.120	.063	.953	-.139	.076	.931	-.144	.106	.901	-.114	.140
$M = 2.29; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	-.132	.406	.048	-.110	.477	.069	-.069	.529	.100	-.093	.473
.200	-.168	.285	.200	-.140		.200	-.094	.420	.200	-.118	.435
.450	-.163	.198	.450	-.168	.240	.450	-.133	.279	.450	-.118	.317
.750	-.140	.111	.750			.750	-.166	.166	.750	-.143	.209
.963	-.133	.104	.953	-.153	.118	.931	-.161	.156	.901	-.126	.176
$M = 2.29; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.130	.401	.048	-.110	.473	.069	-.072	.531	.100	-.095	.476
.200	-.164	.276	.200	-.140		.200	-.096	.420	.200	-.122	.436
.450	-.161	.197	.450	-.165	.240	.450	-.134	.277	.450	-.120	.316
.750	-.137	.113	.750			.750	-.164	.169	.750	-.144	.205
.963	-.131	.103	.953	-.149	.116	.931	-.161	.158	.901	-.127	.179
$M = 2.29; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.107	.333	.048	-.085	.372	.069	-.038	.444	.100	-.069	.397
.200	-.143	.210	.200	-.119		.200	-.067	.354	.200	-.087	.372
.450	-.139	.161	.450	-.154	.182	.450	-.115	.227	.450	-.089	.264
.750	-.124	.077	.750			.750	-.153	.125	.750	-.127	.172
.963	-.125	.064	.953	-.141	.077	.931	-.149	.108	.901	-.118	.142
$M = 2.29; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.079	.252	.048	-.044	.283	.069	-.000	.367	.100	-.038	.304
.200	-.104	.155	.200	-.087		.200	-.031	.281	.200	-.042	.307
.450	-.101	.114	.450	-.125	.129	.450	-.083	.171	.450	-.059	.207
.750	-.104	.042	.750			.750	-.132	.081	.750	-.102	.122
.963	-.107	.030	.953	-.120	.040	.931	-.135	.061	.901	-.105	.097
$M = 2.29; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.042	.171	.048	-.007	.218	.069	.045	.301	.100	-.004	.218
.200	-.062	.103	.200	-.055		.200	.010	.223	.200	-.002	.243
.450	-.068	.066	.450	-.089	.081	.450	-.054	.113	.450	-.019	.161
.750	-.086	.009	.750			.750	-.110	.038	.750	-.073	.075
.963	-.090	.000	.953	-.100	.008	.931	-.117	.021	.901	-.088	.049
$M = 2.29; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.008	.092	.048	-.036	.162	.069	.099	.238	.100	.038	.149
.200	-.014	.051	.200	-.013		.200	.055	.168	.200	.051	.176
.450	-.037	.019	.450	-.044	.032	.450	-.021	.063	.450	.021	.117
.750	-.065	-.022	.750			.750	-.079	-.007	.750	-.045	.035
.963	-.070	-.028	.953	-.076	-.025	.931	-.087	-.018	.901	-.064	.009

TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.069	.035	.048	.089	.109	.069	.153	.180	.100	.088	.099
.200	.024	-.002	.200	.043	.109	.200	.102	.119	.200	.107	.122
.450	-.006	-.022	.450	-.004	-.018	.450	.021	.023	.450	.062	.076
.750	-.038	-.049	.750			.750	-.039	-.046	.750	-.010	.002
.963	-.046	-.054	.953	-.052	-.058	.931	-.050	-.058	.901	-.032	-.021

$M = 2.29; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.130	-.021	.048	.152	.061	.069	.218	.175	.100	.152	.053
.200	.062	-.044	.200	.104	.104	.200	.157	.072	.200	.175	.074
.450	.033	-.059	.450	.035	-.059	.450	.072	-.012	.450	.109	.039
.750	-.011	-.075	.750			.750	.001	-.078	.750	.028	-.030
.963	-.022	-.076	.953	-.019	-.087	.931	-.019	-.090	.901	.006	-.051

$M = 2.29; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.198	-.052	.048	.219	.013	.069	.283	.070	.100	.230	.007
.200	.107	-.091	.200	.165	.013	.200	.215	.027	.200	.240	.027
.450	.069	-.097	.450	.076	-.095	.450	.126	-.048	.450	.157	-.001
.750	.019	-.101	.750			.750	.041	-.104	.750	.073	-.063
.963	.006	-.095	.953	.013	-.118	.931	.017	-.112	.901	.049	-.077

$M = 2.29; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.198	-.051	.048	.219	.013	.069	.283	.070	.100	.230	.008
.200	.107	-.091	.200	.165	.013	.200	.215	.027	.200	.239	.028
.450	.069	-.097	.450	.076	-.095	.450	.127	-.048	.450	.157	-.001
.750	.020	-.101	.750			.750	.040	-.104	.750	.072	-.061
.963	.006	-.095	.953	.013	-.118	.931	.017	-.111	.901	.050	-.077

$M = 2.29; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.325	-.122	.048	.400	-.052	.069	.455	-.001	.100	.424	-.072
.200	.197	-.165	.200	.273		.200	.359	-.039	.200	.385	-.064
.450	.147	-.158	.450	.169	-.149	.450	.230	-.100	.450	.275	-.059
.750	.087	-.135	.750			.750	.121	-.144	.750	.165	-.106
.963	.068	-.127	.953	.079	-.171	.931	.104	-.146	.901	.131	-.112

$M = 2.29; \alpha = -10^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.384	-.145	.048	.481	-.080	.069	.551	-.035	.100	.514	-.112
.200	.247	-.186	.200	.324		.200	.426	-.069	.200	.454	-.103
.450	.186	-.182	.450	.219	-.168	.450	.281	-.122	.450	.325	-.089
.750	.123	-.158	.750			.750	.166	-.161	.750	.206	-.128
.963	.103	-.144	.953	.118	-.184	.931	.142	-.159	.901	.172	-.130

$M = 2.29; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$

.037	.063	-.005	.048	.033	.034	.069	.089	.068	.100	.104	.097
.200	.025	-.025	.200	.030		.200	.046	.028	.200	.078	.065
.450	.007	-.028	.450	.008	-.004	.450	.004	-.029	.450	.013	-.006
.750	-.035	-.049	.750			.750	-.038	-.049	.750	-.040	-.063
.963	-.043	-.046	.953	-.045	-.068	.931	-.039	-.060	.901	-.049	-.071

$M = 2.29; \alpha = 0^\circ; \beta = -6^\circ; \delta_e = 0^\circ$

.037	.061	-.003	.048	.045	.042	.069	.103	.092	.100	.124	.123
.200	.020	-.004	.200	.033		.200	.057	.044	.200	.093	.089
.450	.006	-.020	.450	.002	-.015	.450	.003	-.029	.450	.023	.009
.750	-.035	-.055	.750			.750	-.041	-.066	.750	-.037	-.053
.963	-.043	-.052	.953	-.043	-.073	.931	-.044	-.058	.901	-.049	-.070

TABLE V.- PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

M = 2.29; $\alpha = 0^\circ$; $\beta = -4^\circ$; $\delta_e = 0^\circ$

.037	.065	.006	.048	.066	.057	.069	.121	.118	.100	.140	.144
.200	.019	-.014	.200	.039	.031	.200	.073	.065	.200	.111	.111
.450	.006	-.018	.450	-.002	-.031	.450	.008	-.018	.450	.036	.027
.750	-.034	-.060	.750			.750	-.038	-.064	.750	-.030	-.041
.963	-.042	-.057	.953	-.039	-.072	.931	-.048	-.071	.901	-.045	-.059

M = 2.29; $\alpha = 0^\circ$; $\beta = -2^\circ$; $\delta_e = 0^\circ$

.037	.070	.011	.048	.077	.077	.069	.139	.142	.100	.126	.119
.200	.024	-.015	.200	.042	.042	.200	.089	.086	.200	.123	.124
.450	.003	-.027	.450	-.002	-.027	.450	.015	-.004	.450	.051	.045
.750	-.032	-.060	.750			.750	-.034	-.065	.750	-.019	-.026
.963	-.040	-.060	.953	-.038	-.067	.931	-.046	-.070	.901	-.038	-.047

M = 2.29; $\alpha = 0^\circ$; $\beta = -1^\circ$; $\delta_e = 0^\circ$

.037	.075	.017	.048	.085	.087	.069	.147	.154	.100	.103	.098
.200	.026	-.018	.200	.045	.045	.200	.099	.096	.200	.120	.115
.450	-.000	-.033	.450	-.001	-.031	.450	.020	.004	.450	.059	.053
.750	-.032	-.060	.750			.750	-.033	-.063	.750	-.013	-.019
.963	-.039	-.063	.953	-.038	-.068	.931	-.044	-.071	.901	-.032	-.040

M = 2.29; $\alpha = 0^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$

.037	.081	.028	.048	.094	.099	.069	.156	.164	.100	.092	.089
.200	.029	-.016	.200	.049	.049	.200	.107	.109	.200	.113	.108
.450	-.002	-.034	.450	.002	-.029	.450	.026	.013	.450	.068	.064
.750	-.031	-.057	.750			.750	-.029	-.056	.750	-.005	-.011
.963	-.037	-.063	.953	-.038	-.068	.931	-.041	-.067	.901	-.026	-.032

M = 2.29; $\alpha = 0^\circ$; $\beta = 1^\circ$; $\delta_e = 0^\circ$

.037	.083	.036	.048	.100	.105	.069	.165	.170	.100	.088	.086
.200	.032	-.015	.200	.052	.052	.200	.114	.116	.200	.100	.095
.450	-.005	-.038	.450	.005	-.028	.450	.032	.019	.450	.072	.070
.750	-.031	-.058	.750			.750	-.028	-.052	.750	-.000	-.005
.963	-.036	-.065	.953	-.041	-.072	.931	-.039	-.065	.901	-.022	-.026

M = 2.29; $\alpha = 0^\circ$; $\beta = 2^\circ$; $\delta_e = 0^\circ$

.037	.087	.047	.048	.110	.118	.069	.176	.178	.100	.090	.088
.200	.037	-.009	.200	.056	.056	.200	.124	.128	.200	.091	.088
.450	-.005	-.039	.450	.009	-.023	.450	.029	.029	.450	.076	.074
.750	-.031	-.057	.750			.750	-.024	-.046	.750	.008	.005
.963	-.033	-.063	.953	-.043	-.073	.931	-.035	-.060	.901	-.015	-.018

M = 2.29; $\alpha = 0^\circ$; $\beta = 4^\circ$; $\delta_e = 0^\circ$

.037	.095	.063	.048	.130	.142	.069	.166	.155	.100	.096	.096
.200	.046	.004	.200	.067	.067	.200	.144	.148	.200	.083	.079
.450	-.001	-.044	.450	.015	-.012	.450	.056	.046	.450	.075	.070
.750	-.031	-.062	.750			.750	-.016	-.034	.750	.021	.019
.963	-.028	-.062	.953	-.046	-.079	.931	-.031	-.051	.901	-.001	-.003

M = 2.29; $\alpha = 0^\circ$; $\beta = 6^\circ$; $\delta_e = 0^\circ$

.037	.104	.084	.048	.153	.169	.069	.143	.130	.100	.106	.108
.200	.057	.020	.200	.082	.082	.200	.163	.162	.200	.091	.088
.450	.008	-.040	.450	.023	.002	.450	.076	.067	.450	.073	.062
.750	-.030	-.072	.750			.750	-.006	-.018	.750	.035	.031
.963	-.021	-.063	.953	-.041	-.076	.931	-.020	-.036	.901	.015	.013

TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$$

.037	.121	.108	.048	.179	.195	.069	.136	.124	.100	.119	.122
.200	.069	.038	.200	.103	.103	.200	.168	.157	.200	.100	.100
.450	.019	-.030	.450	.033	.015	.450	.097	.088	.450	.073	.060
.750	-.020	-.072	.750			.750	.010	-.002	.750	.044	.038
.963	-.019	-.075	.953	-.032	-.068	.931	-.008	-.021	.901	.029	.026

$$M = 2.29; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.336	-.122	.048	.405	-.050	.069	.458	.002	.100	.428	-.071
.200	.201	-.173	.200	.276		.200	.364	-.036	.200	.389	-.060
.450	.151	-.178	.450	.170	-.151	.450	.232	-.098	.450	.276	-.058
.750	.088	-.143	.750			.750	.121	-.145	.750	.165	-.107
.963	.071	-.130	.953	.082	-.185	.931	.106	-.148	.901	.130	-.114

$$M = 2.29; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.276	-.093	.048	.301	-.018	.069	.358	.037	.100	.322	-.030
.200	.153	-.148	.200	.222		.200	.285	-.004	.200	.308	-.015
.450	.103	-.141	.450	.120	-.130	.450	.180	-.075	.450	.215	-.030
.750	.053	-.124	.750			.750	.078	-.127	.750	.119	-.086
.963	.037	-.114	.953	.047	-.161	.931	.056	-.133	.901	.088	-.099

$$M = 2.29; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.213	-.062	.048	.214	.024	.069	.282	.085	.100	.221	.013
.200	.110	-.109	.200	.169		.200	.212	.036	.200	.235	.035
.450	.067	-.100	.450	.077	-.100	.450	.128	-.043	.450	.156	.006
.750	.020	-.101	.750			.750	.041	-.104	.750	.075	-.057
.963	.007	-.100	.953	.016	-.126	.931	.017	-.115	.901	.049	-.076

$$M = 2.29; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.147	-.025	.048	.136	.070	.069	.206	.133	.100	.147	.064
.200	.074	-.069	.200	.102		.200	.149	.078	.200	.168	.083
.450	.029	-.061	.450	.040	-.066	.450	.065	-.009	.450	.104	.042
.750	-.011	-.075	.750			.750	.004	-.075	.750	.024	-.028
.963	-.020	-.078	.953	-.021	-.086	.931	-.014	-.091	.901	.001	-.048

$$M = 2.29; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.087	.026	.048	.073	.111	.069	.136	.180	.100	.093	.116
.200	.028	-.016	.200	.044		.200	.089	.121	.200	.110	.139
.450	-.005	-.019	.450	-.001	-.023	.450	.015	.025	.450	.053	.076
.750	-.039	-.051	.750			.750	-.035	-.044	.750	-.017	.002
.963	-.048	-.055	.953	-.051	-.057	.931	-.049	-.061	.901	-.037	-.021

$$M = 2.29; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.022	.088	.048	.027	.155	.069	.087	.235	.100	.041	.169
.200	-.004	.037	.200	-.004		.200	.046	.168	.200	.062	.193
.450	-.032	.018	.450	-.036	.028	.450	-.024	.062	.450	.016	.116
.750	-.060	-.022	.750			.750	-.071	-.008	.750	-.047	.034
.963	-.068	-.031	.953	-.075	-.026	.931	-.078	-.023	.901	-.065	.009

$$M = 2.29; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	-.033	.162	.048	-.016	.210	.069	.045	.299	.100	-.006	.228
.200	-.048	.090	.200	-.057		.200	.006	.222	.200	-.003	.248
.450	-.059	.062	.450	-.076	.076	.450	-.060	.110	.450	-.019	.158
.750	-.083	.007	.750			.750	-.108	.036	.750	-.077	.072
.963	-.087	-.002	.953	-.095	.005	.931	-.113	.018	.901	-.093	.048

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TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

M = 2.29; $\alpha = 6^\circ$; $\beta = 0^\circ$; $\delta_e = 5^\circ$

.037	-.077	.240	.048	-.045	.278	.069	-.004	.366	.100	-.040	.310
.200	-.091	.146	.200	-.088		.200	-.032	.278	.200	-.040	.307
.450	-.087	.109	.450	-.113	.126	.450	-.084	.168	.450	-.059	.204
.750	-.100	.040	.750			.750	-.131	.078	.750	-.100	.120
.963	-.106	.028	.953	-.112	.037	.931	-.135	.058	.901	-.106	.095

M = 2.29; $\alpha = 8^\circ$; $\beta = 0^\circ$; $\delta_e = 5^\circ$

.037	-.104	.326	.048	-.086	.372	.069	-.041	.445	.100	-.072	.402
.200	-.137	.209	.200	-.118		.200	-.069	.355	.200	-.089	.376
.450	-.129	.160	.450	-.151	.183	.450	-.117	.227	.450	-.089	.266
.750	-.122	.078	.750			.750	-.153	.127	.750	-.127	.172
.963	-.123	.064	.953	-.139	.077	.931	-.150	.109	.901	-.120	.144

M = 2.29; $\alpha = 10^\circ$; $\beta = 0^\circ$; $\delta_e = 5^\circ$

.037	-.134	.395	.048	-.116	.471	.069	-.076	.531	.100	-.100	.475
.200	-.168	.273	.200	-.146		.200	-.100	.420	.200	-.129	.436
.450	-.163	.195	.450	-.172	.238	.450	-.139	.277	.450	-.121	.317
.750	-.137	.110	.750			.750	-.171	.165	.750	-.149	.205
.963	-.134	.102	.953	-.151	.117	.931	-.165	.156	.901	-.130	.173

M = 2.75; $\alpha = -8^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$

.037	.287	-.080	.048	.349	-.010	.069	.301	-.053	.100	.271	-.049
.200	.179	-.125	.200	.262		.200	.322	-.051	.200	.256	-.056
.450	.123	-.135	.450	.160	-.108	.450	.231	-.065	.450	.235	-.082
.750	.077	-.116	.750			.750	.131	-.097	.750	.165	-.090
.963	.069	-.112	.953	.076	-.139	.931	.106	-.107	.901	.138	-.077

M = 2.75; $\alpha = -6^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$

.037	.225	-.073	.048	.280	.005	.069	.230	-.028	.100	.215	-.021
.200	.137	-.111	.200	.207		.200	.254	-.028	.200	.199	-.029
.450	.087	-.106	.450	.116	-.095	.450	.178	-.043	.450	.172	-.062
.750	.047	-.097	.750			.750	.094	-.083	.750	.119	-.077
.963	.038	-.093	.953	.043	-.125	.931	.071	-.091	.901	.096	-.065

M = 2.75; $\alpha = -4^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$

.037	.175	-.036	.048	.226	.036	.069	.173	.000	.100	.166	.008
.200	.101	-.066	.200	.158		.200	.187	-.004	.200	.150	-.002
.450	.055	-.084	.450	.081	-.065	.450	.131	-.020	.450	.113	-.039
.750	.019	-.084	.750			.750	.057	-.065	.750	.077	-.059
.963	.013	-.081	.953	.014	-.103	.931	.037	-.076	.901	.059	-.057

M = 2.75; $\alpha = -2^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$

.037	.119	-.013	.048	.188	.086	.069	.125	.034	.100	.120	.041
.200	.058	-.025	.200	.109		.200	.134	.029	.200	.105	.026
.450	.025	-.056	.450	.040	-.041	.450	.096	.013	.450	.069	-.011
.750	-.006	-.064	.750			.750	.022	-.039	.750	.039	-.032
.963	-.010	-.067	.953	-.014	-.077	.931	.004	-.056	.901	.027	-.034

M = 2.75; $\alpha = 0^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$

.037	.069	.047	.048	.140	.143	.069	.081	.074	.100	.079	.078
.200	.015	.006	.200	.067		.200	.078	.069	.200	.064	.062
.450	-.006	-.024	.450	.000	-.004	.450	.057	.050	.450	.028	.023
.750	-.028	-.038	.750			.750	-.006	-.013	.750	.002	-.002
.963	-.032	-.044	.953	-.040	-.052	.931	-.025	-.031	.901	-.006	-.006

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TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
$M = 2.75; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	.030	.094	.048	.081	.191	.069	.042	.122	.100	.042	.112
.200	-.015	.052	.200	.023		.200	.030	.118	.200	.030	.104
.450	-.036	.015	.450	-.029	.037	.450	.016	.088	.450	-.005	.063
.750	-.052	-.007	.750			.750	-.034	.015	.750	-.032	.034
.963	-.052	-.020	.953	-.062	-.020	.931	-.049	-.001	.901	-.036	.026
$M = 2.75; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	-.007	.147	.048	.031	.240	.069	.007	.172	.100	.010	.156
.200	-.049	.101	.200	-.012		.200	-.005	.172	.200	-.002	.150
.450	-.062	.053	.450	-.057	.080	.450	-.019	.126	.450	-.033	.108
.750	-.072	.023	.750			.750	-.057	.047	.750	-.060	.074
.963	-.069	.008	.953	-.084	.013	.931	-.065	.034	.901	-.058	.062
$M = 2.75; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	-.042	.209	.048	-.006	.297	.069	-.022	.228	.100	-.017	.205
.200	-.079	.152	.200	-.047		.200	-.035	.235	.200	-.030	.201
.450	-.088	.093	.450	-.083	.125	.450	-.049	.166	.450	-.058	.160
.750	-.091	.051	.750			.750	-.080	.087	.750	-.082	.116
.963	-.087	.033	.953	-.102	.044	.931	-.078	.070	.901	-.074	.097
$M = 2.75; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	-.066	.282	.048	-.038	.361	.069	-.047	.293	.100	-.041	.261
.200	-.105	.204	.200	-.072		.200	-.059	.307	.200	-.054	.258
.450	-.109	.136	.450	-.103	.172	.450	-.073	.216	.450	-.079	.226
.750	-.108	.082	.750			.750	-.098	.129	.750	-.099	.160
.963	-.102	.066	.953	-.113	.079	.931	-.092	.109	.901	-.083	.138
$M = 2.75; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	-.084	.356	.048	-.063	.430	.069	-.066	.378	.100	-.059	.327
.200	-.127	.254	.200	-.093		.200	-.079	.379	.200	-.073	.329
.450	-.129	.185	.450	-.120	.219	.450	-.091	.270	.450	-.097	.293
.750	-.121	.115	.750			.750	-.113	.172	.750	-.111	.207
.963	-.117	.105	.953	-.125	.117	.931	-.105	.148	.901	-.093	.181
$M = 2.75; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.082	.348	.048	-.066	.423	.069	-.066	.378	.100	-.058	.329
.200	-.123	.252	.200	-.093		.200	-.079	.376	.200	-.073	.330
.450	-.125	.183	.450	-.117	.217	.450	-.093	.270	.450	-.096	.293
.750	-.118	.113	.750			.750	-.111	.171	.750	-.111	.206
.963	-.113	.104	.953	-.120	.115	.931	-.105	.147	.901	-.093	.181
$M = 2.75; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.067	.276	.048	-.042	.355	.069	-.051	.290	.100	-.044	.261
.200	-.109	.200	.200	-.075		.200	-.064	.305	.200	-.057	.258
.450	-.113	.133	.450	-.108	.168	.450	-.076	.214	.450	-.082	.225
.750	-.108	.080	.750			.750	-.101	.129	.750	-.102	.159
.963	-.103	.067	.953	-.113	.079	.931	-.094	.108	.901	-.085	.137
$M = 2.75; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.040	.204	.048	-.011	.293	.069	-.027	.228	.100	-.021	.205
.200	-.083	.148	.200	-.050		.200	-.039	.236	.200	-.034	.201
.450	-.090	.091	.450	-.087	.124	.450	-.052	.165	.450	-.062	.162
.750	-.090	.049	.750			.750	-.085	.087	.750	-.084	.117
.963	-.088	.035	.953	-.102	.043	.931	-.080	.070	.901	-.075	.097

TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
$M = 2.75; \alpha = 4^0; \beta = 0^0; \delta_e = 0^0$											
.037	-.012	.141	.048	.023	.239	.069	.000	.175	.100	.004	.159
.200	-.053	.097	.200	-.019	.017	.200	-.012	.178	.200	-.009	.153
.450	-.063	.053	.450	-.065	.079	.450	-.024	.127	.450	-.040	.112
.750	-.073	.021	.750			.750	-.065	.048	.750	-.064	.078
.963	-.073	.007	.953	-.085	.012	.931	-.069	.034	.901	-.062	.062
$M = 2.75; \alpha = 2^0; \beta = 0^0; \delta_e = 0^0$											
.037	.027	.087	.048	.073	.189	.069	.036	.126	.100	.038	.113
.200	-.011	.047	.200	.017	.035	.200	.029	.127	.200	.026	.107
.450	-.033	.014	.450	-.033	.035	.450	.014	.091	.450	-.009	.068
.750	-.051	-.008	.750			.750	-.039	.015	.750	-.033	.039
.963	-.052	-.019	.953	-.062	-.020	.931	-.055	-.001	.901	-.037	.030
$M = 2.75; \alpha = 0^0; \beta = 0^0; \delta_e = 0^0$											
.037	.069	.045	.048	.117	.140	.069	.073	.081	.100	.072	.079
.200	.029	.002	.200	.053	.053	.200	.076	.083	.200	.058	.068
.450	-.008	-.021	.450	.005	-.004	.450	.050	.055	.450	.024	.030
.750	-.032	-.037	.750			.750	-.014	-.013	.750	-.001	.005
.963	-.033	-.042	.953	-.041	-.048	.931	-.028	-.029	.901	-.009	-.001
$M = 2.75; \alpha = -2^0; \beta = 0^0; \delta_e = 0^0$											
.037	.119	.005	.048	.171	.092	.069	.119	.040	.100	.113	.046
.200	.066	-.034	.200	.102	.043	.200	.131	.044	.200	.100	.032
.450	.024	-.054	.450	.043	-.038	.450	.091	.022	.450	.066	-.005
.750	-.008	-.062	.750			.750	.020	-.037	.750	.037	-.023
.963	-.012	-.061	.953	-.015	-.077	.931	.004	-.053	.901	.024	-.026
$M = 2.75; \alpha = -4^0; \beta = 0^0; \delta_e = 0^0$											
.037	.175	-.031	.048	.222	.051	.069	.167	.007	.100	.159	.013
.200	.104	-.069	.200	.151	.051	.200	.187	.010	.200	.144	.003
.450	.055	-.085	.450	.080	-.062	.450	.130	-.009	.450	.113	-.032
.750	.018	-.082	.750			.750	.056	-.058	.750	.077	-.050
.963	.011	-.076	.953	.013	-.100	.931	.038	-.072	.901	.059	-.049
$M = 2.75; \alpha = -6^0; \beta = 0^0; \delta_e = 0^0$											
.037	.239	-.054	.048	.283	.024	.069	.224	-.022	.100	.210	-.016
.200	.143	-.098	.200	.210	.024	.200	.249	-.020	.200	.195	-.025
.450	.088	-.106	.450	.117	-.084	.450	.180	-.036	.450	.168	-.056
.750	.045	-.096	.750			.750	.094	-.076	.750	.119	-.069
.963	.038	-.091	.953	.041	-.119	.931	.072	-.089	.901	.098	-.060
$M = 2.75; \alpha = -8^0; \beta = 0^0; \delta_e = 0^0$											
.037	.239	-.054	.048	.284	.025	.069	.225	-.023	.100	.212	-.016
.200	.143	-.098	.200	.211	.025	.200	.250	-.019	.200	.195	-.024
.450	.088	-.106	.450	.117	-.084	.450	.180	-.035	.450	.169	-.056
.750	.045	-.097	.750			.750	.095	-.076	.750	.119	-.069
.963	.039	-.090	.953	.042	-.119	.931	.071	-.089	.901	.098	-.059
$M = 2.75; \alpha = -10^0; \beta = 0^0; \delta_e = 0^0$											
.037	.354	-.092	.048	.447	-.025	.069	.400	-.067	.100	.336	-.065
.200	.229	-.130	.200	.321	-.025	.200	.398	-.068	.200	.337	-.074
.450	.165	-.140	.450	.213	-.112	.450	.281	-.079	.450	.310	-.096
.750	.113	-.125	.750			.750	.172	-.102	.750	.212	-.097
.963	.102	-.120	.953	.112	-.138	.931	.146	-.109	.901	.179	-.084

TABLE V.- PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
M = 2.75; $\alpha = 0^\circ$; $\beta = -8^\circ$; $\delta_e = 0^\circ$											
.037 .200 .450 .750 .963	.048 .025 .008 -.026 -.034	-.012 .004 -.004 -.036 -.036	.048 .200 .450 .750 .953	.052 .018 -.001 -.031	.057 .450 -.013 -.055	.069 .200 .450 .750 .931	.116 .070 .006 -.025 -.036	.095 .054 -.012 -.053 -.051	.100 .200 .450 .750 .901	.073 .085 .041 -.019 -.035	.058 .064 .026 -.028 -.044
M = 2.75; $\alpha = 0^\circ$; $\beta = -6^\circ$; $\delta_e = 0^\circ$											
.037 .200 .450 .750 .963	.051 .024 .006 -.026 -.033	.003 -.008 .003 -.043 -.040	.048 .200 .450 .750 .953	.073 .026 .004 -.030	.071 .450 -.017 -.049	.069 .200 .450 .750 .931	.118 .089 .016 -.026 -.034	.087 .073 .002 -.052 -.053	.100 .200 .450 .750 .901	.064 .060 .048 -.008 -.026	.056 .043 .031 -.016 -.033
M = 2.75; $\alpha = 0^\circ$; $\beta = -4^\circ$; $\delta_e = 0^\circ$											
.037 .200 .450 .750 .963	.058 .026 .004 -.023 -.030	.014 -.010 -.010 -.043 -.044	.048 .200 .450 .750 .953	.096 .041 .007 -.029	.087 .450 -.022 -.045	.069 .200 .450 .750 .931	.085 .100 .030 -.023 -.030	.065 .082 .016 -.043 -.054	.100 .200 .450 .750 .901	.070 .058 .040 .003 -.015	.063 .048 .025 -.005 -.020
M = 2.75; $\alpha = 0^\circ$; $\beta = -2^\circ$; $\delta_e = 0^\circ$											
.037 .200 .450 .750 .963	.068 .030 .001 -.023 -.030	.021 -.011 -.024 -.041 -.047	.048 .200 .450 .750 .953	.111 .052 .009 -.033	.111 .450 -.021 -.052	.069 .200 .450 .750 .931	.076 .089 .042 -.018 -.028	.065 .074 .031 -.034 -.047	.100 .200 .450 .750 .901	.069 .058 .032 .005 -.008	.063 .049 .020 -.003 -.011
M = 2.75; $\alpha = 0^\circ$; $\beta = -1^\circ$; $\delta_e = 0^\circ$											
.037 .200 .450 .750 .963	.073 .034 .000 -.024 -.029	.027 -.010 -.026 -.042 -.047	.048 .200 .450 .750 .953	.120 .058 .012 -.035	.123 .450 -.018 -.056	.069 .200 .450 .750 .931	.077 .085 .049 -.015 -.025	.066 .072 .037 -.028 -.043	.100 .200 .450 .750 .901	.071 .060 .029 .004 -.006	.066 .052 .018 -.004 -.010
M = 2.75; $\alpha = 0^\circ$; $\beta = 0^\circ$; $\delta_e = 0^\circ$											
.037 .200 .450 .750 .963	.080 .037 .001 -.022 -.028	.036 -.007 -.027 -.043 -.046	.048 .200 .450 .750 .953	.127 .065 .017 -.035	.133 .450 -.013 -.056	.069 .200 .450 .750 .931	.080 .084 .055 -.009 -.021	.070 .070 .044 -.021 -.036	.100 .200 .450 .750 .901	.077 .063 .029 .005 -.004	.071 .054 .018 -.003 -.008
M = 2.75; $\alpha = 0^\circ$; $\beta = 1^\circ$; $\delta_e = 0^\circ$											
.037 .200 .450 .750 .963	.083 .040 .002 -.024 -.027	.044 -.003 -.031 -.047 -.047	.048 .200 .450 .750 .953	.137 .071 .019 -.036	.145 .450 -.009 -.057	.069 .200 .450 .750 .931	.083 .081 .057 -.005 -.019	.072 .067 .046 -.016 -.030	.100 .200 .450 .750 .901	.081 .066 .030 .004 -.004	.075 .058 .018 -.005 -.009
M = 2.75; $\alpha = 0^\circ$; $\beta = 2^\circ$; $\delta_e = 0^\circ$											
.037 .200 .450 .750 .963	.087 .043 .004 -.024 -.027	.054 .002 -.030 -.051 -.048	.048 .200 .450 .750 .953	.149 .077 .023 -.035	.157 .450 -.004 -.059	.069 .200 .450 .750 .931	.087 .080 .060 .000 -.016	.076 .066 .048 -.009 -.026	.100 .200 .450 .750 .901	.086 .071 .032 .004 -.003	.079 .062 .020 -.005 -.009

TABLE V.- PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
$M = 2.75; \alpha = 0^\circ; \beta = 4^\circ; \delta_e = 0^\circ$											
.037	.101	.076	.048	.174	.183	.069	.097	.086	.100	.097	.091
.200	.053	.017	.200	.096		.200	.084	.072	.200	.081	.071
.450	.011	-.025	.450	.030	.010	.450	.066	.054	.450	.040	.028
.750	-.022	-.054	.750			.750	.013	.006	.750	.007	-.002
.963	-.024	-.051	.953	-.028	-.053	.931	-.004	-.011	.901	.001	-.006
$M = 2.75; \alpha = 0^\circ; \beta = 6^\circ; \delta_e = 0^\circ$											
.037	.119	.100	.048	.201	.209	.069	.109	.098	.100	.106	.102
.200	.065	.036	.200	.118		.200	.093	.081	.200	.089	.080
.450	.022	-.015	.450	.040	.028	.450	.075	.062	.450	.049	.036
.750	-.014	-.054	.750			.750	.029	.020	.750	.014	.003
.963	-.018	-.057	.953	-.019	-.044	.931	.010	.004	.901	.007	-.002
$M = 2.75; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$											
.037	.145	.128	.048	.228	.237	.069	.122	.113	.100	.116	.112
.200	.079	.057	.200	.143		.200	.103	.092	.200	.100	.093
.450	.037	-.001	.450	.057	.047	.450	.084	.072	.450	.058	.045
.750	-.003	-.045	.750			.750	.044	.033	.750	.020	.010
.963	-.007	-.050	.953	-.009	-.032	.931	.027	.021	.901	.014	.004
$M = 2.75; \alpha = -10^\circ; \beta = 0^\circ; \delta_e = 5^\circ$											
.037	.362	-.084	.048	.449	-.020	.069	.400	-.058	.100	.335	-.056
.200	.232	-.116	.200	.323		.200	.397	-.059	.200	.337	-.065
.450	.165	-.125	.450	.213	-.104	.450	.281	-.071	.450	.310	-.086
.750	.112	-.118	.750			.750	.171	-.096	.750	.211	-.089
.963	.101	-.112	.953	.110	-.124	.931	.145	-.101	.901	.178	-.077
$M = 2.75; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 5^\circ$											
.037	.309	-.073	.048	.362	.002	.069	.354	-.044	.100	.269	-.039
.200	.192	-.117	.200	.274		.200	.326	-.042	.200	.257	-.047
.450	.130	-.131	.450	.166	-.099	.450	.234	-.056	.450	.239	-.075
.750	.081	-.119	.750			.750	.135	-.090	.750	.170	-.083
.963	.072	-.108	.953	.078	-.131	.931	.110	-.100	.901	.143	-.071
$M = 2.75; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 5^\circ$											
.037	.250	-.053	.048	.284	.029	.069	.224	-.020	.100	.204	-.014
.200	.148	-.102	.200	.214		.200	.251	-.015	.200	.191	-.024
.450	.091	-.115	.450	.120	-.083	.450	.182	-.032	.450	.167	-.056
.750	.048	-.108	.750			.750	.096	-.075	.750	.120	-.068
.963	.040	-.098	.953	.043	-.124	.931	.073	-.089	.901	.098	-.059
$M = 2.75; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$											
.037	.189	-.027	.048	.224	.066	.069	.164	.010	.100	.155	.015
.200	.111	-.077	.200	.156		.200	.184	.015	.200	.139	.003
.450	.058	-.092	.450	.084	-.063	.450	.132	-.004	.450	.110	-.031
.750	.019	-.087	.750			.750	.058	-.057	.750	.076	-.047
.963	.014	-.085	.953	.014	-.106	.931	.042	-.072	.901	.059	-.045
$M = 2.75; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$											
.037	.123	.010	.048	.159	.105	.069	.113	.047	.100	.108	.049
.200	.074	-.048	.200	.095		.200	.126	.055	.200	.094	.034
.450	.025	-.060	.450	.047	-.037	.450	.086	.028	.450	.061	.000
.750	-.007	-.060	.750			.750	.020	-.035	.750	.034	-.018
.963	-.011	-.065	.953	-.013	-.081	.931	.006	-.051	.901	.022	-.022

TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 5^\circ$											
.037	.077	.043	.048	.101	.140	.069	.068	.086	.100	.068	.081
.200	.034	-.008	.200	.045		.200	.076	.095	.200	.055	.071
.450	-.005	-.021	.450	.008	-.005	.450	.044	.058	.450	.021	.036
.750	-.032	-.037	.750			.750	-.016	-.013	.750	-.002	.011
.963	-.035	-.044	.953	-.037	-.050	.931	-.028	-.030	.901	-.011	.001
$M = 2.75; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$											
.037	.027	.080	.048	.058	.183	.069	.031	.129	.100	.034	.116
.200	-.002	.040	.200	.007		.200	.031	.137	.200	.022	.110
.450	-.030	.013	.450	-.025	.032	.450	.010	.090	.450	-.012	.072
.750	-.052	-.009	.750			.750	-.043	.013	.750	-.034	.043
.963	-.053	-.022	.953	-.059	-.020	.931	-.056	-.003	.901	-.038	.032
$M = 2.75; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$											
.037	-.014	.131	.048	.025	.235	.069	-.002	.178	.100	.002	.160
.200	-.046	.091	.200	-.022		.200	-.011	.184	.200	-.010	.154
.450	-.058	.049	.450	-.067	.075	.450	-.023	.125	.450	-.040	.114
.750	-.068	.019	.750			.750	-.065	.046	.750	-.063	.079
.963	-.070	.006	.953	-.078	.013	.931	-.072	.032	.901	-.061	.063
$M = 2.75; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = 5^\circ$											
.037	-.039	.199	.048	-.014	.291	.069	-.029	.232	.100	-.026	.208
.200	-.079	.145	.200	-.051		.200	-.041	.242	.200	-.038	.204
.450	-.085	.090	.450	-.087	.123	.450	-.053	.166	.450	-.064	.166
.750	-.087	.049	.750			.750	-.084	.087	.750	-.085	.118
.963	-.084	.032	.953	-.097	.044	.931	-.081	.070	.901	-.075	.097
$M = 2.75; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = 5^\circ$											
.037	-.085	.345	.048	-.071	.425	.069	-.071	.383	.100	-.063	.331
.200	-.127	.254	.200	-.098		.200	-.084	.376	.200	-.077	.334
.450	-.128	.183	.450	-.124	.218	.450	-.098	.268	.450	-.100	.293
.750	-.122	.113	.750			.750	-.117	.171	.750	-.114	.206
.963	-.114	.103	.953	-.125	.115	.931	-.110	.148	.901	-.098	.181
$M = 3.22; \alpha = -10^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	.327	-.042	.048	.417	.007	.069	.314	-.030	.100	.313	-.025
.200	.211	-.062	.200	.313		.200	.317	-.043	.200	.292	-.030
.450	.150	-.073	.450	.197	-.060	.450	.269	-.051	.450	.239	-.053
.750	.097	-.080	.750			.750	.168	-.060	.750	.193	-.061
.963	.092	-.074	.953	.098	-.077	.931	.141	-.056	.901	.171	-.055
$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	.260	-.046	.048	.349	.015	.069	.257	-.030	.100	.253	-.020
.200	.172	-.074	.200	.258		.200	.248	-.042	.200	.236	-.027
.450	.118	-.085	.450	.157	-.067	.450	.215	-.053	.450	.184	-.055
.750	.072	-.086	.750			.750	.134	-.065	.750	.141	-.068
.963	.068	-.080	.953	.074	-.086	.931	.109	-.058	.901	.126	-.056
$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = -5^\circ$											
.037	.208	-.039	.048	.291	.030	.069	.206	-.011	.100	.200	-.003
.200	.133	-.076	.200	.205		.200	.192	-.024	.200	.183	-.013
.450	.085	-.088	.450	.120	-.061	.450	.163	-.042	.450	.137	-.043
.750	.046	-.085	.750			.750	.096	-.058	.750	.095	-.062
.963	.040	-.078	.953	.043	-.087	.931	.076	-.050	.901	.082	-.053

TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
M = 3.22; $\alpha = -4^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$											
.037	.165	-.019	.048	.233	.055	.069	.161	.011	.100	.155	.021
.200	.101	-.058	.200	.157	.055	.200	.143	-.002	.200	.140	.008
.450	.056	-.073	.450	.087	-.046	.450	.118	-.022	.450	.098	-.026
.750	.023	-.074	.750	.075	.055	.750	.064	-.044	.750	.059	-.048
.963	.018	-.068	.953	.021	-.079	.931	.045	-.038	.901	.048	-.045
M = 3.22; $\alpha = -2^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$											
.037	.128	-.005	.048	.189	.074	.069	.117	.039	.100	.112	.046
.200	.060	-.026	.200	.123	.074	.200	.101	.024	.200	.100	.029
.450	.028	-.051	.450	.050	-.024	.450	.080	.000	.450	.063	-.004
.750	.000	-.062	.750	.037	.055	.750	.037	-.027	.750	.027	-.032
.963	-.004	-.058	.953	-.003	-.066	.931	.018	-.029	.901	.018	-.035
M = 3.22; $\alpha = 0^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$											
.037	.091	.055	.048	.117	.101	.069	.080	.073	.100	.078	.071
.200	.026	.009	.200	.090	.101	.200	.064	.056	.200	.065	.062
.450	.001	-.022	.450	.022	.007	.450	.043	.028	.450	.032	.025
.750	-.020	-.039	.750	.010	-.022	.750	.010	-.002	.750	.000	-.008
.963	-.020	-.036	.953	-.022	-.043	.931	-.002	-.010	.901	-.005	-.013
M = 3.22; $\alpha = 2^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$											
.037	.058	.093	.048	.073	.147	.069	.049	.110	.100	.046	.099
.200	.002	.045	.200	.050	.147	.200	.034	.091	.200	.037	.094
.450	-.023	.010	.450	-.003	.036	.450	.010	.062	.450	.007	.055
.750	-.038	-.015	.750	.010	.036	.750	-.013	.024	.750	-.021	.018
.963	-.038	-.014	.953	-.042	-.018	.931	-.020	.010	.901	-.027	.009
M = 3.22; $\alpha = 4^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$											
.037	.024	.133	.048	.037	.200	.069	.020	.153	.100	.019	.138
.200	-.026	.091	.200	.015	.200	.200	.007	.135	.200	.011	.132
.450	-.044	.048	.450	-.028	.075	.450	-.015	.103	.450	-.017	.090
.750	-.057	.016	.750	.010	.075	.750	-.037	.053	.750	-.040	.049
.963	-.054	.011	.953	-.055	.012	.931	-.034	.035	.901	-.044	.040
M = 3.22; $\alpha = 6^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$											
.037	.002	.179	.048	.013	.258	.069	-.002	.199	.100	.000	.180
.200	-.049	.137	.200	-.011	.258	.200	-.014	.181	.200	-.010	.175
.450	-.061	.086	.450	-.047	.116	.450	-.034	.147	.450	-.035	.130
.750	-.068	.047	.750	-.064	.042	.750	-.053	.083	.750	-.055	.084
.963	-.066	.037	.953	-.064	.042	.931	-.046	.066	.901	-.052	.075
M = 3.22; $\alpha = 8^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$											
.037	-.012	.238	.048	-.009	.327	.069	-.020	.254	.100	-.018	.228
.200	-.066	.189	.200	-.031	.327	.200	-.031	.236	.200	-.027	.226
.450	-.075	.126	.450	-.061	.161	.450	-.050	.198	.450	-.048	.176
.750	-.079	.080	.750	-.074	.072	.750	-.064	.122	.750	-.065	.128
.963	-.073	.063	.953	-.074	.072	.931	-.058	.103	.901	-.058	.117
M = 3.22; $\alpha = 10^\circ$; $\beta = 0^\circ$; $\delta_e = -5^\circ$											
.037	-.024	.296	.048	-.023	.396	.069	-.035	.314	.100	-.033	.283
.200	-.080	.238	.200	-.048	.396	.200	-.046	.300	.200	-.042	.281
.450	-.086	.165	.450	-.075	.208	.450	-.063	.250	.450	-.061	.231
.750	-.089	.115	.750	-.083	.107	.750	-.075	.164	.750	-.074	.181
.963	-.080	.094	.953	-.083	.107	.931	-.070	.143	.901	-.062	.165

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TABLE V.- PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
$M = 3.22; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.025	.299	.048	-.027	.399	.069	-.036	.320	.100	-.040	.280
.200	-.081	.239	.200	-.053		.200	-.047	.307	.200	-.048	.283
.450	-.086	.166	.450	-.075	.210	.450	-.063	.254	.450	-.063	.237
.750	-.087	.116	.750			.750	-.075	.168	.750	-.074	.187
.963	-.076	.097	.953	-.082	.111	.931	-.071	.146	.901	-.062	.170
$M = 3.22; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.019	.236	.048	-.017	.328	.069	-.027	.260	.100	-.024	.230
.200	-.080	.189	.200	-.042		.200	-.039	.240	.200	-.033	.228
.450	-.085	.125	.450	-.073	.162	.450	-.060	.200	.450	-.055	.181
.750	-.087	.084	.750			.750	-.075	.123	.750	-.071	.133
.963	-.077	.068	.953	-.081	.076	.931	-.066	.107	.901	-.062	.122
$M = 3.22; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	-.005	.178	.048	-.002	.262	.069	-.009	.204	.100	-.006	.184
.200	-.063	.140	.200	-.023		.200	-.025	.184	.200	-.016	.180
.450	-.072	.088	.450	-.061	.118	.450	-.046	.152	.450	-.042	.134
.750	-.077	.050	.750			.750	-.063	.086	.750	-.063	.089
.963	-.072	.040	.953	-.073	.045	.931	-.053	.070	.901	-.058	.080
$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.019	.131	.048	.030	.205	.069	.011	.156	.100	.016	.141
.200	-.037	.091	.200	.004		.200	-.002	.137	.200	.005	.136
.450	-.053	.051	.450	-.040	.076	.450	-.026	.107	.450	-.024	.093
.750	-.062	.019	.750			.750	-.047	.056	.750	-.048	.053
.963	-.060	.015	.953	-.062	.015	.931	-.039	.038	.901	-.049	.045
$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.050	.093	.048	.067	.161	.069	.041	.115	.100	.042	.105
.200	-.008	.049	.200	.040		.200	.024	.099	.200	.030	.099
.450	-.025	.016	.450	-.015	.041	.450	.001	.070	.450	.000	.060
.750	-.045	-.007	.750			.750	-.024	.030	.750	-.028	.024
.963	-.043	-.007	.953	-.048	-.010	.931	-.027	.015	.901	-.034	.016
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.082	.054	.048	.123	.122	.069	.072	.077	.100	.070	.073
.200	.029	.010	.200	.074		.200	.056	.061	.200	.057	.064
.450	.000	-.018	.450	.014	.010	.450	.034	.037	.450	.025	.028
.750	-.023	-.034	.750			.750	.002	.004	.750	-.007	-.004
.963	-.025	-.031	.953	-.026	-.038	.931	-.010	-.008	.901	-.011	-.008
$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.119	.026	.048	.177	.094	.069	.108	.048	.100	.104	.052
.200	.064	-.014	.200	.114		.200	.091	.032	.200	.091	.036
.450	.027	-.037	.450	.046	-.010	.450	.071	.013	.450	.054	.005
.750	-.005	-.049	.750			.750	.029	-.014	.750	.020	-.021
.963	-.008	-.044	.953	-.008	-.052	.931	.014	-.020	.901	.011	-.028
$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.170	-.008	.048	.234	.066	.069	.152	.019	.100	.146	.025
.200	.101	-.041	.200	.157		.200	.137	.005	.200	.131	.012
.450	.053	-.060	.450	.083	-.033	.450	.114	-.012	.450	.090	-.017
.750	.017	-.067	.750			.750	.060	-.034	.750	.052	-.039
.963	.014	-.061	.953	.016	-.069	.931	.044	-.032	.901	.042	-.038

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TABLE V.- PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.218	-.023	.048	.288	.048	.069	.198	.000	.100	.190	.005
.200	.134	-.058	.200	.204		.200	.184	-.014	.200	.174	-.004
.450	.084	-.074	.450	.118	-.046	.450	.159	-.028	.450	.130	-.033
.750	.041	-.078	.750			.750	.093	-.046	.750	.091	-.052
.963	.037	-.070	.953	.040	-.077	.931	.075	-.039	.901	.079	-.044

$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.276	-.039	.048	.354	.029	.069	.253	-.019	.100	.243	-.014
.200	.174	-.077	.200	.262		.200	.247	-.032	.200	.226	-.021
.450	.119	-.087	.450	.157	-.060	.450	.215	-.043	.450	.180	-.048
.750	.071	-.087	.750			.750	.132	-.058	.750	.139	-.062
.963	.063	-.080	.953	.068	-.085	.931	.110	-.049	.901	.123	-.050

$M = 3.22; \alpha = -10^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.342	-.053	.048	.426	.009	.069	.314	-.037	.100	.305	-.030
.200	.219	-.084	.200	.319		.200	.322	-.050	.200	.286	-.036
.450	.155	-.094	.450	.204	-.072	.450	.273	-.059	.450	.241	-.063
.750	.101	-.095	.750			.750	.173	-.069	.750	.195	-.071
.963	.095	-.088	.953	.103	-.092	.931	.146	-.060	.901	.174	-.059

$M = 3.22; \alpha = 0^\circ; \beta = -8^\circ; \delta_e = 0^\circ$

.037	.062	.008	.048	.096	.088	.069	.066	.050	.100	.058	.051
.200	.025	-.013	.200	.036		.200	.068	.045	.200	.049	.038
.450	.004	-.007	.450	.007	-.019	.450	.037	.022	.450	.026	.011
.750	-.019	-.029	.750			.750	-.015	-.028	.750	.002	-.012
.963	-.024	-.031	.953	-.020	-.033	.931	-.024	-.040	.901	-.008	-.017

$M = 3.22; \alpha = 0^\circ; \beta = -4^\circ; \delta_e = 0^\circ$

.037	.070	.018	.048	.115	.102	.069	.067	.054	.100	.062	.055
.200	.025	-.013	.200	.050		.200	.061	.042	.200	.051	.042
.450	.002	-.018	.450	.010	-.017	.450	.041	.025	.450	.024	.012
.750	-.018	-.029	.750			.750	-.008	-.020	.750	.000	-.013
.963	-.021	-.033	.953	-.022	-.038	.931	-.020	-.035	.901	-.009	-.017

$M = 3.22; \alpha = 0^\circ; \beta = -2^\circ; \delta_e = 0^\circ$

.037	.083	.028	.048	.129	.117	.069	.073	.059	.100	.071	.062
.200	.032	-.010	.200	.064		.200	.060	.046	.200	.058	.048
.450	.005	-.023	.450	.014	-.009	.450	.043	.029	.450	.027	.016
.750	-.017	-.031	.750			.750	.002	-.011	.750	-.001	-.011
.963	-.020	-.033	.953	-.021	-.042	.931	-.013	-.025	.901	-.007	-.015

$M = 3.22; \alpha = 0^\circ; \beta = -1^\circ; \delta_e = 0^\circ$

.037	.088	.034	.048	.134	.119	.069	.077	.063	.100	.075	.066
.200	.036	-.007	.200	.071		.200	.061	.048	.200	.061	.051
.450	.006	-.027	.450	.018	-.005	.450	.042	.028	.450	.029	.018
.750	-.018	-.037	.750			.750	.005	-.007	.750	-.001	-.010
.963	-.020	-.034	.953	-.022	-.045	.931	-.009	-.020	.901	-.008	-.013

$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 0^\circ$

.037	.094	.040	.048	.138	.116	.069	.081	.067	.100	.078	.067
.200	.040	-.003	.200	.079		.200	.064	.051	.200	.066	.055
.450	.008	-.026	.450	.022	-.001	.450	.044	.029	.450	.033	.020
.750	-.018	-.040	.750			.750	.008	-.003	.750	.000	-.009
.963	-.020	-.036	.953	-.021	-.046	.931	-.006	-.016	.901	-.006	-.014

TABLE V.- PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	Cp at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$M = 3.22; \alpha = 0^\circ; \beta = 1^\circ; \delta_e = 0^\circ$

.037	.093	.040	.048	.138	.115	.069	.080	.067	.100	.077	.067
.200	.040	-.003	.200	.079	.200	.200	.063	.051	.200	.064	.054
.450	.008	-.027	.450	.022	-.001	.450	.043	.029	.450	.032	.020
.750	-.019	-.041	.750			.750	.007	-.004	.750	.000	-.010
.963	-.021	-.036	.953	-.021	-.046	.931	-.006	-.016	.901	-.007	-.014

$M = 3.22; \alpha = 0^\circ; \beta = 2^\circ; \delta_e = 0^\circ$

.037	.105	.058	.048	.133	.111	.069	.089	.075	.100	.084	.073
.200	.049	.008	.200	.056	.200	.200	.072	.059	.200	.070	.060
.450	.011	-.023	.450	.031	.013	.450	.048	.033	.450	.037	.026
.750	-.017	-.044	.750			.750	.014	.002	.750	.004	-.005
.963	-.019	-.041	.953	-.017	-.042	.931	.002	-.007	.901	-.003	-.011

$M = 3.22; \alpha = 0^\circ; \beta = 4^\circ; \delta_e = 0^\circ$

.037	.120	.081	.048	.136	.115	.069	.097	.084	.100	.087	.077
.200	.059	.023	.200	.116	.200	.200	.080	.067	.200	.075	.065
.450	.017	-.017	.450	.042	.026	.450	.053	.039	.450	.043	.030
.750	-.013	-.045	.750			.750	.021	.009	.750	.010	.000
.963	-.016	-.043	.953	-.012	-.035	.931	.011	.002	.901	.001	-.006

$M = 3.22; \alpha = 0^\circ; \beta = 6^\circ; \delta_e = 0^\circ$

.037	.141	.107	.048	.143	.121	.069	.109	.095	.100	.072	.064
.200	.072	.042	.200	.141	.200	.200	.091	.078	.200	.061	.052
.450	.032	-.004	.450	.058	.044	.450	.061	.047	.450	.040	.025
.750	-.004	-.039	.750			.750	.029	.017	.750	.012	.000
.963	-.007	-.040	.953	-.002	-.025	.931	.019	.010	.901	.006	-.003

$M = 3.22; \alpha = 0^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.037	.170	.140	.048	.154	.132	.069	.088	.068	.100	.070	.062
.200	.090	.066	.200	.169	.200	.200	.088	.070	.200	.056	.049
.450	.049	.012	.450	.080	.065	.450	.069	.046	.450	.026	.015
.750	.010	-.026	.750			.750	.043	.029	.750	.004	-.007
.963	.008	-.031	.953	.011	-.011	.931	.033	.023	.901	-.001	-.005

$M = 3.22; \alpha = 5^\circ; \beta = 8^\circ; \delta_e = 0^\circ$

.037	.077	.257	.048	.058	.241	.069	.030	.184	.100	.010	.147
.200	.016	.180	.200	.058	.200	.200	.022	.169	.200	.001	.134
.450	-.013	.097	.450	.016	.157	.450	.005	.138	.450	-.022	.086
.750	-.030	.040	.750			.750	-.016	.097	.750	-.036	.060
.963	-.030	.029	.953	-.030	.055	.931	-.016	.086	.901	-.029	.054

$M = 3.22; \alpha = 5^\circ; \beta = 6^\circ; \delta_e = 0^\circ$

.037	.054	.221	.048	.041	.229	.069	.033	.189	.100	.023	.165
.200	-.005	.155	.200	.045	.200	.200	.018	.170	.200	.013	.157
.450	-.030	.082	.450	-.004	.133	.450	-.008	.127	.450	-.018	.110
.750	-.046	.033	.750			.750	-.029	.092	.750	-.040	.073
.963	-.046	.022	.953	-.046	.043	.931	-.021	.082	.901	-.032	.066

$M = 3.22; \alpha = 5^\circ; \beta = 4^\circ; \delta_e = 0^\circ$

.037	.035	.194	.048	.031	.226	.069	.023	.188	.100	.015	.163
.200	-.021	.139	.200	.028	.200	.200	.003	.166	.200	.005	.157
.450	-.043	.074	.450	-.019	.119	.450	-.017	.129	.450	-.021	.112
.750	-.057	.028	.750			.750	-.037	.084	.750	-.042	.069
.963	-.057	.022	.953	-.052	.036	.931	-.028	.069	.901	-.036	.061

TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower
$M = 3.22; \alpha = 5^\circ; \beta = 2^\circ; \delta_e = 0^\circ$											
.037	.024	.172	.048	.025	.227	.069	.016	.180	.100	.014	.161
.200	-.030	.125	.200	.016	.200	.200	.002	.161	.200	.005	.156
.450	-.049	.069	.450	-.029	.106	.450	-.021	.126	.450	-.022	.111
.750	-.060	.028	.750			.750	-.042	.078	.750	-.045	.067
.963	-.060	.023	.953	-.056	.032	.931	-.034	.062	.901	-.042	.059
$M = 3.22; \alpha = 5^\circ; \beta = 1^\circ; \delta_e = 0^\circ$											
.037	.014	.163	.048	.019	.232	.069	.008	.181	.100	.010	.164
.200	-.040	.123	.200	.003	.200	.200	-.006	.161	.200	.000	.158
.450	-.055	.070	.450	-.039	.102	.450	-.029	.128	.450	-.027	.114
.750	-.064	.031	.750			.750	-.048	.076	.750	-.050	.070
.963	-.063	.027	.953	-.061	.031	.931	-.038	.058	.901	-.048	.061
$M = 3.22; \alpha = 5^\circ; \beta = 0^\circ; \delta_e = 0^\circ$											
.037	.008	.154	.048	.018	.233	.069	.003	.179	.100	.007	.164
.200	-.043	.117	.200	-.005	.200	.200	-.011	.161	.200	-.003	.157
.450	-.057	.069	.450	-.044	.099	.450	-.032	.129	.450	-.029	.113
.750	-.066	.033	.750			.750	-.050	.071	.750	-.052	.071
.963	-.064	.027	.953	-.063	.029	.931	-.042	.053	.901	-.050	.061
$M = 3.22; \alpha = 5^\circ; \beta = -1^\circ; \delta_e = 0^\circ$											
.037	.004	.146	.048	.016	.235	.069	-.003	.177	.100	.003	.159
.200	-.047	.113	.200	-.012	.200	.200	-.016	.159	.200	-.008	.156
.450	-.059	.069	.450	-.050	.094	.450	-.036	.128	.450	-.034	.113
.750	-.066	.036	.750			.750	-.054	.066	.750	-.056	.070
.963	-.065	.025	.953	-.064	.028	.931	-.046	.050	.901	-.054	.061
$M = 3.22; \alpha = 5^\circ; \beta = -2^\circ; \delta_e = 0^\circ$											
.037	.000	.141	.048	.012	.237	.069	-.009	.175	.100	-.003	.157
.200	-.051	.112	.200	-.020	.200	.200	-.022	.159	.200	-.012	.156
.450	-.060	.071	.450	-.054	.092	.450	-.040	.129	.450	-.039	.114
.750	-.066	.038	.750			.750	-.058	.063	.750	-.060	.073
.963	-.065	.024	.953	-.066	.030	.931	-.050	.047	.901	-.057	.064
$M = 3.22; \alpha = 5^\circ; \beta = -4^\circ; \delta_e = 0^\circ$											
.037	-.003	.129	.048	.002	.223	.069	-.017	.171	.100	-.012	.146
.200	-.054	.105	.200	-.031	.200	.200	-.028	.157	.200	-.020	.148
.450	-.059	.075	.450	-.061	.087	.450	-.043	.119	.450	-.045	.110
.750	-.064	.032	.750			.750	-.062	.053	.750	-.063	.072
.963	-.065	.021	.953	-.065	.031	.931	-.057	.041	.901	-.059	.061
$M = 3.22; \alpha = 5^\circ; \beta = -6^\circ; \delta_e = 0^\circ$											
.037	-.009	.122	.048	-.010	.205	.069	-.020	.167	.100	-.018	.136
.200	-.050	.102	.200	-.038	.200	.200	-.031	.158	.200	-.025	.140
.450	-.053	.082	.450	-.063	.085	.450	-.043	.108	.450	-.048	.108
.750	-.062	.024	.750			.750	-.067	.048	.750	-.065	.068
.963	-.065	.020	.953	-.065	.030	.931	-.064	.039	.901	-.059	.054
$M = 3.22; \alpha = 5^\circ; \beta = -8^\circ; \delta_e = 0^\circ$											
.037	-.011	.114	.048	-.018	.177	.069	-.024	.166	.100	-.020	.127
.200	-.049	.105	.200	-.045	.200	.200	-.022	.165	.200	-.027	.134
.450	-.049	.077	.450	-.064	.081	.450	-.045	.095	.450	-.048	.114
.750	-.062	.023	.750			.750	-.072	.048	.750	-.062	.061
.963	-.066	.016	.953	-.064	.021	.931	-.070	.036	.901	-.059	.045

TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING - Continued

Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

$$M = 3.22; \alpha = -10^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.352	-.048	.048	.434	.013	.069	.311	-.034	.100	.300	-.030
.200	.226	-.075	.200	.325		.200	.321	-.047	.200	.282	-.034
.450	.161	-.096	.450	.208	-.067	.450	.276	-.055	.450	.239	-.059
.750	.104	-.087	.750			.750	.177	-.064	.750	.196	-.065
.963	.098	-.079	.953	.106	-.081	.931	.149	-.057	.901	.175	-.057

$$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.291	-.041	.048	.360	.032	.069	.252	-.023	.100	.239	-.018
.200	.183	-.078	.200	.270		.200	.249	-.035	.200	.223	-.025
.450	.126	-.091	.450	.164	-.063	.450	.218	-.046	.450	.178	-.054
.750	.076	-.093	.750			.750	.138	-.061	.750	.141	-.065
.963	.069	-.085	.953	.076	-.085	.931	.113	-.051	.901	.126	-.051

$$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.233	-.024	.048	.296	.054	.069	.197	-.004	.100	.187	.001
.200	.144	-.067	.200	.211		.200	.187	-.017	.200	.171	-.010
.450	.090	-.086	.450	.125	-.051	.450	.162	-.030	.450	.130	-.039
.750	.048	-.090	.750			.750	.100	-.049	.750	.093	-.057
.963	.042	-.081	.953	.047	-.084	.931	.080	-.041	.901	.081	-.044

$$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.178	-.001	.048	.232	.076	.069	.147	.020	.100	.139	.024
.200	.106	-.051	.200	.158		.200	.132	.005	.200	.124	.010
.450	.056	-.072	.450	.087	-.034	.450	.112	-.010	.450	.088	-.020
.750	.021	-.076	.750			.750	.063	-.035	.750	.052	-.043
.963	.017	-.070	.953	.021	-.078	.931	.047	-.033	.901	.043	-.038

$$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.124	.032	.048	.169	.103	.069	.106	.048	.100	.100	.051
.200	.070	-.028	.200	.110		.200	.090	.033	.200	.088	.033
.450	.030	-.049	.450	.052	-.012	.450	.069	.013	.450	.054	.003
.750	-.002	-.058	.750			.750	.031	-.016	.750	.020	-.024
.963	-.004	-.052	.953	-.001	-.064	.931	.017	-.023	.901	.012	-.027

$$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.082	.061	.048	.119	.140	.069	.069	.083	.100	.068	.076
.200	.038	.003	.200	.068		.200	.053	.066	.200	.055	.068
.450	.006	-.018	.450	.017	.012	.450	.033	.043	.450	.024	.032
.750	-.020	-.034	.750			.750	.001	.007	.750	-.007	-.001
.963	-.023	-.032	.953	-.021	-.040	.931	-.010	-.006	.901	-.012	-.007

$$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.049	.083	.048	.078	.175	.069	.039	.118	.100	.042	.107
.200	.005	.042	.200	.035		.200	.026	.101	.200	.031	.101
.450	-.017	.012	.450	-.012	.038	.450	.004	.073	.450	.000	.062
.750	-.037	-.011	.750			.750	-.020	.030	.750	-.026	.025
.963	-.039	-.011	.953	-.039	-.015	.931	-.026	.012	.901	-.031	.016

$$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ; \delta_e = 5^\circ$$

.037	.023	.126	.048	.035	.214	.069	.012	.159	.100	.117	.146
.200	-.029	.047	.200	.006		.200	.000	.141	.200	.007	.141
.450	-.040	.049	.450	-.034	.076	.450	-.021	.110	.450	-.021	.097
.750	-.053	.017	.750			.750	-.040	.056	.750	-.044	.055
.963	-.051	.012	.953	-.052	.013	.931	-.037	.037	.901	-.047	.046

TABLE V. - PRESSURE COEFFICIENTS ON MISSILE WING - Concluded

Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$		Chord station $\frac{x_w}{c}$	C_p at $\frac{y}{b/2} = -$	
	.357 Upper	.366 Lower		.513 Upper	.522 Lower		.688 Upper	.698 Lower		.805 Upper	.815 Lower

 $M = 3.22; \alpha = 6^\circ; \beta = 0^\circ; \delta_e = 5^\circ$

.037	-.000	.174	.048	-.007	.266	.069	-.006	.208	.100	-.002	.190
.200	-.051	.137	.200	-.017	.200	.200	-.020	.188	.200	-.011	.185
.450	-.060	.085	.450	-.051	.118	.450	-.040	.154	.450	-.037	.138
.750	-.065	.049	.750	-.055	.075	.750	-.055	.087	.750	-.057	.090
.963	-.063	.038	.953	-.063	.043	.931	-.049	.069	.901	-.055	.081

 $M = 3.22; \alpha = 8^\circ; \beta = 0^\circ; \delta_e = 5^\circ$

.037	-.018	.231	.048	-.016	.326	.069	-.024	.261	.100	-.022	.233
.200	-.071	.184	.200	-.039	.200	.200	-.037	.242	.200	-.031	.231
.450	-.076	.123	.450	-.068	.161	.450	-.056	.200	.450	-.053	.182
.750	-.091	.080	.750	-.070	.071	.750	-.070	.121	.750	-.070	.132
.963	-.074	.064	.953	-.075	.071	.931	-.064	.104	.901	-.063	.122

 $M = 3.22; \alpha = 10^\circ; \beta = 0^\circ; \delta_e = 5^\circ$

.037	-.025	.295	.048	-.028	.399	.069	-.037	.324	.100	-.041	.281
.200	-.082	.235	.200	-.053	.210	.200	-.049	.310	.200	-.044	.286
.450	-.086	.163	.450	-.078	.210	.450	-.065	.255	.450	-.063	.239
.750	-.088	.115	.750	-.082	.109	.750	-.078	.167	.750	-.076	.188
.963	-.079	.095	.953	-.082	.109	.931	-.072	.146	.901	-.065	.171

TABLE VI. - PRESSURE COEFFICIENTS ON BOOSTER PITCH THRUST CHAMBER

(a) Without bug-eye fairings

Nozzle station $\frac{x_N}{L_N}$	Cp at $\theta =$ —			
	0°	90°	180°	270°
M = 2.75; $\alpha = -8^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; With jet simulator				
.119	.214	.035	-.089	-.089
.367	.007	.029	-.094	-.096
.731	-.094	-.067	-.098	-.105
M = 2.75; $\alpha = -6^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; With jet simulator				
.119	.115	.047	-.091	-.086
.367	-.016	.064	-.094	-.094
.731	-.096	-.037	-.101	-.107
M = 2.75; $\alpha = -4^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; With jet simulator				
.119	.003	.071	-.070	-.061
.367	-.058	.071	-.087	-.084
.731	-.089	.002	-.101	-.107
M = 2.75; $\alpha = -2^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; With jet simulator				
.119	-.033	.096	-.068	-.051
.367	-.042	.090	-.093	-.082
.731	-.082	.000	-.108	-.103
M = 2.75; $\alpha = 0^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; With jet simulator				
.119	-.028	.104	-.079	-.037
.367	-.033	.116	-.089	-.073
.731	-.075	-.005	-.010	-.098
M = 2.75; $\alpha = 2^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; With jet simulator				
.119	-.032	.108	-.077	-.028
.367	-.030	.111	-.073	-.065
.731	-.072	-.021	-.091	-.096
M = 2.75; $\alpha = 4^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; With jet simulator				
.119	-.051	.114	-.051	-.028
.367	-.026	.099	-.060	-.061
.731	-.072	-.037	-.084	-.096

Nozzle station $\frac{x_N}{L_N}$	Cp at $\theta =$ —			
	0°	90°	180°	270°
M = 2.75; $\alpha = -8^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; Without jet simulator				
.119	.075	.043	-.098	-.098
.367	-.007	.034	-.101	-.103
.731	-.096	-.066	-.105	-.112
M = 2.75; $\alpha = -6^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; Without jet simulator				
.119	-.016	.049	-.100	-.090
.367	-.026	.072	-.103	-.101
.731	-.096	-.036	-.108	-.114
M = 2.75; $\alpha = -4^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; Without jet simulator				
.119	-.064	.068	-.088	-.084
.367	-.055	.075	-.101	-.103
.731	-.090	-.005	-.110	-.115
M = 2.75; $\alpha = -2^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; Without jet simulator				
.119	-.059	.092	-.083	-.081
.367	-.040	.083	-.098	-.100
.731	-.081	-.005	-.113	-.112
M = 2.75; $\alpha = 0^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; Without jet simulator				
.119	-.060	.097	-.089	-.076
.367	-.031	.107	-.089	-.094
.731	-.077	-.009	-.112	-.110
M = 2.75; $\alpha = 2^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; Without jet simulator				
.119	-.067	.099	-.079	-.065
.367	-.031	.111	-.081	-.086
.731	-.072	-.026	-.106	-.106
M = 2.75; $\alpha = 4^\circ$; $\beta = 0^\circ$; $\delta_{PT} = -10^\circ$; $\delta_{YT} = 0^\circ$; Without jet simulator				
.119	-.069	.104	-.057	-.059
.367	-.029	.102	-.076	-.079
.731	-.076	-.040	-.101	-.107

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TABLE VI. - PRESSURE COEFFICIENTS ON BOOSTER PITCH THRUST CHAMBER - Continued

(a) Without bug-eye fairings - Concluded

Nozzle station $\frac{x_N}{L_N}$	C_p at $\theta = \text{---}$			
	0°	90°	180°	270°
$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ With jet simulator				
.119	.018	.070	-.060	-.050
.367	-.024	.075	-.073	-.075
.731	-.079	-.062	-.079	-.077
$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ With jet simulator				
.119	-.024	.083	-.058	-.056
.367	-.005	.062	-.073	-.073
.731	-.050	-.039	-.077	-.075
$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ With jet simulator				
.119	-.033	.073	-.058	-.054
.367	.011	.062	-.073	-.071
.731	-.041	-.026	-.081	-.077
$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ With jet simulator				
.119	-.026	.051	-.060	-.052
.367	.028	.043	-.071	-.071
.731	-.026	-.010	-.081	-.077
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ With jet simulator				
.119	-.026	.058	-.058	-.051
.367	.011	.049	-.068	-.068
.731	-.035	-.003	-.077	-.075
$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ With jet simulator				
.119	-.024	.062	-.050	-.043
.367	.011	.060	-.066	-.062
.731	-.035	-.016	-.075	-.073
$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ With jet simulator				
.119	-.026	.064	-.041	-.039
.367	.014	.070	-.062	-.058
.731	-.031	-.026	-.071	-.084

Nozzle station $\frac{x_N}{L_N}$	C_p at $\theta = \text{---}$			
	0°	90°	180°	270°
$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ Without jet simulator				
.119	-.030	.069	-.069	-.071
.367	-.022	.075	-.075	-.077
.731	-.073	-.058	-.079	-.075
$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ Without jet simulator				
.119	-.037	.079	-.069	-.071
.367	-.001	.058	-.077	-.077
.731	-.054	-.041	-.082	-.077
$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ Without jet simulator				
.119	-.033	.072	-.071	-.062
.367	.011	.062	-.077	-.075
.731	-.041	-.029	-.083	-.079
$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ Without jet simulator				
.119	-.027	.050	-.067	-.058
.367	.028	.043	-.073	-.075
.731	-.027	-.014	-.084	-.082
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ Without jet simulator				
.119	-.022	.055	-.058	-.058
.367	.020	.045	-.070	-.070
.731	-.035	-.001	-.079	-.077
$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ Without jet simulator				
.119	-.024	.060	-.052	-.052
.367	.014	.056	-.071	-.066
.731	-.033	-.014	-.077	-.075
$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$ Without jet simulator				
.119	-.026	.062	-.047	-.049
.367	.014	.068	-.068	-.064
.731	-.031	-.026	-.075	-.075

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TABLE VI. - PRESSURE COEFFICIENTS ON BOOSTER PITCH THRUST CHAMBER - Continued

(b) With bug-eye fairings

Nozzle station $\frac{x_N}{z_N}$	Cp at $\theta = -$			
	0°	90°	180°	270°
$M = 2.75; \alpha = -8^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.079	.083	-.084	-.088
.367	-.069	.172	-.098	-.105
.731	-.112	-.113	-.115	-.117
$M = 2.75; \alpha = -6^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.093	.190	-.100	-.081
.367	-.076	.185	-.101	-.100
.731	-.113	-.113	-.113	-.117
$M = 2.75; \alpha = -4^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.105	.171	-.100	-.074
.367	-.083	.212	-.105	-.098
.731	-.107	-.110	-.113	-.117
$M = 2.75; \alpha = -2^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.103	.208	-.069	-.059
.367	-.070	.171	-.077	-.089
.731	-.093	-.113	-.113	-.115
$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.098	.224	-.026	-.052
.367	-.055	.065	-.069	-.093
.731	-.067	-.113	-.115	-.115
$M = 2.75; \alpha = 2^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.091	.138	.034	-.060
.367	-.040	-.035	-.094	-.098
.731	-.060	-.117	-.115	-.113
$M = 2.75; \alpha = 4^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.086	.130	-.038	-.074
.367	-.033	-.077	-.094	-.098
.731	-.101	-.115	-.110	-.108

Nozzle station $\frac{x_N}{z_N}$	Cp at $\theta = -$			
	0°	90°	180°	270°
$M = 2.75; \alpha = -8^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	.012	.038	-.099	-.102
.367	-.019	.070	-.107	-.116
.731	-.068	-.106	-.116	-.121
$M = 2.75; \alpha = -6^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.036	.048	-.088	-.092
.367	-.031	.108	-.104	-.111
.731	-.057	-.092	-.116	-.119
$M = 2.75; \alpha = -4^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.071	.070	-.092	-.088
.367	-.052	.105	-.106	-.112
.731	-.033	-.059	-.118	-.119
$M = 2.75; \alpha = -2^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.071	.095	-.081	-.085
.367	-.035	.096	-.102	-.109
.731	-.014	-.025	-.120	-.120
$M = 2.75; \alpha = 0^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.071	.098	-.090	-.083
.367	-.023	.132	-.106	-.109
.731	.008	-.095	-.118	-.118
$M = 2.75; \alpha = 2^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.071	.103	-.102	-.090
.367	-.018	.141	-.109	-.114
.731	.017	-.111	-.118	-.118
$M = 2.75; \alpha = 4^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.071	.131	-.095	-.102
.367	-.002	.160	-.106	-.111
.731	.000	-.111	-.114	-.116

TABLE VI. - PRESSURE COEFFICIENTS ON BOOSTER PITCH THRUST CHAMBER - Continued

(b) With bug-eye fairings - Continued

Nozzle station $\frac{x_N}{z_N}$	Cp at $\theta =$ —			
	0°	90°	180°	270°
$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.061	.113	-.057	-.057
.367	-.040	.066	-.074	-.070
.731	-.072	-.083	-.081	-.081
$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.057	.149	-.063	-.055
.367	-.042	.095	-.072	-.066
.731	-.063	-.078	-.076	-.078
$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.053	.171	-.063	-.048
.367	-.023	.164	-.063	-.066
.731	-.051	-.078	-.076	-.078
$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.053	.156	-.023	-.042
.367	-.021	.158	-.051	-.061
.731	-.042	-.074	-.076	-.076
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.050	.170	-.006	-.040
.367	-.016	.110	-.059	-.061
.731	-.040	-.083	-.081	-.078
$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.044	.132	.005	-.040
.367	.001	.085	-.063	-.066
.731	-.020	-.083	-.081	-.076
$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.042	.132	-.036	-.053
.367	.021	-.005	-.068	-.070
.731	-.012	-.081	-.078	-.076

Nozzle station $\frac{x_N}{z_N}$	Cp at $\theta =$ —			
	0°	90°	180°	270°
$M = 3.22; \alpha = -8^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.040	.050	-.068	-.072
.367	-.031	.031	-.078	-.078
.731	-.035	-.066	-.081	-.083
$M = 3.22; \alpha = -6^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.031	.059	-.073	-.066
.367	-.003	.023	-.077	-.077
.731	-.012	-.068	-.068	-.081
$M = 3.22; \alpha = -4^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.031	.059	-.072	-.066
.367	-.003	.022	-.076	-.076
.731	-.012	-.070	-.068	-.081
$M = 3.22; \alpha = -2^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.033	.061	-.063	-.066
.367	.010	.044	-.076	-.074
.731	.001	-.068	-.078	-.081
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.031	.057	-.063	-.072
.367	.016	.072	-.076	-.081
.731	.029	-.038	-.083	-.085
$M = 3.22; \alpha = 2^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.031	.074	-.066	-.072
.367	.022	.104	-.074	-.083
.731	.031	-.072	-.083	-.083
$M = 3.22; \alpha = 4^\circ; \beta = 0^\circ;$ $\delta_{PT} = -10^\circ; \delta_{YT} = 0^\circ$				
.119	-.031	.076	-.072	-.074
.367	.029	.121	-.078	-.083
.731	.037	-.078	-.030	-.085

TABLE VI. - PRESSURE COEFFICIENTS ON BOOSTER PITCH THRUST CHAMBER - Continued

(b) With bug-eye fairings - Continued

Nozzle station $\frac{x_N}{z_N}$	Cp at $\theta =$ —			
	0°	90°	180°	270°
M = 2.75; $\alpha = 0^\circ$; $\beta = -8^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.038	.012	-.077	-.060
.367	-.036	-.081	-.108	-.094
.731	-.098	-.123	-.120	-.118
M = 2.75; $\alpha = 0^\circ$; $\beta = -6^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.024	.046	-.091	-.074
.367	-.011	-.046	-.110	-.110
.731	-.098	-.120	-.118	-.115
M = 2.75; $\alpha = 0^\circ$; $\beta = -4^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.045	.071	-.101	-.069
.367	-.044	-.014	-.096	-.101
.731	-.106	-.120	-.113	-.113
M = 2.75; $\alpha = 0^\circ$; $\beta = -2^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.088	.138	-.088	-.062
.367	-.059	-.016	-.098	-.101
.731	-.108	-.118	-.115	-.115
M = 2.75; $\alpha = 0^\circ$; $\beta = -1^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.104	.174	-.081	-.063
.367	-.072	.065	-.094	-.101
.731	-.104	-.117	-.117	-.115
M = 2.75; $\alpha = 0^\circ$; $\beta = 0^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.101	.231	-.043	-.053
.367	-.059	.068	-.070	-.094
.731	-.098	-.118	-.118	-.117
M = 2.75; $\alpha = 0^\circ$; $\beta = 1^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.093	.214	-.004	-.052
.367	-.042	.137	-.035	-.088
.731	-.066	-.119	-.117	-.113

Nozzle station $\frac{x_N}{z_N}$	Cp at $\theta =$ —			
	0°	90°	180°	270°
M = 2.75; $\alpha = 0^\circ$; $\beta = -8^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.030	.003	-.082	-.061
.367	.000	-.096	-.114	-.114
.731	-.100	-.126	-.121	-.121
M = 2.75; $\alpha = 0^\circ$; $\beta = -4^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.049	.068	-.103	-.070
.367	-.044	-.014	-.100	-.101
.731	-.108	-.121	-.114	-.114
M = 2.75; $\alpha = 0^\circ$; $\beta = -2^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.049	.069	-.103	-.068
.367	-.044	-.014	-.100	-.100
.731	-.108	-.121	-.114	-.114
M = 2.75; $\alpha = 0^\circ$; $\beta = -1^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.093	.175	-.082	-.061
.367	-.075	.057	-.087	-.100
.731	-.099	-.115	-.115	-.120
M = 2.75; $\alpha = 0^\circ$; $\beta = 0^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.100	.203	-.037	-.053
.367	-.056	.075	-.075	-.096
.731	-.078	-.117	-.117	-.115
M = 2.75; $\alpha = 0^\circ$; $\beta = 1^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.097	.209	.014	-.056
.367	-.039	.136	-.043	-.091
.731	-.067	-.116	-.117	-.116

TABLE VI. - PRESSURE COEFFICIENTS ON BOOSTER PITCH THRUST CHAMBER - Continued

(b) With bug-eye fairings - Continued

Nozzle station $\frac{x_N}{z_N}$	C _p at $\theta =$ —			
	0°	90°	180°	270°
M = 2.75; $\alpha = 0^\circ$; $\beta = 2^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.086	.202	.049	-.062
.367	-.036	.085	.001	-.086
.731	-.067	-.117	-.110	-.112
M = 2.75; $\alpha = 0^\circ$; $\beta = 4^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.076	.159	.085	-.079
.367	.013	.041	.034	-.096
.731	-.091	-.112	-.112	-.113
M = 2.75; $\alpha = 0^\circ$; $\beta = 6^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.076	.155	.113	-.100
.367	.012	.027	.089	-.108
.731	-.098	-.112	-.119	-.117
M = 2.75; $\alpha = 0^\circ$; $\beta = 8^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.074	.179	.114	-.101
.367	.012	.025	.090	-.110
.731	-.096	-.112	-.118	-.118
M = 3.22; $\alpha = 0^\circ$; $\beta = -8^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.025	.025	-.072	-.048
.367	-.001	-.023	-.087	-.053
.731	-.020	-.093	-.091	-.089
M = 3.22; $\alpha = 0^\circ$; $\beta = -6^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.012	.029	-.065	-.042
.367	.016	-.012	-.082	-.054
.731	-.025	-.086	-.084	-.084
M = 3.22; $\alpha = 0^\circ$; $\beta = -4^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 0^\circ$				
.119	-.008	.057	-.063	-.029
.367	.001	.033	-.080	-.063
.731	-.022	-.085	-.082	-.082

Nozzle station $\frac{x_N}{z_N}$	C _p at $\theta =$ —			
	0°	90°	180°	270°
M = 2.75; $\alpha = 0^\circ$; $\beta = 2^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.086	.199	.061	-.065
.367	-.033	.080	.003	-.089
.731	-.068	-.116	-.112	-.114
M = 2.75; $\alpha = 0^\circ$; $\beta = 6^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.091	.155	.105	-.112
.367	.012	.024	.085	-.117
.731	-.100	-.112	-.124	-.123
M = 2.75; $\alpha = 0^\circ$; $\beta = 8^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.074	.176	.096	-.105
.367	.010	.022	.068	-.115
.731	-.098	-.115	-.122	-.122
M = 3.22; $\alpha = 0^\circ$; $\beta = -8^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.023	.029	-.063	-.044
.367	-.001	-.014	-.083	-.057
.731	-.020	-.089	-.087	-.087
M = 3.22; $\alpha = 0^\circ$; $\beta = -6^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.010	.029	-.061	-.035
.367	.018	-.010	-.082	-.054
.731	-.027	-.086	-.084	-.084
M = 3.22; $\alpha = 0^\circ$; $\beta = -4^\circ$; $\delta_{PT} = 0^\circ$; $\delta_{YT} = 8^\circ$				
.119	-.010	.052	-.065	-.025
.367	.001	.031	-.080	-.065
.731	-.050	-.086	-.094	-.084

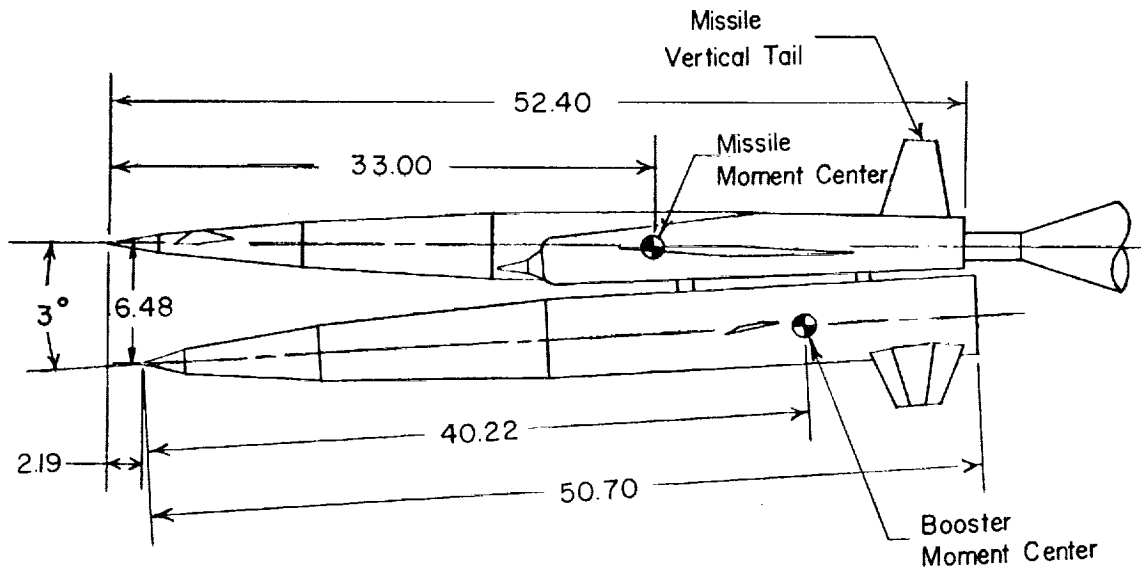
TABLE VI. - PRESSURE COEFFICIENTS ON BOOSTER PITCH THRUST CHAMBER - Concluded

(b) With bug-eye fairings - Concluded

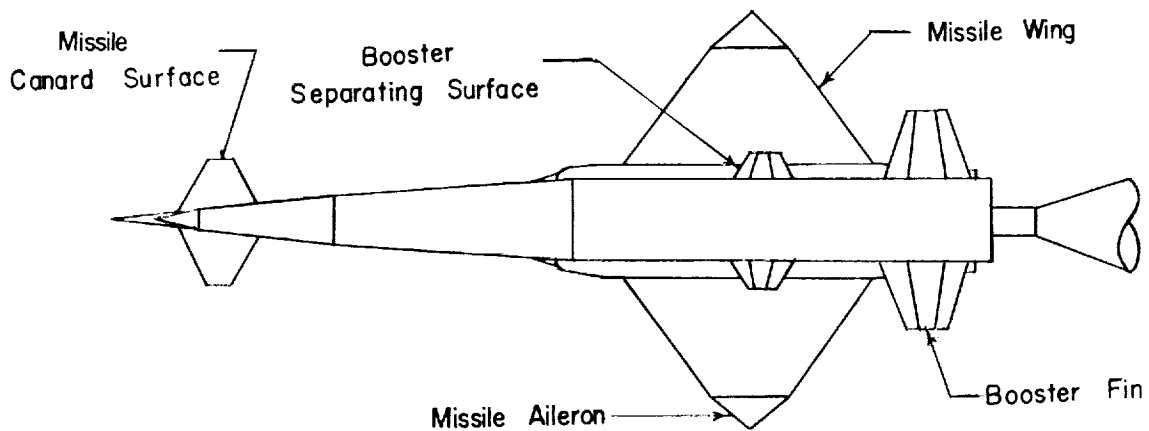
Nozzle station $\frac{x_N}{z_N}$	C_p at $\theta = -$			
	0°	90°	180°	270°
$M = 3.22; \alpha = 0^\circ; \beta = -2^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.027	.106	-.053	-.038
.367	-.010	.003	-.074	-.070
.731	-.059	-.087	-.082	-.082
$M = 3.22; \alpha = 0^\circ; \beta = -1^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.048	.127	-.057	-.040
.367	-.031	.108	-.072	-.066
.731	-.059	-.081	-.085	-.083
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.055	.161	-.025	-.035
.367	-.025	.108	-.063	-.063
.731	-.055	-.085	-.085	-.082
$M = 3.22; \alpha = 0^\circ; \beta = 1^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.053	.144	.001	-.040
.367	-.016	.117	-.048	-.063
.731	-.055	-.087	-.085	-.080
$M = 3.22; \alpha = 0^\circ; \beta = 2^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.048	.164	.054	-.046
.367	-.008	.136	-.025	-.060
.731	-.046	-.087	-.078	-.080
$M = 3.22; \alpha = 0^\circ; \beta = 4^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.042	.140	.076	-.065
.367	.016	.153	.061	-.076
.731	-.010	-.076	-.082	-.082
$M = 3.22; \alpha = 0^\circ; \beta = 6^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.033	.097	.131	-.076
.367	.044	.125	.071	-.082
.731	-.001	-.070	-.085	-.080
$M = 3.22; \alpha = 0^\circ; \beta = 8^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 0^\circ$				
.119	-.029	.108	.114	-.082
.367	.065	.097	.099	-.084
.731	.012	-.072	-.084	-.084

Nozzle station $\frac{x_N}{z_N}$	C_p at $\theta = -$			
	0°	90°	180°	270°
$M = 3.22; \alpha = 0^\circ; \beta = -2^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 8^\circ$				
.119	-.035	.107	-.057	-.059
.367	-.018	-.001	-.074	-.069
.731	-.063	-.086	-.084	-.082
$M = 3.22; \alpha = 0^\circ; \beta = -1^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 8^\circ$				
.119	-.054	.124	-.059	-.039
.367	-.035	.110	-.071	-.065
.731	-.061	-.080	-.084	-.082
$M = 3.22; \alpha = 0^\circ; \beta = 0^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 8^\circ$				
.119	-.054	.158	-.027	-.037
.367	-.025	.105	-.067	-.065
.731	-.054	-.084	-.084	-.082
$M = 3.22; \alpha = 0^\circ; \beta = 1^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 8^\circ$				
.119	-.052	.141	.001	-.042
.367	-.016	.116	-.050	-.065
.731	-.056	-.086	-.082	-.080
$M = 3.22; \alpha = 0^\circ; \beta = 2^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 8^\circ$				
.119	-.048	.160	.052	-.048
.367	-.005	.135	-.027	-.065
.731	-.050	-.086	-.080	-.080
$M = 3.22; \alpha = 0^\circ; \beta = 4^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 8^\circ$				
.119	-.042	.135	.069	-.067
.367	.018	.149	.069	-.078
.731	-.014	-.075	-.084	-.082
$M = 3.22; \alpha = 0^\circ; \beta = 6^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 8^\circ$				
.119	-.035	.092	.118	-.080
.367	.041	.122	.075	-.086
.731	-.005	-.071	-.090	-.084
$M = 3.22; \alpha = 0^\circ; \beta = 8^\circ;$ $\delta_{PT} = 0^\circ; \delta_{YT} = 8^\circ$				
.119	-.029	.103	.109	-.086
.367	.065	.094	.094	-.088
.731	.005	-.073	-.088	-.088

Side view

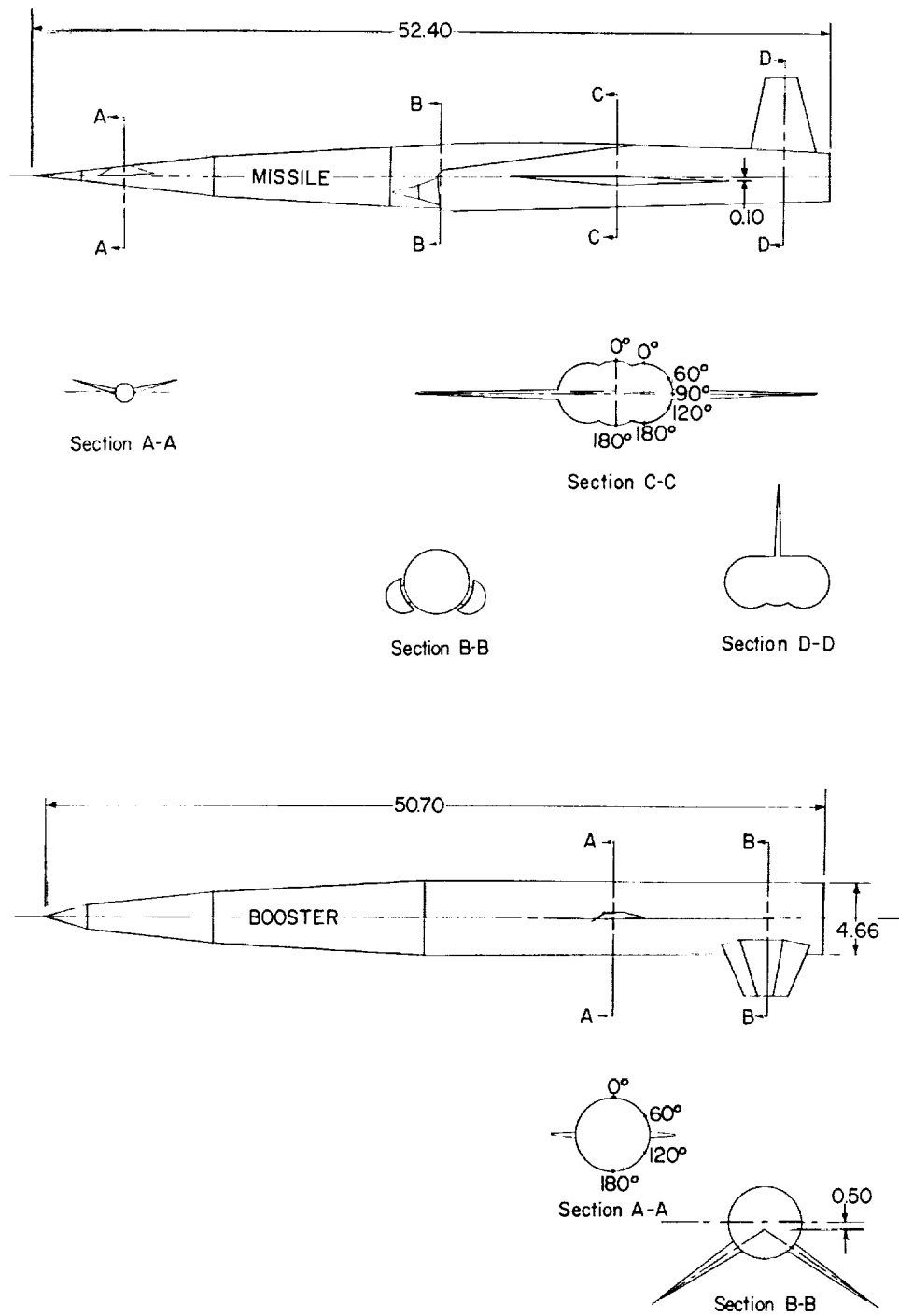


Bottom view



(a) General views.

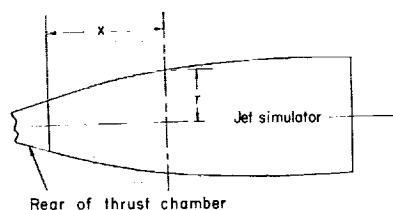
Figure 1.- Sketches of 0.05-scale model of SXM-64A missile. All dimensions in inches.



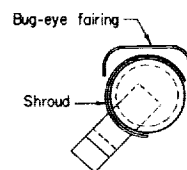
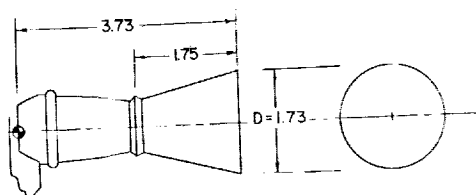
(b) Body details.

Figure 1.- Continued.

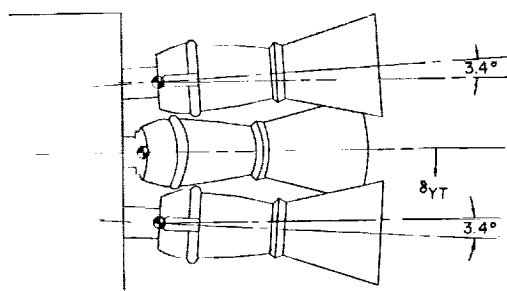
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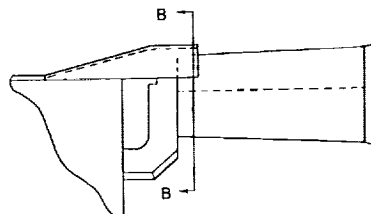
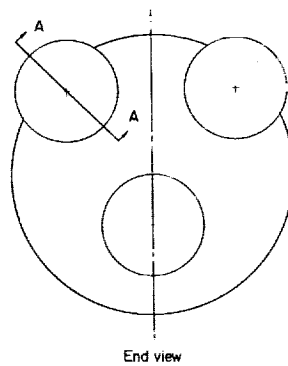
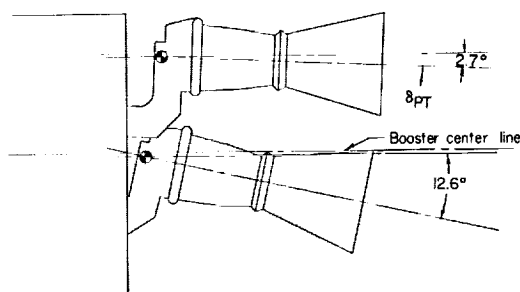
x	r	x	r
0	0.865	5.046	1.860
0.721	1.095	5.766	1.900
1.441	1.298	6.488	1.931
2.163	1.460	7.208	1.954
2.883	1.600	7.929	1.960
3.604	1.708	8.650	1.960
4.325	1.794	9.371	1.954
		10.091	1.946



Section B-B



Top view

Chamber modifications
Viewed at section A-A

End view

(c) Booster thrust chambers.

Figure 1.- Concluded.

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